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AN 01-20EF-1
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PILOT'S FLIGHT OPERATING INSTRUCTIONS

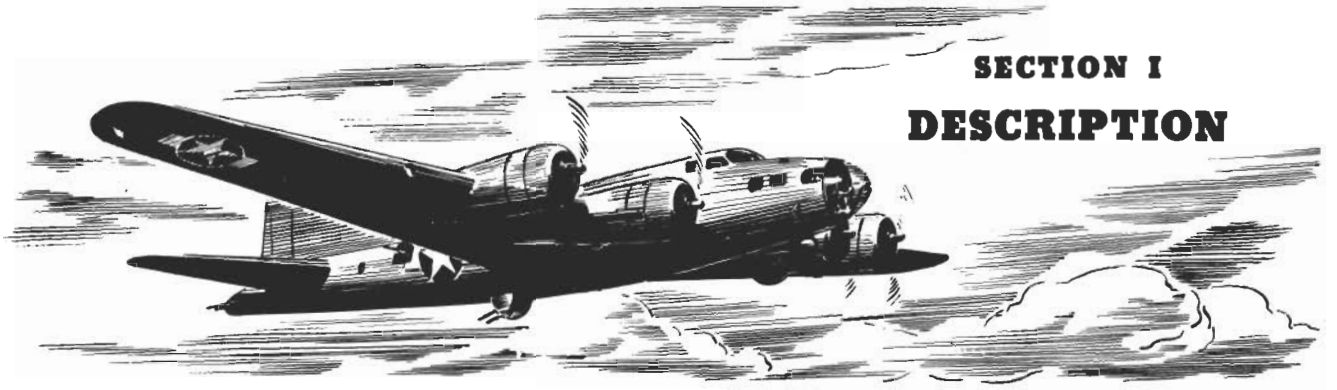
FOR

ARMY MODELS
B-17F and G

BRITISH MODEL
FORTRESS II

This publication shall not be carried in aircraft on combat missions or when there is a reasonable chance of its falling into the hands of the enemy.

AUGUST 1, 1943



SECTION I
DESCRIPTION

Figure 1 - B-17F in Flight

1. AIRPLANE.

a. Model B-17F and G bombardment airplanes are four-engine-midwing monoplanes. The approximate over-all dimensions are: length, 74 feet 9 inches; height, taxying position, 19 feet 1 inch; span, 103 feet 9 inches.

b. Electrically operated landing gear, tail gear, wing flaps, bomb bay doors, and hydraulically operated brakes and cowl flaps are provided.

c. The crew includes pilot, copilot, navigator, bombardier, upper turret gunner, lower turret gunner, radio operator, side gunner(s), and tail gunner. The airplane can be entered either through the main entrance door on the right side of the airplane just forward of the horizontal stabilizer, or through the front hatch in the bottom of the fuselage below the pilot's compartment.

d. Defensive armament of the B-17F consists of three turrets, each mounting two .50 calibre machine guns, and five single flexibly mounted .50 mounted .50 calibre machine guns. The B-17G has an additional power turret just below the nose of the airplane and controlled from the bombardier's compartment.

e. Provisions are made for loading 2000-pound or smaller bombs on racks within the bomb bay, and one bomb, up to 4000 pounds may be carried under each wing.

f. Automatic flight control equipment is provided.

2. POWER PLANT.

a. ENGINES. - The Wright model R-1820-97 engines are air-cooled, nine-cylinder radial aircraft engines, equipped with integral reduction gears through which the propellers are driven.

b. TURBOSUPERCHARGERS. - A type B-2 General Electric turbosupercharger is provided for each engine to boost manifold pressure for take-off and high-altitude flight. Superchargers are controlled by automatic hydraulic regulators adjusted from the pilot's control pedestal.



Figure 3 - Power Plant

c. PROPELLERS. - The Hamilton standard three-blade propellers are hydromatically controlled with constant-speed and full feathering provisions.

d. AUTOMATIC ENGINE CONTROL. - Should engine control cables be shot away, four of the controls will automatically assume predetermined positions: throttles, wide open; superchargers, 65 percent power; intercoolers, cold; and propellers, 1850 rpm. Functioning of the automatic control at one unit will not affect placement of controls at other units, or of similar controls on other engines.



Figure 2
Three-quarter
Rear View

3. HYDRAULIC SYSTEM

a. SERVICE SYSTEM. - Hydraulic pressure for operating brakes and cowl flaps is supplied by an electric motor-driven pump, or by an accumulator while the pump is not operating.

(1) When the hydraulic pump switch on the pilot's control panel is in the "AUTO" position, pressure is automatically regulated by a pressure cut-out switch, starting the pump when pressure drops to 600 pounds and stopping the pump when the pressure builds up to 800 pounds. In case the automatic pressure switch fails, pressure may be maintained by holding the hydraulic pump switch in the "MANUAL" position. A relief valve opens if pressure in the system reaches 900 pounds.

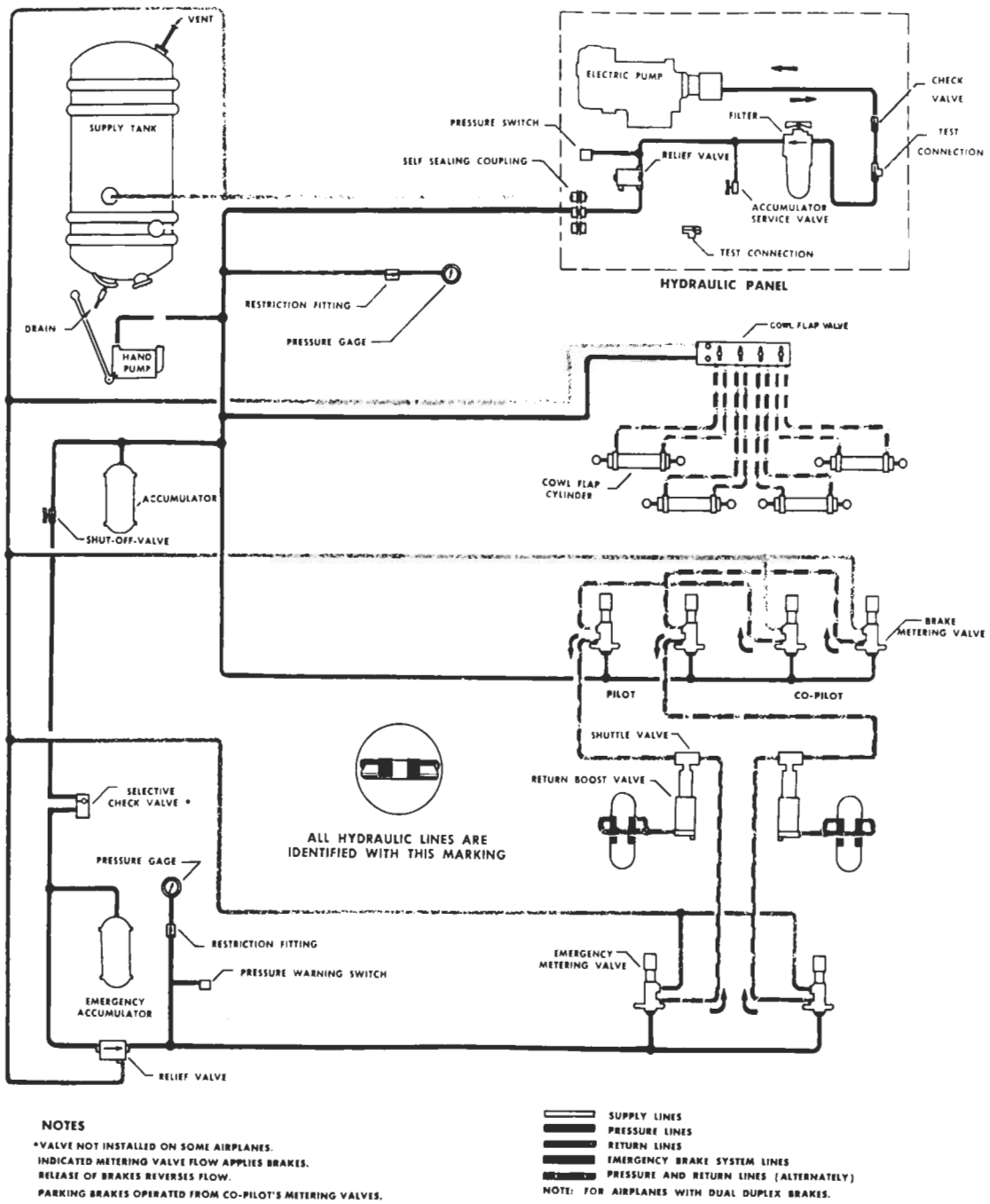


Figure 4 - Hydraulic Flow Diagram

WARNING

Should leakage occur in the hydraulic system, the pump must be stopped to prevent loss of fluid. Remove the hydraulic pump switch fuse in the station 4 fuse panel, or disconnect the electrical receptacle at the pressure switch.

(2) In some airplanes the hydraulic pump is controlled by an "ON-OFF" switch on the pilot's control panel. This switch must be "ON" to maintain pressure automatically.

b. EMERGENCY BRAKE SYSTEM. - A spare accumulator and auxiliary metering valves provide emergency brake operation. A red warning lamp on the pilot's instrument panel lights when pressure in the emergency system falls to approximately 700 pounds per square inch. To charge the emergency accumulator, open the manual shut-off valve. If a selective check valve is installed, place it in "SERVICING" position, unless it is lockwired in "NORMAL" position. (These units are located on the right side wall at the rear of the control cabin. See figure 5.) Build up 800 pounds pressure in the system, then return the selective check valve to "NORMAL" position and close the manual shut-off valve.

NOTE

The emergency brake system has been eliminated from the later model airplanes.

c. PRESSURE GAGES. - Pressure in the service and emergency brake systems is indicated by two gages on the pilot's instrument panel.

d. HAND PUMP. - A hand pump on the side wall at the right of the copilot is used to supply pressure for ground service operations, and to recharge the accumulators if the electric pump fails.

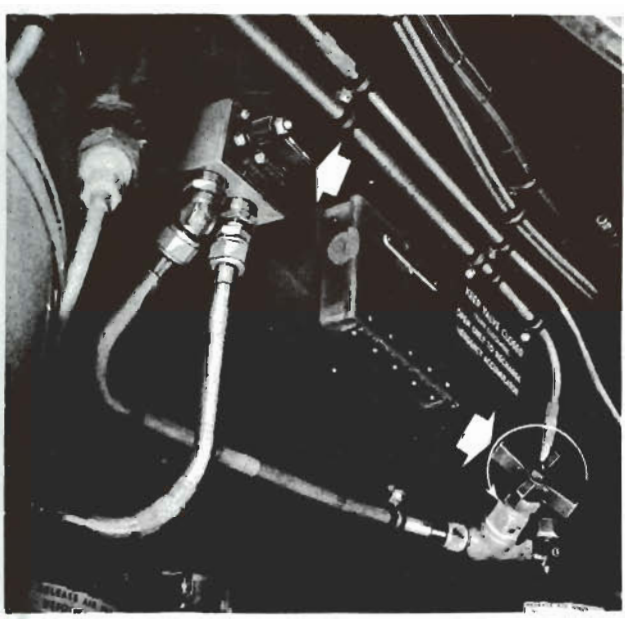


Figure 5 - Servicing Emergency Accumulator

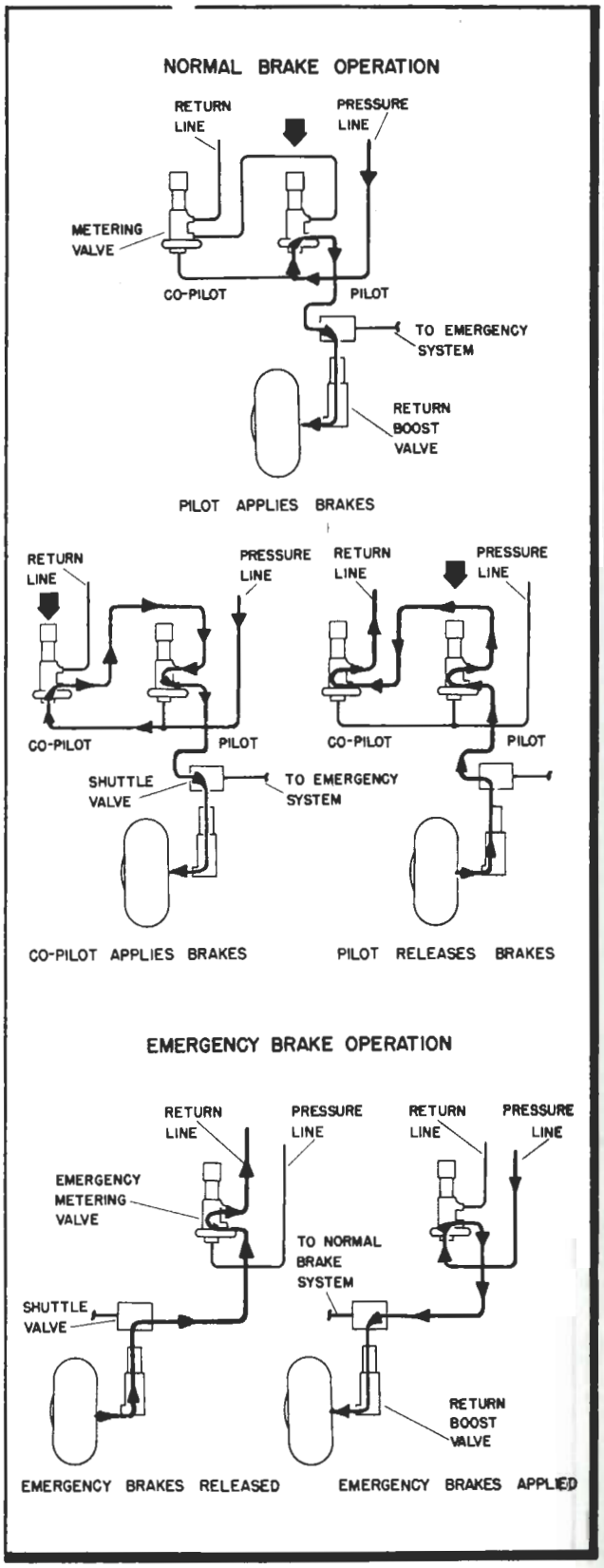


Figure 6 - Brake Operation Diagram

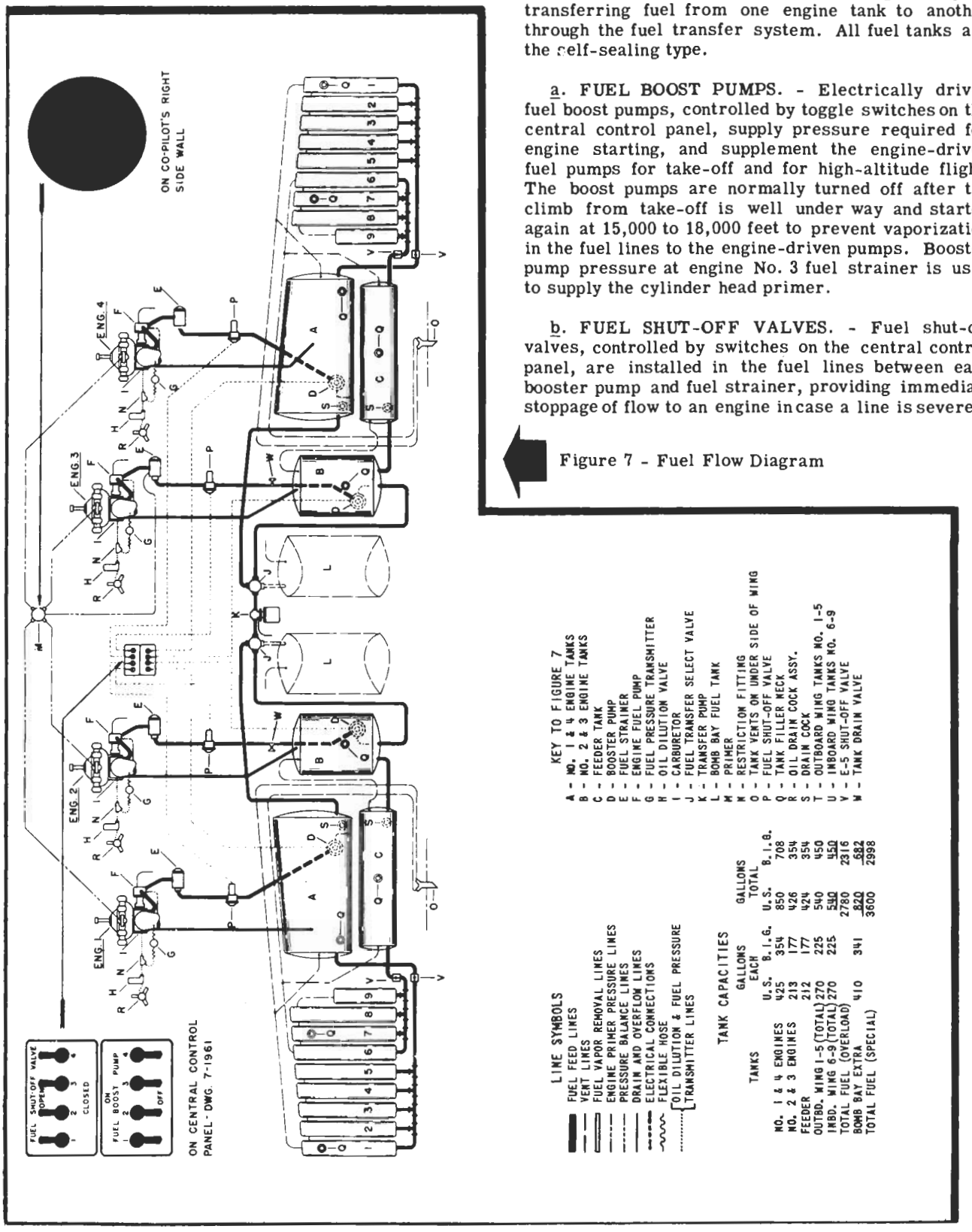
4. FUEL SYSTEM

The fuel system consists of four independent single-engine systems as shown in figure 7. The fuel supply for one engine can be used for another engine only by transferring fuel from one engine tank to another through the fuel transfer system. All fuel tanks are the self-sealing type.

a. FUEL BOOST PUMPS. - Electrically driven fuel boost pumps, controlled by toggle switches on the central control panel, supply pressure required for engine starting, and supplement the engine-driven fuel pumps for take-off and for high-altitude flight. The boost pumps are normally turned off after the climb from take-off is well under way and started again at 15,000 to 18,000 feet to prevent vaporization in the fuel lines to the engine-driven pumps. Booster pump pressure at engine No. 3 fuel strainer is used to supply the cylinder head primer.

b. FUEL SHUT-OFF VALVES. - Fuel shut-off valves, controlled by switches on the central control panel, are installed in the fuel lines between each booster pump and fuel strainer, providing immediate stoppage of flow to an engine in case a line is severed.

Figure 7 - Fuel Flow Diagram



c. PRIMER. - The cylinder head primer has positions corresponding to each of the four engines, and an "OFF" position in which the primer handle is locked. To operate, push the handle down, turn the valve to the engine position required, and then withdraw the handle and pump the charge to the engine.

IMPORTANT

Pressure from No. 3 fuel booster pump is on the suction side of the primer and overpriming will result, if the handle is left in the withdrawn position. Therefore, each priming operation must terminate with the handle returned to the locked position.

d. FUEL TRANSFER SYSTEM.

(1) Fuel is transferred by means of an electric motor-driven pump and two selector valves. The motor switch and selector valve handles are in the rear of the control cabin below the door leading to the bomb bay. Direct transfer can only be made across the center line of the airplane. (See figure 8 for fuel transfer procedure.)

WARNING

Do not use bomb bay valve position when bomb bay tanks are not installed. It is recommended that a 6-inch length of hose, plugged at the outer end, be attached to the bomb bay valve ports.

(2) An emergency hand-operated fuel pump, mounted on the rear bulkhead of the bomb bay, can be substituted for the electric-driven transfer pump by disconnecting the electric pump lines from the fuel transfer selector valves at the forward end of the bomb bay and connecting the hand pump lines. The hand pump can also be used as a refueling pump. (See figure 60.)

(3) Airplanes equipped with auxiliary wing fuel cells have shut-off valves in the lines leading from each group of cells. These valves are controlled by handles in the radio compartment or in the bomb bay near bulkhead No. 5. (See figure 59.) Keep auxiliary cell shut-off valves "CLOSED" (handles out) at all times except when transferring fuel from auxiliary to main tanks. Transfer fuel only when fuel level of main tanks has dropped to 100 gallons per engine. After transfer, return valve to "CLOSED" (handle out) position.

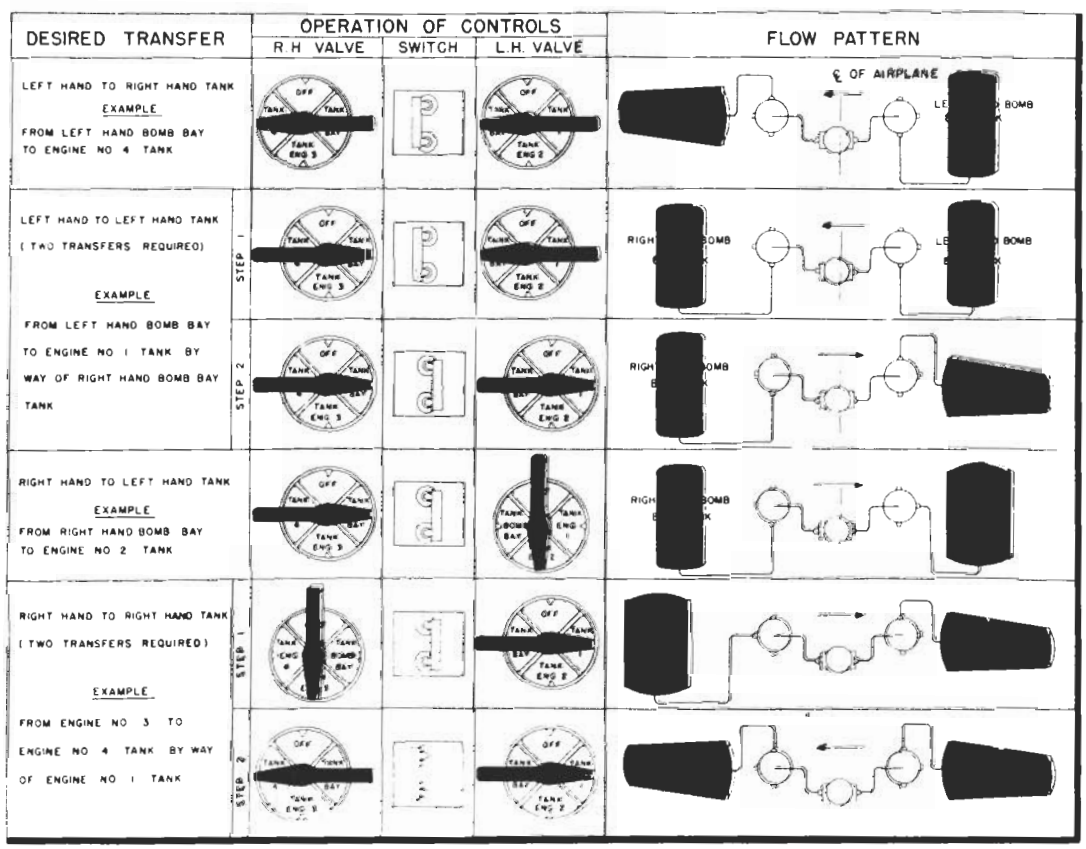
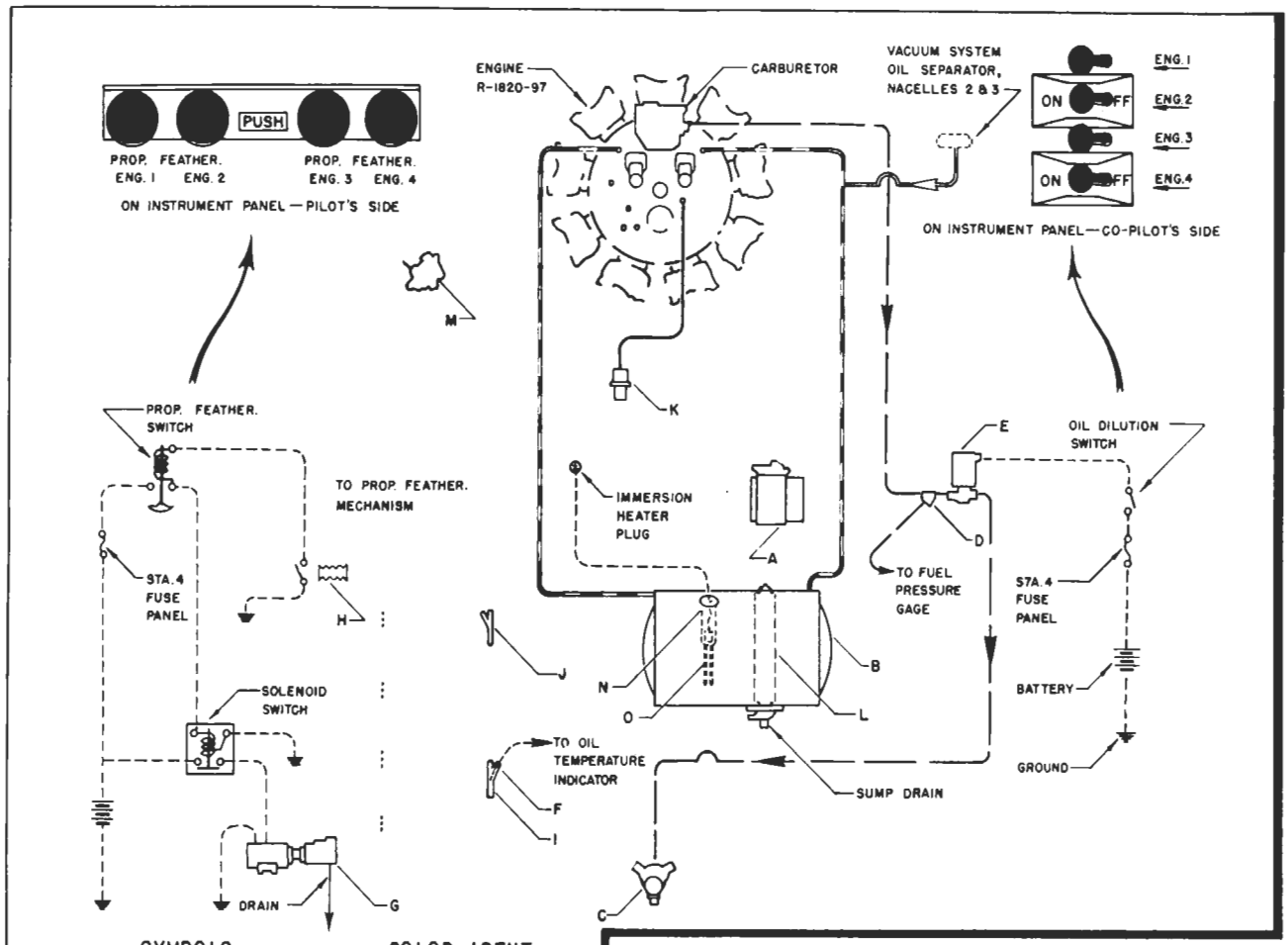


Figure 8 - Fuel Transfer Diagram

5. OIL SYSTEM



SYMBOLS	COLOR IDENT.
OIL LINES	YELLOW
VENT LINES	YELLOW & BLACK
OIL DILUTION LINES	RED
OIL DISCHARGE LINES	
OIL LINES: HIGH PRES.	YELLOW & BLACK
ELECTRICAL WIRING	

KEY TO FIGURE 9

A - OIL TEMPERATURE REGULATOR
B - OIL TANK
C - DRAIN VALVE
D - RESTRICTION FITTING
E - OIL DILUTION VALVE
F - OIL TEMPERATURE BULB
G - PROPELLER FEATHERING PUMP
H - PRESSURE CUTOUT SWITCH
I - TUBE ASSEMBLY
J - TUBE ASSEMBLY
K - OIL PRESSURE TRANSMITTER
L - OIL TEMPERATURE ACCELERATING WELL
M - SUPERCHARGER REGULATOR
N - HEATER RECEPTACLE ASSEMBLY
O - ELECTRICAL IMMERSION OIL HEATER

Figure 9 - Oil Flow Diagram

a. Each engine is equipped with a self-sealing oil tank having a capacity of 37 gallons plus approximately 10 percent expansion space. The total of 148 gallons for all four tanks is required for maximum fuel load with wing tanks and bomb bay tanks full. The propeller feathering pump receives its oil supply from the "in" line. Oil for operating the supercharger regulators is supplied from the engine oil system.

b. The oil temperature regulator is in the "out" line. Operation of oil-cooler shutters is fully automatic and, therefore, no oil-cooler controls are provided in the cockpit. Each tank is equipped with an oil heat accelerator or "hopper" which keeps the oil from circulating through the tank.

c. An oil dilution system is provided for diluting the oil with gasoline at the end of an engine run to provide easier starting.

d. Fill oil tanks with Specification No. AN-VV-O-446, grade 1120 for normal operations, grade 1100A for cold weather.

6. ELECTRICAL SYSTEM

a. A 24-volt d-c system distributes power from four engine-driven generators and from three storage batteries in the leading edges of the wing, just outboard of the fuselage. Three solenoid-operated battery switches are controlled by toggle switches on the pilot's control panel.

b. A gasoline engine-driven generator unit stowed in the rear fuselage compartment may be operated on the ground to provide auxiliary electric power for recharging batteries and for limited radio operation.

c. Alternating current for the Autosyn instruments, drift meter, radio compass, and warning signals transformer is furnished by two inverters under the pilot's and copilot's seats. A double-throw switch on the pilot's control panel selects the inverter to be used: in "NORMAL" position the left inverter is on; in "ALTERNATE" position the right inverter is on. Both inverters are off when the switch is centered.

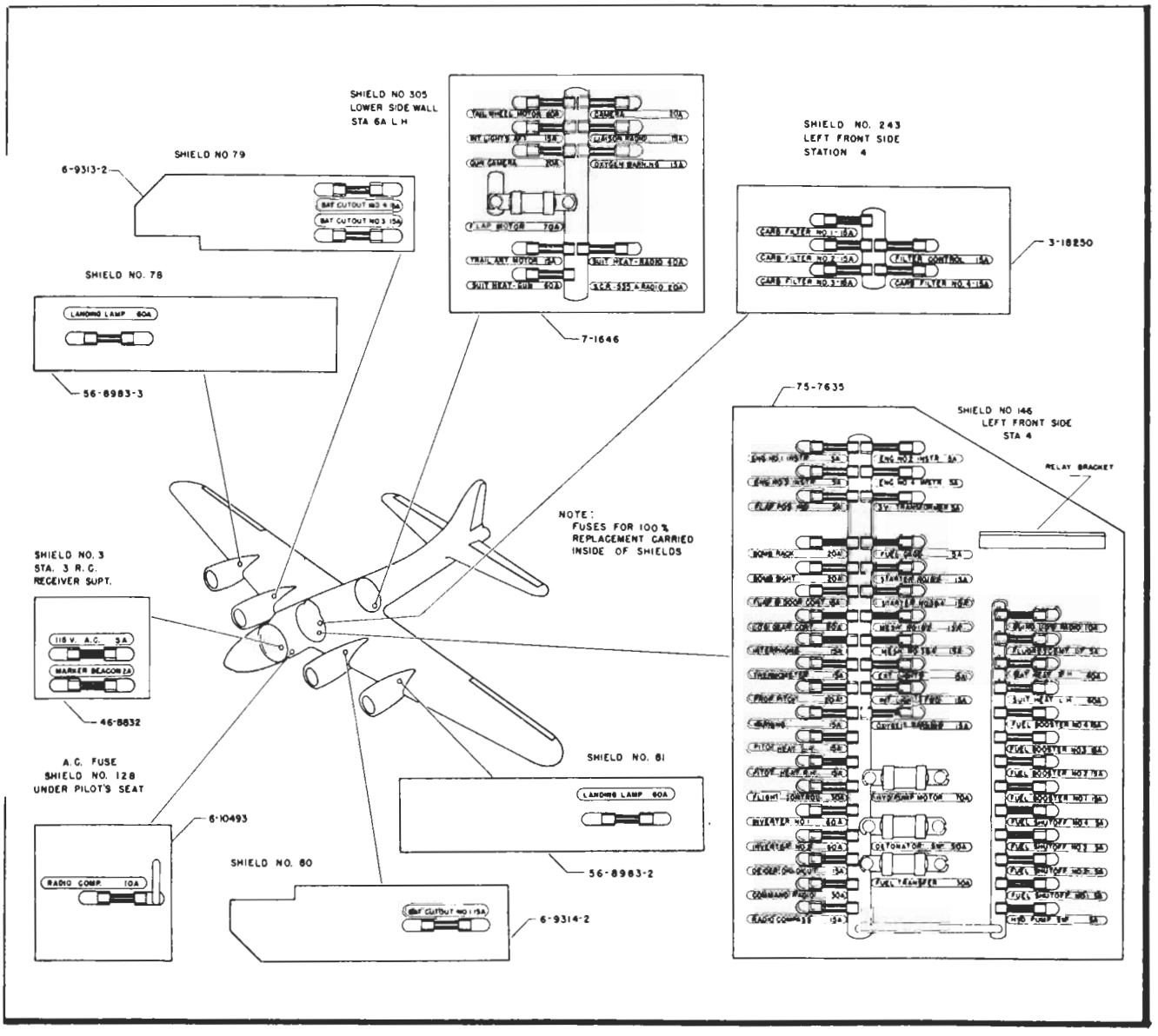


Figure 10 - Fuse Location Diagram

7. HEATING

a. GLYCOL HEATING SYSTEM. - Cabin heat is supplied by a hot air system in which heat is transferred to the ventilating air from a glycol system in the No. 2 nacelle. Flow of heated air to the cabin is controlled by a damper at the pilot's left. Defroster air is controlled by a red knob in the "v" of the pilot's windshield and by a control near the outlet in the bombardier's air duct. Fill glycol tank with approved mixture only; do not dilute with water.

CAUTION

During starting and ground operation of engines, the cabin heat control must be in the "OFF" or "COLD" position to prevent glycol in the system from boiling away.

b. AUXILIARY HEATING SYSTEM. - A similar glycol system, installed in the No. 3 nacelle of some airplanes, supplies eight radiator-fan heating and defrosting units in various locations in the airplane. Fan motors are thermostatically controlled and the flow of heating air is regulated by a damper at each unit.

c. SUIT HEATER OUTLET. - Ten receptacles for plugging in electric suit heaters are provided at various crew stations. The heat output of each suit is controlled by a rheostat on the receptacle box.



Figure 12 - Suit Heater Receptacle

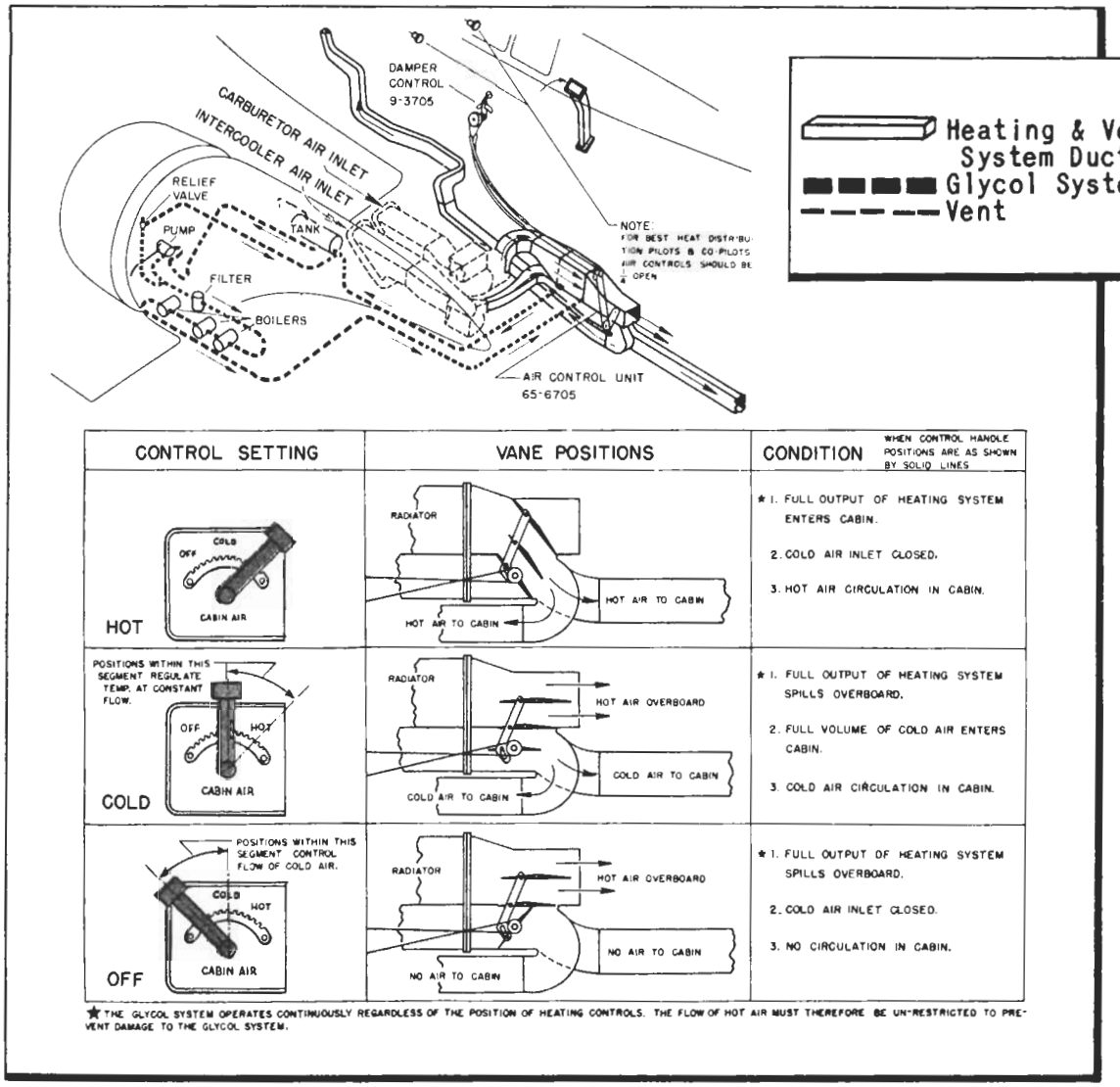


Figure 11 - Heating System Diagram

9. OXYGEN SYSTEM

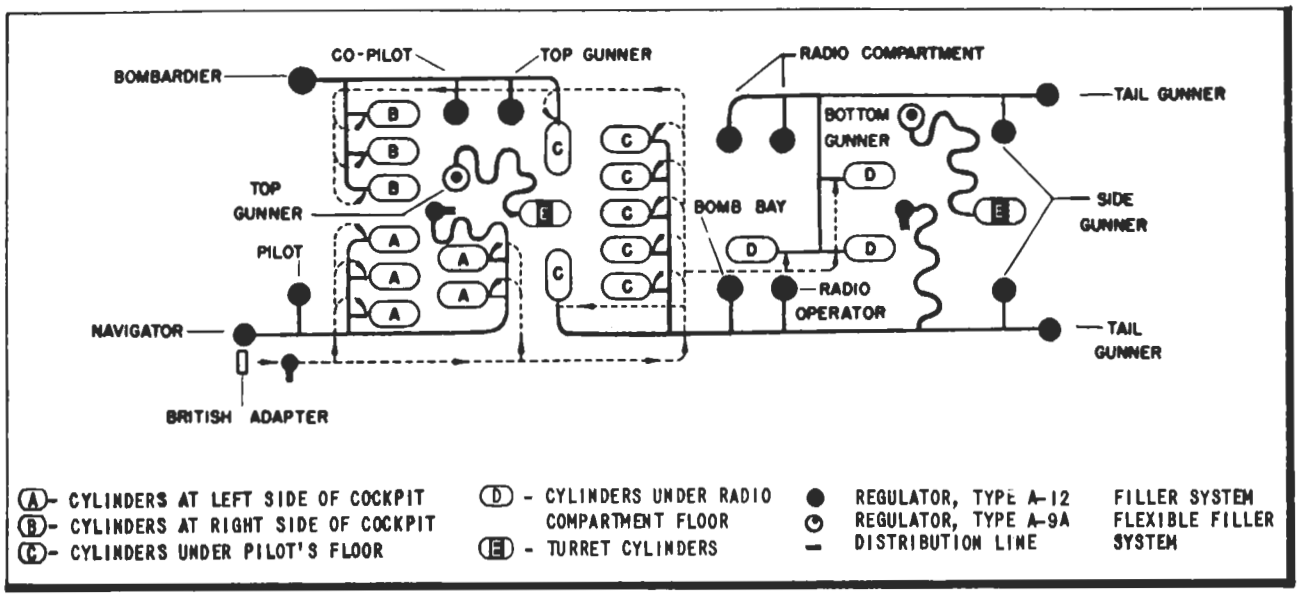


Figure 14 - Oxygen Flow Diagram

a. SUPPLY SYSTEM. - Breathing oxygen is stored in 18 type G-1 cylinders and is distributed by four self-contained systems, each serving two or more crew stations, which prevent complete loss of supply should a distribution line be severed. A check valve at each cylinder prevents loss of system pressure through a punctured cylinder. Each fully charged G-1 cylinder will supply one man with oxygen for 5 hours at 30,000 feet. The main system is filled to 400 pounds per square inch pressure through a filler valve just aft of the forward entrance hatch. On some airplanes a separate type F-1 cylinder at each power turret provides 2-1/2 hours of oxygen for one man at 30,000 feet and is refilled from the main system through a valve on a flexible hose. (See figure 15) Portable oxygen units provided for each crew member may be filled at the recharging valve at any demand regulator.

b. REGULATORS. - A type A-12 demand regulator and an indicator panel are located at each crew station. (See figure 16 for operation.) Power turrets are equipped with A-9A constant-flow regulators in airplanes having separate turret cylinders.

c. INDICATOR PANELS. - When oxygen flows from the regulator, the ball in the indicator bounces up in the glass tube; when flow stops, the ball falls. Do not be surprised if the indicator shows no oxygen flowing when the airplane is on the ground and the auto-mix is "ON," as the regulator is not necessarily supposed to add oxygen at ground level. The gage shows the pressure in the supply cylinders for that station. The warning signal lights when that pressure falls below 100 pounds per square inch.

NOTE

In some airplanes 15 constant-flow type A-9A regulators are provided. This installation has a relief valve in the filler system, and does not have the indicator panels or the portable units, but is essentially the same as the demand system.

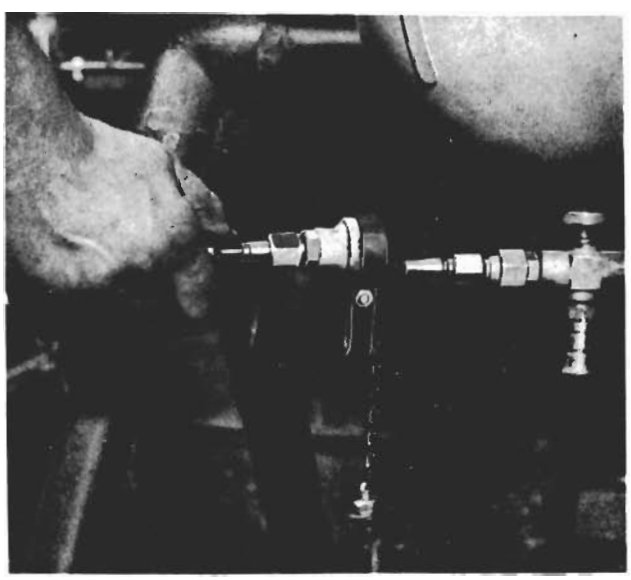


Figure 15 - Refilling Turret Oxygen Cylinder

USE OXYGEN INTELLIGENTLY



Figure 16 - Use of Oxygen

CAUTION

EXERCISE EXTREME CAUTION TO INSURE THAT OXYGEN EQUIPMENT DOES NOT BECOME CONTAMINATED WITH OIL OR GREASE. FIRE OR EXPLOSION MAY RESULT WHEN EVEN SLIGHT TRACES OF OIL OR GREASE COME IN CONTACT WITH OXYGEN UNDER PRESSURE.

1. Have your own mask which has been checked for fit by the oxygen officer.

2. Carry your bail-out cylinder charged to 1800 pounds.

3. Check to see that there is a portable "walk-around" unit at each station, filled to 400 pounds, and in working order.

4. Check system pressure before flight; it should be 400 pounds.

5. Check function of demand regulator in both "ON" and "OFF" positions. Flow gage should function when auto-mix is "OFF."

6. Check knurled collar on elbow connecting mask hose to regulator for tightness.

7. Open emergency valve to check flow; then close. This valve should not be open except in case of emergency.

8. Turn regulator to auto-mix "ON" position.

9. Use auto-mix "OFF" only -
   When oxygen officer advises the use of pure oxygen before take-off, in which case, use it all the way up as protection against "bends."

   When treating men for shock, loss of blood, or as protection against poisonous gas.

10. Start using oxygen at 10,000 feet. At night use oxygen from ground up, with auto-mix in "ON" position.

11. In flight above 10,000 feet, always use "walk-around" unit when moving from one station to another.



To use portable unit; first: Check pressure gage of portable unit; 2nd: Inhale deeply, then disconnect mask from regular hose and quickly open spring cover of regulator connection and snap in male fitting on end of mask hose. Clamp portable unit to clothing.



DON'T DELAY— When pointer on pressure gage reaches red area refill cylinder.

Figure 17 - Portable Oxygen Unit in Use

MAN HOURS OF AVAILABLE OXYGEN

BLACK FIGURES INDICATE AUTO-MIX "ON" RED FIGURES INDICATE AUTO-MIX "OFF"

CAUTION—The auto-mix in the off position rapidly diminishes the available oxygen supply. Do not use this position unless it is necessary to get *pure oxygen!*

AIRCO REGULATORS
TYPE A-12

PIONEER REGULATORS
TYPE A-12

GROUP I (5 G-1 Cylinders)
Pilot, Navigator and Top Turret Filler

Alt. Ft. \ Gage Pres.	400	350	300	250	200	150	100	50	
	41.5	35.6	29.4	23.6	17.8	12.0	5.8		
40,000	41.5	35.6	29.4	23.6	17.8	12.0	5.8		E
	29.5	25.3	20.9	16.8	12.6	8.5	4.0		
35,000	29.5	25.3	20.9	16.8	12.6	8.5	4.0		M
	21.5	18.5	15.2	12.2	9.2	6.0	3.0		
30,000	22.0	18.9	15.6	12.5	10.4	6.2	3.0		E
	16.5	14.1	11.5	9.0	7.0	4.7	2.0		
25,000	21.0	18.0	14.9	11.9	9.0	6.0	2.9		R
	13.0	11.1	9.2	7.4	5.5	3.7	1.5		
20,000′	23.5	20.2	16.6	13.3	10.1	6.8	3.2		G
	10.0	8.6	7.0	5.7	4.0	3.9	1.4		
15,000	28.5	24.5	20.2	16.2	12.2	8.2	3.9		E
	8.0	6.8	5.6	4.5	3.4	2.3	1.1		
10,000	48.5	41.7	34.4	27.6	20.8	14.0	6.7		N
	6.5	5.5	4.6	3.7	2.8	1.8	1.0		
5,000	—	—	—	—	—	—	—		C
	5.5	4.7	3.9	3.1	2.3	1.5	0.7		
S. L.	—	—	—	—	—	—	—		Y

Alt. Ft. \ Gage Pres.	400	350	300	250	200	150	100	50	
	41.5	35.6	29.4	23.6	17.8	12.0	5.8		
40,000	41.5	35.6	29.4	23.6	17.8	12.0	5.8		E
	29.5	25.3	20.9	16.8	12.6	8.5	4.0		
35,000	30.0	25.8	21.3	17.1	12.9	8.7	4.2		M
	21.5	18.5	15.2	12.2	9.2	6.0	3.0		
30,000	22.5	19.3	15.9	12.8	9.6	6.5	3.1		E
	16.5	14.1	11.5	9.0	7.0	4.7	2.0		
25,000	22.0	18.4	15.6	12.5	9.4	6.3	3.0		R
	13.0	11.1	9.2	7.4	5.5	3.7	1.5		
20,000	39.0	33.5	26.6	22.2	16.7	11.3	5.4		G
	10.0	8.6	7.0	5.7	4.0	3.9	1.4		
15,000	38.0	32.6	26.9	21.6	16.3	11.0	5.3		E
	8.0	6.8	5.6	4.5	3.4	2.3	1.1		
10,000	37.5	32.2	26.6	21.3	16.1	10.8	5.2		N
	6.5	5.5	4.6	3.7	2.8	1.8	1.0		
5,000	28.5	24.5	20.2	16.1	12.2	8.2	3.9		C
	5.5	4.7	3.9	2.3	2.3	1.5	0.7		
S. L.	30.0	25.8	21.3	17.1	12.9	8.7	4.2		Y

GROUP II (4 G-1 Cylinders)
Co-pilot, Bombardier and Top Gunner

Alt. Ft. \ Gage Pres.	400	350	300	250	200	150	100	50	
	33.2	28.6	23.6	19.0	14.2	9.6	4.6		
40,000	33.2	28.5	23.6	18.9	14.2	9.6	4.6		E
	23.6	20.2	16.8	13.4	10.2	6.8	3.4		
35,000	23.6	20.3	16.7	13.4	10.1	6.8	3.3		M
	17.2	14.8	12.2	9.8	7.4	5.0	2.4		
30,000	17.6	15.1	12.5	10.0	7.6	5.0	2.4		E
	13.2	11.2	9.2	7.4	5.6	3.8	1.8		
25,000	16.8	14.4	11.9	9.6	7.2	4.8	3.3		R
	10.4	9.0	7.4	6.0	4.4	3.0	1.4		
20,000	18.8	16.2	13.3	10.7	8.1	5.4	2.6		G
	8.0	6.8	5.6	4.6	3.4	2.4	1.2		
15,000	22.8	19.6	16.2	13.0	9.9	6.6	3.2		E
	6.4	5.4	4.6	3.6	2.8	1.8	0.8		
10,000	38.8	33.4	27.5	22.1	16.7	11.2	5.4		N
	5.2	4.4	3.6	3.0	2.2	1.4	0.8		
5,000	—	—	—	—	—	—	—		C
	4.4	3.8	3.2	2.4	1.8	1.2	0.6		
S. L.	—	—	—	—	—	—	—		Y

Alt. Ft. \ Gage Pres.	400	350	300	250	200	150	100	50	
	33.2	28.6	23.6	19.0	14.2	9.6	4.6		
40,000	33.2	28.5	23.6	18.9	14.2	9.6	4.6		E
	23.6	20.2	16.8	13.4	10.2	6.8	3.4		
35,000	24.0	20.6	19.0	13.7	10.3	6.9	3.3		M
	17.2	14.8	12.2	9.8	7.4	5.0	2.4		
30,000	18.0	15.5	12.8	10.2	7.7	5.2	2.5		E
	13.2	11.2	9.2	7.4	5.6	3.8	1.8		
25,000	17.6	14.7	12.5	10.0	7.6	7.1	2.4		R
	10.4	9.0	7.4	6.0	4.4	3.0	1.4		
20,000	31.2	26.8	22.1	17.8	13.4	9.0	4.3		G
	8.0	6.8	5.6	4.6	3.4	2.4	1.2		
15,000	30.4	26.1	21.6	17.3	13.0	8.8	4.2		E
	6.4	5.4	4.6	3.6	2.8	1.8	0.8		
10,000	30.0	25.9	21.3	17.1	12.9	8.7	4.2		N
	5.2	4.4	3.6	3.0	2.2	1.4	0.8		
5,000	22.8	19.6	16.2	13.0	9.8	6.6	3.1		C
	4.4	3.8	3.2	2.4	1.8	1.2	0.6		
S. L.	24.0	20.6	17.0	13.7	10.3	7.0	3.3		Y

MAN HOURS OF AVAILABLE OXYGEN

BLACK FIGURES INDICATE AUTO-MIX "ON" RED FIGURES INDICATE AUTO-MIX "OFF"

NOTE: Each turret cylinder, Type F-1, will supply one man for approximately 2 hours at 30,000 feet, 2½ hours at 25,000 feet, 3 hours at 20,000 feet.

AIRCO REGULATORS
TYPE A-12

PIONEER REGULATORS
TYPE A-12

GROUP III (6 G-1 Cylinders)
Bomb Bay, Radio Operator, Side Gunner, Tail Gunner, and Ball Turret Filler

Gage Pres. / Alt.. Ft.	400	350	300	250	200	150	100	50
	49.8	42.8	35.4	28.4	21.4	14.4	7.0	E
40,000	49.8	42.8	35.4	28.4	21.2	14.4	6.9	
	35.4	30.4	25.0	20.2	15.2	10.2	5.0	M
35,000	35.4	30.4	25.0	20.1	15.1	10.2	4.9	
	25.8	22.2	18.2	15.6	11.0	7.4	2.8	E
30,000	26.4	22.6	18.7	15.0	11.3	7.5	3.6	
	19.8	16.8	13.8	11.2	8.4	5.6	2.8	R
25,000	25.2	21.6	17.8	14.3	10.8	7.2	3.4	
	15.6	13.6	11.0	8.8	6.6	4.4	2.2	G
20,000	28.2	24.2	19.9	16.0	12.1	8.1	3.9	
	12.0	10.4	8.6	6.8	5.2	3.4	1.6	E
15,000	34.2	29.4	24.2	19.4	14.7	9.9	4.7	
	9.6	8.2	6.8	5.4	4.2	2.8	1.4	N
10,000	58.2	50.0	41.2	33.1	25.0	16.8	8.1	
	7.8	6.6	5.6	4.2	3.4	2.2	1.2	C
5,000	—	—	—	—	—	—	—	
	6.6	5.6	4.6	3.8	2.8	1.8	0.8	Y
S. L.	—	—	—	—	—	—	—	

Gage Pres. / Alt. Ft.	400	350	300	250	200	150	100	50
	49.8	42.8	35.4	28.4	21.4	14.4	7.0	E
40,000	49.8	42.8	35.4	28.4	21.3	14.4	6.9	
	35.4	30.4	25.0	20.2	15.2	10.2	5.0	M
35,000	36.0	30.9	25.5	20.5	15.4	10.4	5.0	
	25.8	22.2	18.2	15.6	11.0	7.4	2.8	E
30,000	27.0	23.2	19.1	15.3	11.5	7.8	3.7	
	19.8	16.8	13.8	11.2	8.4	5.6	2.8	R
25,000	26.4	22.0	18.7	15.0	11.3	7.6	3.8	
	15.6	13.6	11.0	8.8	6.6	4.4	2.2	G
20,000	46.8	40.2	33.1	26.6	20.1	13.5	6.5	
	12.0	10.4	8.6	6.8	5.2	3.4	1.6	E
15,000	45.6	39.1	31.7	25.9	19.5	13.2	6.3	
	9.6	8.2	6.8	5.4	4.2	2.8	1.4	N
10,000	45.0	38.7	31.9	25.6	19.3	13.0	6.3	
	7.8	6.6	5.6	4.2	3.4	2.2	1.2	C
5,000	32.2	29.4	24.2	19.4	14.7	9.9	4.5	
	6.6	5.6	4.6	3.8	2.8	1.8	0.8	Y
S. L.	36.0	31.9	25.5	20.5	15.4	10.4	5.0	

GROUP IV (3 G-1 Cylinders)
Radio Compartment (2 Outlets), Side Gunner and Tail Gunner

Gage Pres. / Alt. Ft.	400	350	300	250	200	150	100	50
	24.9	21.4	17.7	14.2	10.7	7.2	3.5	E
40,000	24.9	21.4	17.7	14.2	10.7	7.2	3.5	
	17.7	15.2	12.5	10.1	7.6	5.1	2.5	M
35,000	17.7	15.2	12.5	10.1	7.6	5.1	2.5	
	12.9	11.1	9.1	7.3	5.5	3.7	1.4	E
30,000	13.2	11.3	9.4	7.5	5.7	3.8	1.8	
	9.9	8.4	6.9	5.6	4.2	2.8	1.4	R
25,000	12.6	10.8	8.9	7.2	5.4	3.6	1.7	
	7.8	6.8	5.5	4.4	3.3	2.2	1.1	G
20,000	14.1	12.1	10.0	8.0	6.1	4.1	1.9	
	6.0	5.2	4.3	3.4	2.6	1.7	0.8	E
15,000	17.1	14.7	12.1	9.7	7.3	4.9	2.4	
	4.8	4.1	3.4	2.7	2.1	1.4	0.7	N
10,000	29.1	25.0	20.5	16.6	12.3	8.4	4.0	
	3.9	3.3	2.8	2.1	1.7	1.1	0.6	C
5,000	—	—	—	—	—	—	—	
	3.3	2.8	2.3	1.9	1.4	0.9	0.4	Y
S. L.	—	—	—	—	—	—	—	

Gage Pres. / Alt. Ft.	400	350	300	250	200	150	100	50
	24.9	21.4	17.7	14.2	10.7	7.2	3.5	E
40,000	24.9	21.4	17.7	14.2	10.7	7.2	3.5	
	17.7	15.2	12.5	10.1	7.6	5.1	2.5	M
35,000	18.0	15.5	12.8	10.3	7.7	5.2	2.5	
	12.9	11.1	9.1	7.3	5.5	3.7	1.8	E
30,000	13.5	11.6	9.6	7.7	5.8	3.9	1.9	
	9.9	8.4	6.9	5.6	4.2	2.8	1.4	R
25,000	13.2	11.0	9.4	7.5	5.7	3.8	1.8	
	7.8	6.8	5.4	4.4	3.3	2.2	1.1	G
20,000	23.4	20.1	16.6	13.3	10.0	6.8	3.3	
	6.0	5.2	4.3	3.4	2.6	1.7	0.8	E
15,000	22.8	19.6	16.2	13.0	9.8	6.6	3.2	
	4.8	4.1	3.4	2.7	2.1	1.4	0.7	N
10,000	22.5	19.3	16.0	12.8	9.7	6.5	3.1	
	3.9	3.3	2.8	2.1	1.7	1.1	0.6	C
5,000	16.1	14.7	12.1	9.7	7.3	4.9	2.3	
	3.3	2.8	2.3	1.9	1.4	0.9	0.4	Y
S. L.	18.0	15.5	12.8	10.3	7.7	5.2	2.5	

10. COMMUNICATIONS EQUIPMENT

a. GENERAL. - A radio and interphone system provides for communications between crew members within the airplane; between the airplane and ground stations or other airplanes; reception of weather, range, and marker beacon signals; and ground and interphone identification.

b. INTERPHONE SYSTEM. - Interphone jack boxes are installed at 11 locations in the airplane. With any selector switch in "CALL" position, that station may be heard at all other stations regardless of the position of their selector switches. With all switches adjusted to "INTER," any station may be heard at all other stations. Any station may listen to the liaison, command, or radio compass receiver by adjusting the selector switch to those positions. Any station can modulate the command radio transmitter; however, modulation of the liaison transmitter is provided for pilot, copilot, navigator, and radio operator. All stations are provided with throat microphones, which, with the exception of those for the pilot and copilot, are controlled by "PUSH-TO-TALK" switches on the cords. They are connected to the jack boxes by extension cords.

c. OTHER COMMUNICATIONS EQUIPMENT. Instruction for operating other communication equipment will be found in the section covering the compartment in which the equipment is located.



Figure 18
Interphone
Jack Box

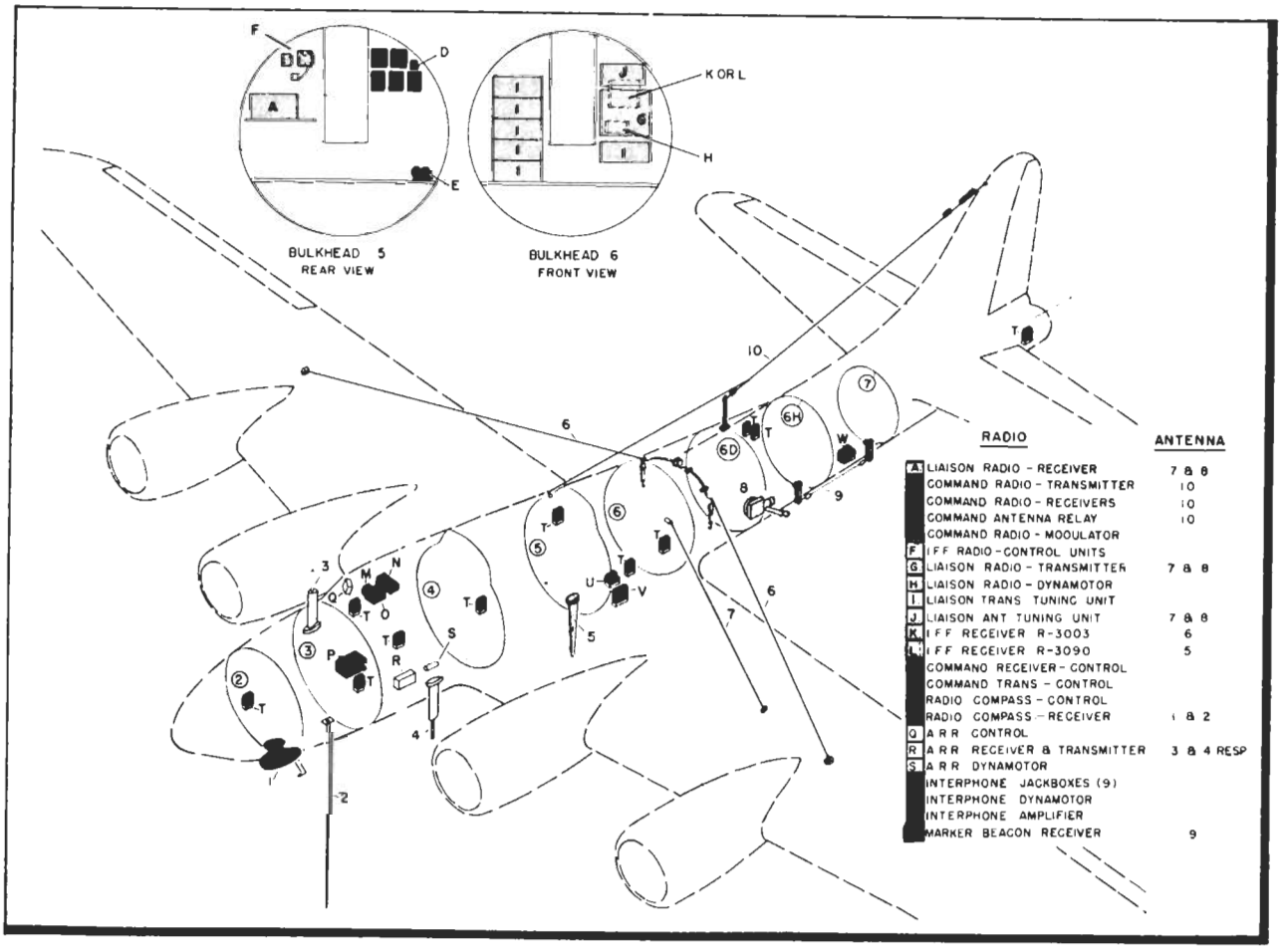
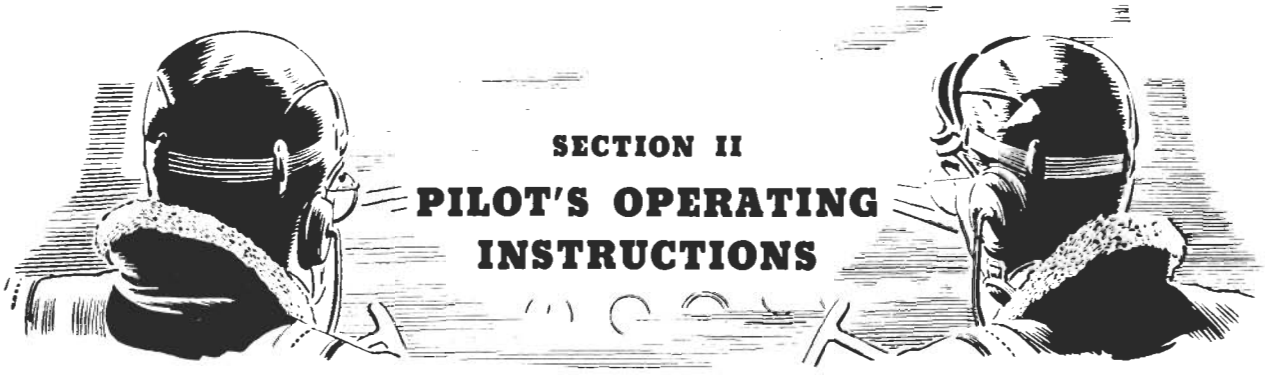


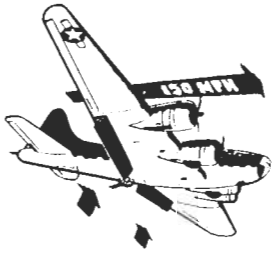
Figure 19 - Communications Equipment



SECTION II

PILOT'S OPERATING INSTRUCTIONS

1. RESTRICTIONS



DON'T lower flaps at speeds in excess of 147 mph!



DON'T dive in excess of 270 mph (with modified elevators).

WARNING

Some airplanes are restricted to 220-mph maximum diving speed, pending modification of the elevators. See warning placard in airplane.



DON'T exceed 46 inches Hg manifold pressure!



DON'T exceed 30 inches Hg below 2100 rpm!



DON'T stall the airplane! (except for training purposes.)



DON'T spin!



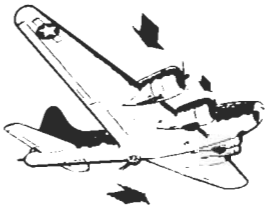
DON'T roll!



DON'T loop!



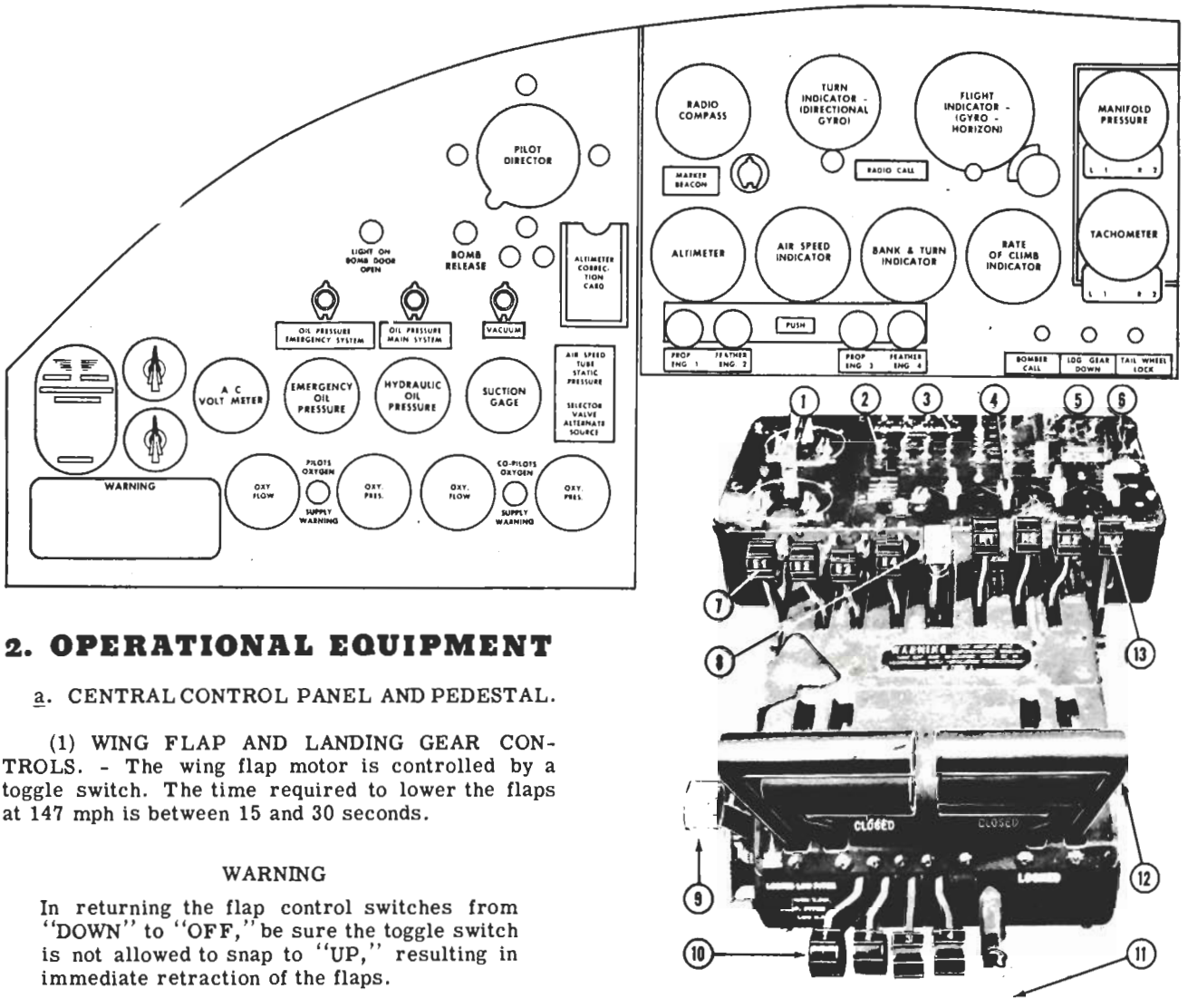
DON'T attempt inverted flight!



DON'T fly the airplane at maximum gross weight (64,500 pounds) UNLESS auxiliary wing tanks are full!

CAUTION

All power settings given in this section are for use with 100 octane fuel only. See appendix III for restrictions to be observed when using 91 octane fuel.



2. OPERATIONAL EQUIPMENT

a. CENTRAL CONTROL PANEL AND PEDESTAL.

(1) WING FLAP AND LANDING GEAR CONTROLS. - The wing flap motor is controlled by a toggle switch. The time required to lower the flaps at 147 mph is between 15 and 30 seconds.

WARNING

In returning the flap control switches from "DOWN" to "OFF," be sure the toggle switch is not allowed to snap to "UP," resulting in immediate retraction of the flaps.

(2) The main landing wheels and tail wheel are operated simultaneously by a toggle switch. A hinged guard prevents accidental moving of the switch to the "UP" position. Warning that the landing gear is not fully extended is given by a green indicator lamp failing to light, and by a horn which sounds if any throttle is closed.

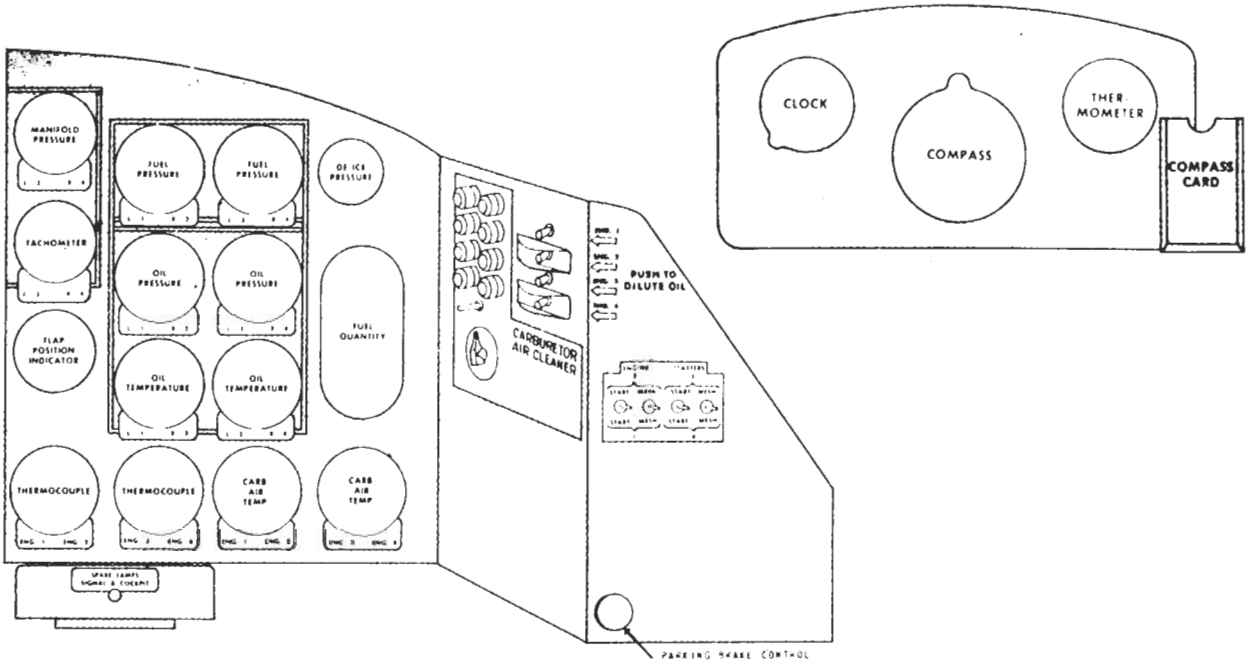
(3) COWL FLAP VALVES. - Cowl flaps are operated by four valves, each valve controlling the flaps on one nacelle. The valve must be turned to "LOCKED" when the desired position of the flaps is reached. Slight "cracking" of the control valve will result in relatively slow travel of the flaps when close adjustment is desired.

(4) FUEL BOOST CONTROLS. - The fuel boost pumps, operated by four toggle switches, provide fuel pressure for starting engines and for maximum power, and also prevent vaporization in the lines to engine-driven pumps due to hot fuel or high altitudes. Booster pressure at the No. 3 nacelle fuel strainer also supplies fuel to the priming system.

KEY TO FIGURE 21

1. IGNITION SWITCHES
2. FUEL BOOST PUMP SWITCHES
3. FUEL SHUT-OFF VALVE SWITCHES
4. COWL FLAP CONTROL VALVES
5. LANDING GEAR SWITCH
6. WING FLAP SWITCH
7. TURBO SUPERCHARGER CONTROLS
8. TURRO AND MIXTURE CONTROL LOCK
9. THROTTLE CONTROL LOCK
10. PROPELLER PITCH CONTROLS
11. PROPELLER PITCH CONTROL LOCK
12. THROTTLE CONTROLS
13. MIXTURE CONTROLS

Figure 21 - Control Panel and Pedestal



(5) FUEL SHUT-OFF VALVE SWITCHES. - Solenoid valves, operated by four toggle switches permit immediate shut-off of the fuel at the tank when necessary. Failure of electrical power causes the valves to "OPEN" allowing fuel to flow.

(6) IDENTIFICATION LIGHTS. - Two switches and a keying button permit signalling with any combination of the four lights.

(7) PROPELLER FEATHERING SWITCHES.

(a) Each propeller is feathered individually by one of the four red push button switches above the central control panel on the instrument panel. Pushing the switch in starts an electric pump in the nacelle which supplies hydraulic power for the feathering operation. When the propeller is fully feathered the push button automatically releases, stopping the pump. To stop the operation before feathering is complete, pull out the switch button by hand.

(b) To unfeather a propeller, the push-button switch must be manually held in the closed position until unfeathering has been accomplished.

NOTE

When unfeathering a propeller on a cold engine, do not allow the engine speed to exceed minimum governing speed until oil pressure and oil temperature appear satisfactory. Turn off the ignition after feathering any propeller if the engine is to remain inoperative for any length of time. Do not operate more than one propeller feathering switch at a time, except in emergencies.

(8) TURBOSUPERCHARGER CONTROLS. - The supercharger regulators are operated by engine oil pressure. With warm oil in the engine the minimum time for operating the regulator control from the low boost to the high boost position should be 5 seconds. If the oil is somewhat cooler than normal engine temperatures, this should be extended to 15 seconds.

b. COPILOT'S AUXILIARY PANEL.

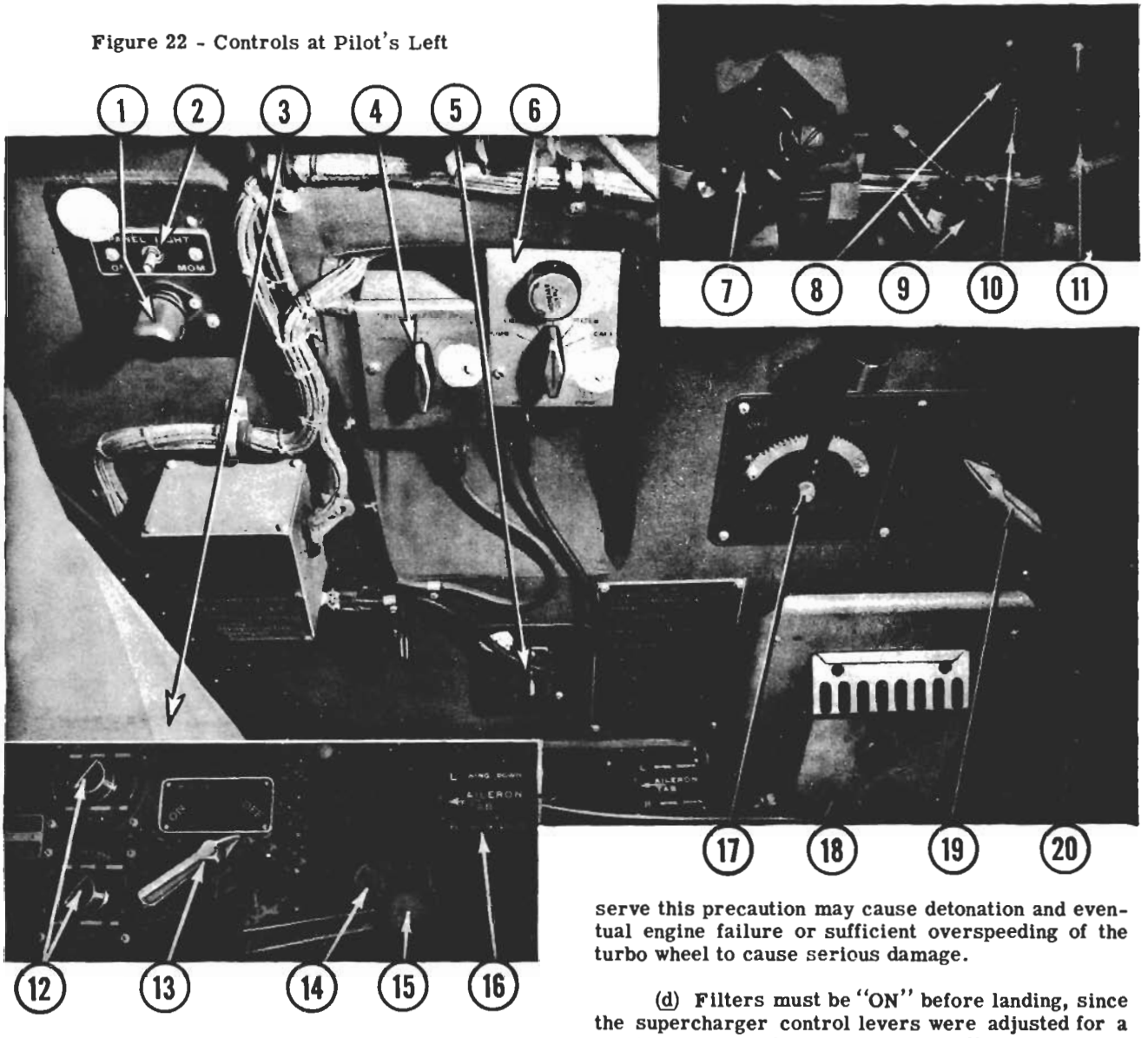
(1) CARBURETOR AIR FILTER CONTROLS.

(a) Carburetor air filter valve motors are controlled by one double-throw toggle switch located on the side of the auxiliary panel, forward of the copilot. When all the valves are "ON" permitting only filtered air to enter the supercharger intakes, four amber lamps are lighted. Four green lamps light when the control valves are "OFF," admitting only unfiltered air to the supercharger intakes. Any lamp failing to light indicates that the corresponding valve has not completed its travel to the full open or full closed position.

(b) Air filters should be "ON" for all ground operations and for dust conditions up to 8000 feet.

(c) Use of the filters above 8000 feet should be avoided, since operation above that altitude is accompanied by a rise in carburetor air inlet temperature, increasing the possibility of detonation. (This condition is aggravated by abnormally high outside air temperatures.) The turbo also has a tendency to overspeed. IN ALL CASES, THE FILTERS MUST BE CLOSED ABOVE 15,000 FEET! Failure to ob-

Figure 22 - Controls at Pilot's Left



KEY TO FIGURE 22

1. PANEL LIGHT
2. PANEL LIGHT SWITCH
3. PILOT'S SEAT
4. FILTER SELECTOR SWITCH
5. PROPELLER ANTI-ICER SWITCH
6. INTERPHONE JACKBOX
7. OXYGEN REGULATOR
8. WINDSHIELD WIPER CONTROLS
9. PORTABLE OXYGEN UNIT RECHARGER
10. WINDSHIELD ANTI-ICER SWITCH
11. WINDSHIELD ANTI-ICER FLOW CONTROL
12. PROPELLER ANTI-ICER RHEOSTATS
13. SURFACE DE-ICER CONTROL
14. AILERON TRIM TAB CONTROL
15. PILOT'S SEAT ADJUSTMENT LEVER
16. AILERON TRIM TAB INDICATOR
17. CABIN AIR CONTROL
18. SUIT HEATER OUTLET
19. VACUUM SELECTOR VALVE
20. EMERGENCY BOMB RELEASE

serve this precaution may cause detonation and eventual engine failure or sufficient overspeeding of the turbo wheel to cause serious damage.

(d) Filters must be "ON" before landing, since the supercharger control levers were adjusted for a maximum manifold pressure at take-off with the filters "ON." If emergency power is attempted with the filters "OFF," manifold pressures above the recommended maximum of 46 inches will be obtained.

(2) OIL DILUTION SWITCHES.

(a) Four momentary contact toggle switches on the side of the copilot's auxiliary panel operate solenoid valves in the corresponding nacelle, admitting fuel to the engine oil in line. This operation is performed AFTER an engine run, immediately prior to shutting it off.

(b) Do not dilute oil over 4 minutes. The supercharger controls should be operated continuously during this period to cause diluted oil to flow to the regulators. The propeller control should be moved

from extreme increase to extreme decrease rpm slowly several times to fill the propeller dome with diluted oil and prevent sluggish response of the propeller when starting the engine.

(3) STARTER SWITCHES. - Two START and two MESH switches control the engine starters. The START switch energizes the starter motor, rotating the inertia flywheel. The MESH switch engages the starter and engine jaws while the START switch is held on.

NOTE

Some airplanes have a "START-OFF-MESH" switch for each engine starter.

(4) PARKING BRAKE. - The pull handle at the bottom of the instrument panel sets the copilot's brake metering valves when the foot pedals are depressed. This utilizes the regular braking system; therefore, hydraulic system pressure must be available when the parking brake is required for any length of time. When necessary, set the parking brake handle and pump the system pressure to at least 400 pounds per square inch (minimum pressure for full braking control).

WARNING

Do not set parking brake while brake drums are hot.

(5) FUEL INDICATOR. - A liquidometer indicator, on the extreme right side of the instrument panel, shows the available fuel supply in any one of the six main fuel tanks. A six-position switch directly below the indicating dial, selects the tank to be checked.

(6) INSTRUMENT LIGHTING.

(a) Three spot lamps light the instrument panel and a fourth on the ceiling lights the compass panel. Two types of light are available: for flood lighting with visible fluorescent light, rotate the shutter to the left; for ultra-violet activation of the luminous paint on the instrument dials, rotate the shutter in the opposite direction approximately one-quarter turn.

(b) The spot lights are controlled by switches, two on the pilot's instrument panel, and one on the copilot's auxiliary panel. To operate, hold the switch in the "START" position for approximately 2 seconds; then, release the switch allowing it to spring back to the "ON" position.

c. CONTROLS AT PILOT'S LEFT.

(1) CABIN AIR CONTROL. - Heat and ventilation are controlled by a lever on the side wall. (See figure 11 for operation.)

CAUTION

Be sure the heater control is "OFF" or "COLD" for all starting and ground operations.

(2) VACUUM PUMP CONTROL. - The "GYRO INSTRUMENTS" selector valve on the side wall permits use of either vacuum pump for the gyro instruments, suction from the other pump being connected to the surface de-icer system. (See figure 13.)

(3) DE-ICER CONTROL. - The de-icer valve on the floor panel controls the operation of the surface de-icer shoes. In the "ON" position it starts the de-icer distributor and connects the exhaust pressure from both vacuum pumps, and the suction from one vacuum pump to the distributor valve. In the "OFF" position the distributor motor is turned off and the pressure from the vacuum pumps is bypassed overboard. Suction remains connected to the distributor valve in order to keep the de-icer shoes deflated.

(4) PROPELLER ANTI-ICER CONTROL. - A toggle switch on the side wall controls the two propeller anti-icer pumps. Two rheostats on the floor panel control the speed of the pump motors and may be used to turn the motors off if desired. Normally the rheostats should be left adjusted to a predetermined rate of flow and the pump motors turned on or off by means of the toggle switch.

(5) WINDSHIELD WIPER AND ANTI-ICER. - Windshield wiper and anti-icer controls are on a panel at the pilot's left.

(a) A toggle switch controls the operation of the wiper motor, "OFF," "SLOW," or "FAST," and a circuit breaker is provided to protect motor in case of an overload.

(b) An "ON-OFF" switch controls the alcohol pump, and flow is regulated by a needle valve.

CAUTION

Do not operate wipers on dry glass!

(6) EMERGENCY BOMB RELEASE. - An emergency bomb release handle is at the pilot's left. Pulling the handle immediately releases bomb door latches, and continued pulling will release all bombs SALVO the instant the doors are fully open. Bomb bay fuel tanks may be dropped by the release handle.

d. PILOT'S CONTROL PANEL.

(1) ALARM BELL CONTROL. - A toggle switch operates three alarm bells: one under the navigator's table, one above the radio operator's table, and one in the tail wheel compartment inside the dorsal fin.

(2) PHONE CALL. - Another toggle switch operates four amber phone call signal lamps: three ad-

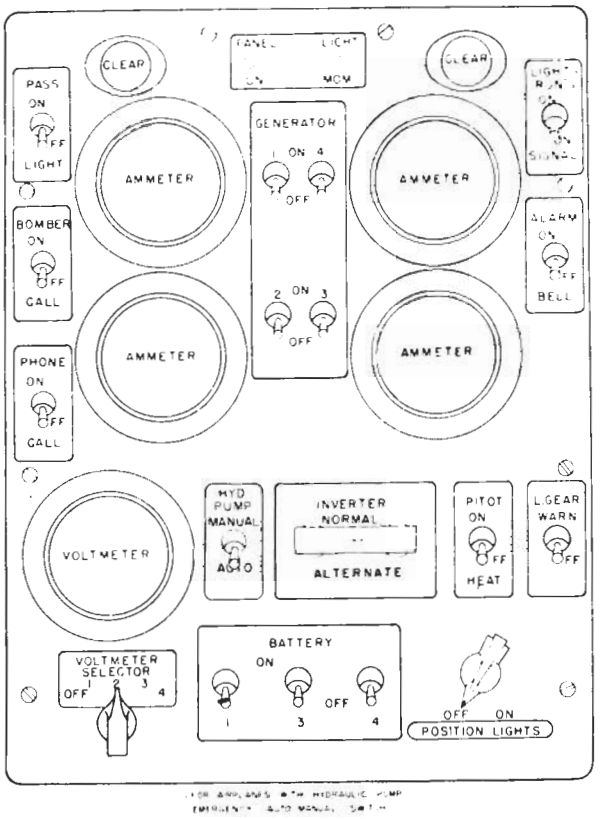


Figure 23 - Pilot's Control Panel

jacent to the alarm bells, and the fourth at the tail gunner's right.

(3) BOMBARDIER CALL. - A toggle switch on the pilot's control panel operates an amber call lamp on the bombardier's control panel; and a toggle switch on the bombardier's panel operates an amber call lamp on the pilot's instrument panel.

(4) LANDING GEAR WARNING HORN RESET. - A switch on the control panel permits the silencing of the landing gear warning horn when it is desired to continue flight with one or more throttles closed. Operation of this switch does not prevent repetition of the warning for subsequent closing of any throttle while the landing gear is up. The switch is reset when the throttles are opened.

(5) INVERTER SWITCH. - A double-throw switch selects which of two inverters is to be used: in "NORMAL" position the left inverter is on; in "ALTERNATE" position the right inverter is on.

(6) HYDRAULIC PUMP SWITCH. - With this switch in the "AUTO" position, pressure is automatically regulated between 600 and 800 pounds. In case of failure of the automatic pressure, cut-out pressure may be maintained by holding the switch in the "MANUAL" position.

WARNING

In case of leakage stop the pump to prevent loss of fluid. Remove switch fuse at station 4 fuse panel or disconnect receptable at switch. In some airplanes the hydraulic pump is controlled by an "ON-OFF" switch.

(7) CARBURETOR ANTI-ICER.

(a) Carburetor icing may occur in outside air temperatures up to 50°F (10°C), with humidity greater than 50 percent. Ice formation in the carburetor adaptor or at the fuel nozzle, indicated by engine roughness and a drop in manifold pressure, may be eliminated by moving the intercooler shutters to "HOT," or by setting the turbos "FULL ON" and adjusting power with the throttles. Apply full power and climb above icing condition if possible. Below 15,000 feet the air filters may be opened to provide a further increase of carburetor air temperature.

WARNING

DO NOT EXCEED ALLOWABLE LIMITS FOR MANIFOLD PRESSURE, ENGINE RPM, AND CYLINDER HEAD TEMPERATURE.

(b) Some airplanes are equipped with carburetor anti-icers consisting of pumps controlled by toggle switches on the pilot's control panel. One supplies inboard engines; the other, outboard engines. Approximately 4 gallons of isopropyl alcohol per hour are sprayed into the pressure duct of each carburetor, the entire system sustaining a total of 2 hours operation. This equipment should be used as follows:

1. To start an engine after severe carburetor icing or engine stoppage.

2. To determine cause of power loss or engine roughness; if adjustment of engine controls and use of

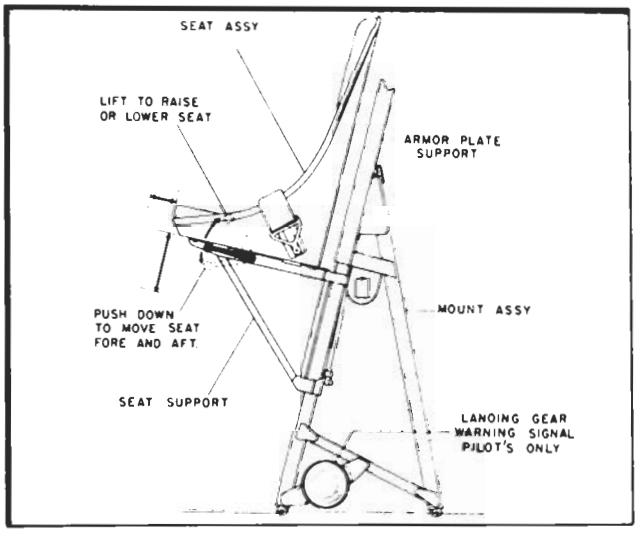


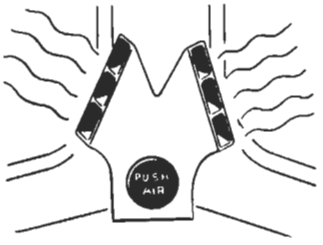
Figure 24 - Pilot's Seat Adjustment

alcohol system does not relieve condition, it can be assumed the trouble is not caused by icing.

3. To clear out engines quickly after a glide at low power through icing conditions.

4. To obtain full power under icing conditions.

5. As an alternate method of ice elimination if use of fuel turbo or carburetor air filter is prohibited.



e. DEFROSTER CONTROL. - Hot air for defrosting the pilot's and copilot's windshields is controlled by a red button in the vee of the windshield.

f. TRIM TAB CONTROLS.

(1) Complete aileron tab travel requires about 3-3/4 turns of the knob located on the pilot's floor panel.

(2) Complete rudder tab travel requires about seven turns of the wheel located on the floor in front of the control pedestal.

(3) The elevator trim tab wheel on the left side of the control pedestal requires about six turns for complete travel. It has a friction brake to prevent creeping.

g. LOCKS.

(1) AILERON LOCK. - The aileron is locked in neutral position by a pin which is manually inserted in a hole in the left control column, holding the center spoke of that wheel in a padded slot. The pin is clipped to the pilot's control column when not in use.

(2) RUDDER AND ELEVATOR LOCK. - The rudder and elevator locking lever operates by cable control to place a pin in a socket on a segment at each of the control quadrants. The locking lever, which is recessed into the floor aft of the engine control pedestal, is locked in either the "UP" or "DOWN" position. The lever may be moved to the "UP" or "LOCKED" position, regardless of the attitude of the control surfaces. Under this condition, the control surfaces will automatically lock when the rudder is in the "NEUTRAL" position and the elevator is in the "DOWN" position.

(3) TAIL WHEEL LOCK. - The tail wheel locking lever operates a single cable to retrace a spring-loaded locking pin from a socket in the treadle. The locking lever which is recessed into the floor aft of the control pedestal, latches in the "UP" position only and may be moved into the "DOWN" position regardless of the attitude of the tail wheel, which will lock when centered. To release the locking handle, press the knob on the end of it. A red signal light on the pilot's instrument panel is "OFF" when the tail wheel is locked.

h. AUTOMATIC FLIGHT CONTROL EQUIPMENT. The automatic flight control panel is located on the front of the control pedestal. To engage A.F.C.E.:

(1) Throw "ON" master and stabilizer switches.

(2) CAREFULLY TRIM AIRPLANE FOR STRAIGHT AND LEVEL FLIGHT.

(3) Turn "ON" tell-tale lights.

(4) After master and stabilizer switches have been "ON" for 10 minutes, throw "ON" PDI and servo switches.

(5) Center PDI by turning plane and resuming straight and level flight.

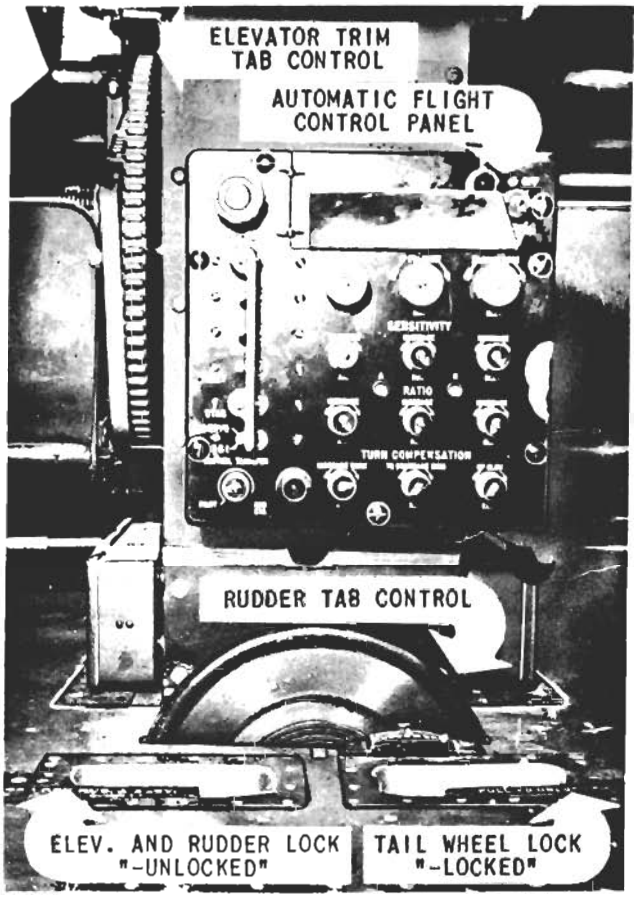


Figure 25 - Lower Control Pedestal

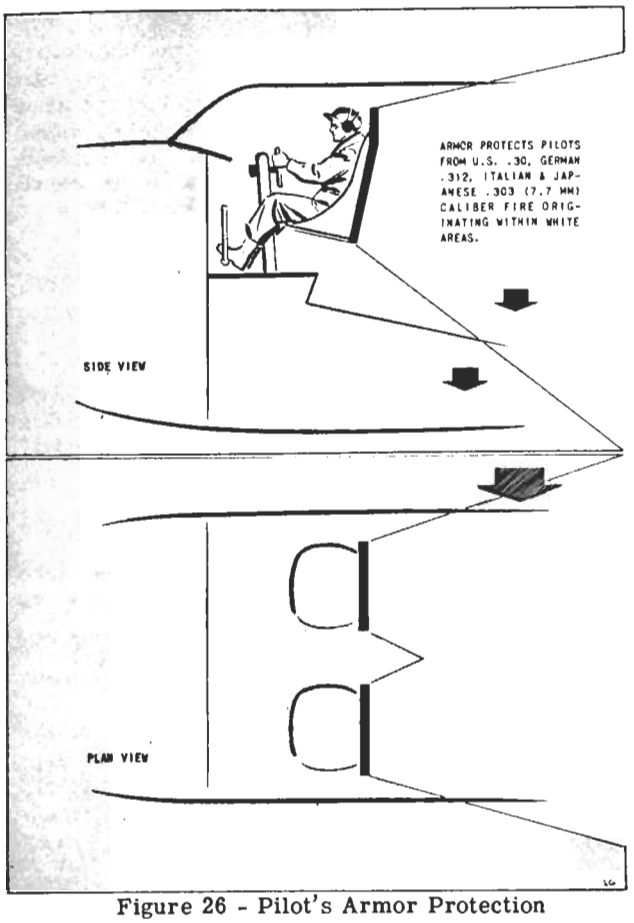


Figure 26 - Pilot's Armor Protection

(6) With PDI on "ZERO," adjust rudder centering knob until both rudder tell-tale lights go "OUT." Immediately throw rudder switch "ON."

(7) With wings level, adjust aileron centering knob until both aileron tell-tale lights go "OUT." Immediately throw aileron switch "ON."

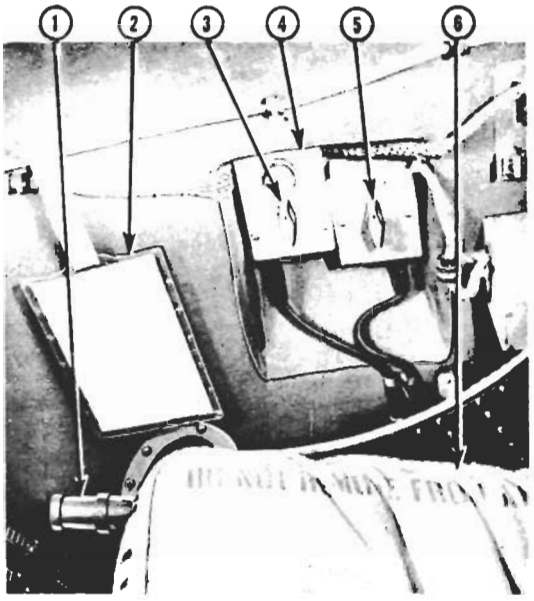
(8) With airplane flying level, adjust elevator centering knob until both elevator tell-tale lights go "OUT." Immediately throw elevator switch "ON."

(9) Observe PDI, artificial horizon, and rate-of-climb or altimeter instruments. Then carefully retrim all centering knobs, until ship is flying as straight and level as possible, with PDI on "CENTER."

(10) With autopilot engaged, all course corrections must be made with turn control ONLY. Always turn knob with a slow steady movement.

WARNING

Do not engage A.F.C.E. motors until all "tell-tale" lights are off.



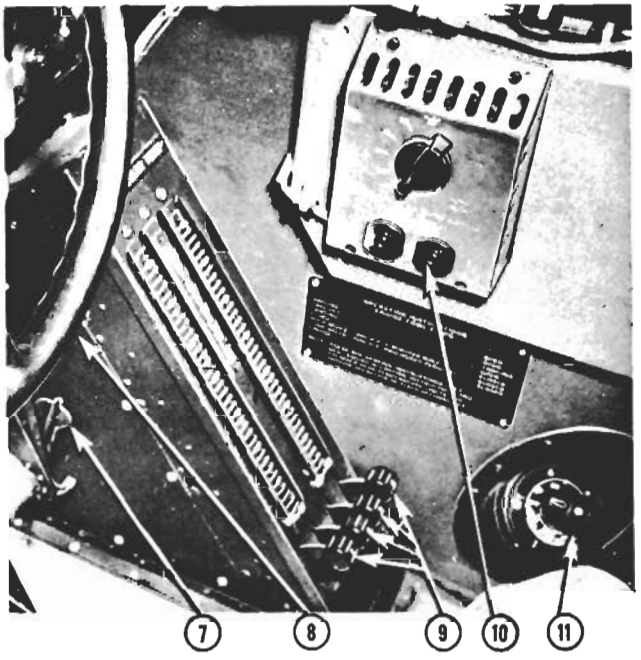


Figure 27 - Controls at Copilot's Right

KEY TO FIGURE 27

1. HYDRAULIC HAND PUMP
2. CHECK LIST
3. INTERPHONE SELECTOR SWITCH
4. INTERPHONE JACKBOX
5. FILTER SELECTOR SWITCH
6. COPILOT'S SEAT
7. RUDDER PEDAL ADJUSTMENT
8. COPILOT'S CONTROL WHEEL
9. INTERCOOLER CONTROLS
10. SUIT HEATER OUTLET
11. ENGINE PRIMER

i. CONTROLS AT COPILOT'S RIGHT.

(1) PRIMER. - The cylinder head primer has four positions corresponding to the four engines, and an "OFF" position. The primer handle is locked only in the "OFF" position. To operate, push the handle down, turn the valve to the engine position required, and then withdraw the handle and pump the charge to the cylinder.

IMPORTANT

Overpriming will result if the handle is left in the withdrawn position. Therefore, each priming operation must terminate with the handle returned to the locked position.

(2) CARBURETOR TEMPERATURE CONTROLS. The intercooler shutters are controlled from a stand in front of the copilot. Each cable is operated by a slide latching in any desired position. To release the latch, pull handle out.

(3) HYDRAULIC HAND PUMP. - The hydraulic hand pump is manually operated to furnish pressure in case of failure of the electric pump.

(4) KEY CASE. - A key case on the side wall contains two keys which fit all door locks in the airplane.

j. RUDDER PEDAL ADJUSTMENT. - Rudder pedal tilt may be varied to any of five positions by a locking pin and sector at the outside corner of each pedal.

k. PILOT'S COMMUNICATIONS CONTROLS.

(1) GENERAL.

(a) All communications equipment may be operated to some extent from the pilot's compartment. Receiver and transmitter frequency selection may be controlled with the exception of the liaison equipment which must have both its transmitter and receiver frequencies set by the radio operator.

CAUTION

For normal operation of all communications equipment, the filter selector switch should be set at "BOTH." To receive the radio range without possibility of voice interference, set the selector switch to "RANGE." To receive voice without range interference, set selector switch to "VOICE."

NOTE

The head set extension cord should be plugged into the filter selector control box as shown in figure 28 and not into the interphone jackbox or the receiver control box.

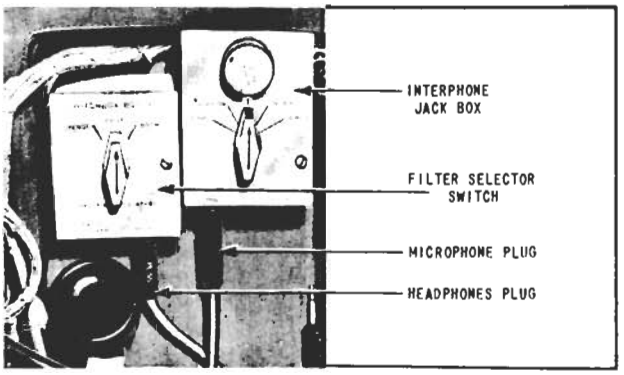


Figure 28 - Microphone and Headset Plugs

IMPORTANT

When the throat microphone is being used for either interphone or radio communication, it must be adjusted so that its two circular elements are held snugly against each side of the throat just above the "Adam's apple." SPEAK SLOWLY, DISTINCTLY, AND IN A NORMAL TONE OF VOICE. Shouting will seriously distort the voice signal.

(b) A possible means of limiting noise level in all radio equipment, caused by adverse conditions such as rain, snow, ice, or sand, is to direct the radio operator to proceed as follows:

1. Place the antenna change-over switch to the fixed antenna position.

2. Release approximately 50 feet of the trailing wire antenna.

3. Ground the trailing wire antenna post directly to the airplane structure (for instance, the metal support for the transmitter tuning units).

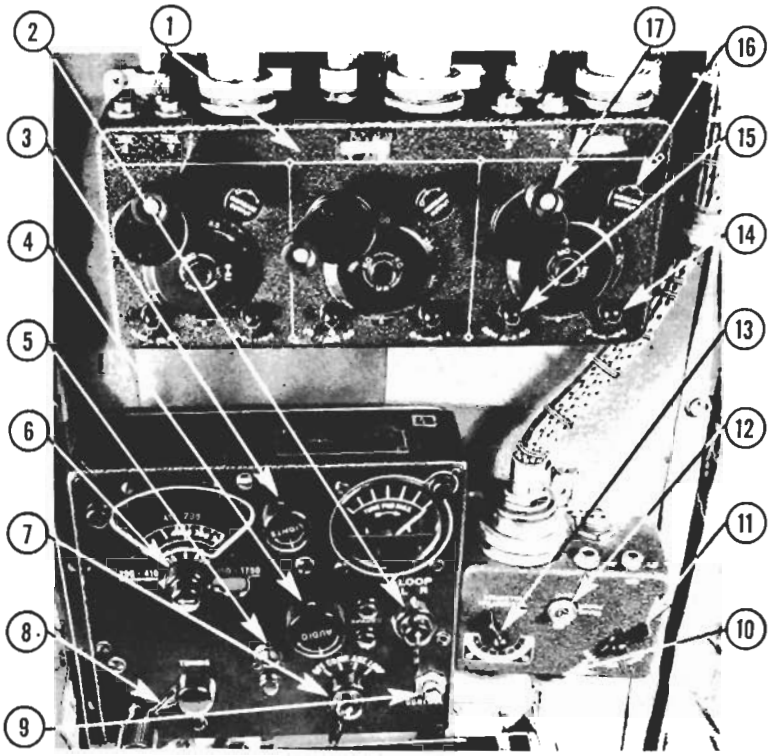
CAUTION

Do not extend retractable rod antenna at speeds greater than 240 mph.

(2) INTERPHONE EQUIPMENT RC-36. - An interphone jack box is provided for both pilot and copilot. Refer to section I, paragraph 10.

(3) COMMAND SET SCR-274-N. - The command set is designed for short-range operation and is used for communicating with nearby aircraft for tactical purposes and with ground stations for navigational and traffic control purposes.

(a) RECEIVING. - The interphone jack box (figure 22) switch must first be placed in the "COMMAND" position. The receiver control box (figure 29) is divided into three sections, each controlling the par-



KEY TO FIGURE 29

1. COMMAND RECEIVER CONTROL UNIT
2. LOOP CONTROL SWITCH
3. LIGHT CONTROL SWITCH
4. VOLUME CONTROL
5. CONTROL INDICATOR LAMP
6. BAND SELECTOR KNOB
7. POWER SWITCH
8. TUNING CRANK
9. CONTROL PUSH BUTTON
10. TRANSMITTING KEY
11. TRANSMISSION SELECTOR SWITCH (TONE-CW-VOICE)
12. TRANSMITTER POWER SWITCH
13. CHANNEL SELECTOR SWITCH
14. A-B CHANNEL SWITCH
15. SIGNAL SELECTOR SWITCH
16. VOLUME CONTROL
17. TUNING CRANK

Figure 29 - Radio Controls, Pilot's Compartment Ceiling

ticular receiver to which it is connected. Reception of a signal of a specific frequency as indicated on the dial is accomplished by the use of the section of the receiver control box which controls the particular receiver involved. The desired receiver is turned on and off by a switch in the left forward corner of the control box section used. This switch, in addition to having an "OFF" position, has two selective positions marked "CW" and "MCW," which indicate the type of signal which is to be received. The "A-B" switch should be left in the "A" position at all times and need not be turned off when the receivers are turned off.

NOTE

When tuning receiver for a definite frequency, always turn dial a little to each side of the frequency calibration mark to find the point where the signal is the strongest.

(b) TRANSMITTING.

1. Before transmitting, adjust radio receiver to the same frequency as the station with which you desire to talk, and listen in to be sure that the operator is not talking to someone else. If the station is transmitting, take advantage of the opportunity to more accurately set the airplane receiver on the assigned frequency, and when the other operator is finished, proceed with your transmission.

2. Throw the "OFF-ON" switch (figure 29) on the transmitter control box to the "ON" position. Select type of transmission desired with switch marked "TONE-CW-VOICE." With the switch in the "VOICE" position, the microphone from any interphone jack box switched to "COMMAND" position will be operative and voice will be transmitted when the push-to-talk button on the control wheel is pressed. With the switch turned to the "CW" position, a continuous wave, or unmodulated signal, will be transmitted and with the switch in the "TONE" position, a modulated tone signal is transmitted. Greatest effective range can be obtained on "CW." Range is most limited when operating on "VOICE."

3. On both the "CW" and "TONE" positions, the microphones are inoperative, and signalling by code is accomplished by a key which is located on the forward end of the transmitter control box.

NOTE

To reduce battery drain and to increase dynamotor life, the "TONE-CW-VOICE" switch should be left on "VOICE" unless continued use on "CW" or "TONE" is expected.

(4) RADIO COMPASS SCR-269.

(a) Set the interphone jack box switch (figure 22) to the "COMP" position, if aural reception of the

radio compass receiver is desired. If only visual indication is desired, the switch does not have to be set at the "COMP" position.

(2) The radio compass equipment is designed to perform the following functions.

1. Aural reception from the fixed antenna or from the rotatable loop. For signal reception during interference caused by precipitation static or proximity of signals, the loop will prove superior.

2. Aural null directional indication of an incoming signal with the loop only in use.

3. Visual unidirectional indication of an incoming signal.

(c) The receiving unit is turned on or off by a switch on the face of the remote control box, which, in addition to having an "OFF" position, has three other positions: "COMP," "ANT," and "LOOP."

1. With the switch at the "COMP" position, both the rotatable loop and the fixed antenna are in use.

2. In the position marked "ANT" only the fixed antenna is in use.

3. With the switch turned to the "LOOP" position, only the rotatable loop is in use.

(d) If the green indicator on the face of the control box does not light, depress button marked "CONTROL" to establish control of the set at this unit. Select frequency band desired as indicated in kilocycles on the face of control box and tune by use of the crank to the desired frequency. The loop may be rotated to any position as indicated on the radio compass azimuth indicator by use of switch marked "LOOP L-R." (See figure 29.) This variable operation is possible only when operating on "LOOP" position of the selector switch. During periods of severe precipitation static operate on "LOOP." For best aural reception rotate the loop by means of the "LOOP L R" switch until a maximum signal is obtained. Proper volume may be obtained by use of knob marked "AUDIO."

(5) MARKER BEACON EQUIPMENT RC-43. Since the operation of the marker beacon equipment is fully automatic, no manual operation is necessary. As the ship passes over a fixed point from which a marker beacon signal is being transmitted, the signal is picked up by the receiver, causing the indicator to flash on. Observe the pilot not to have passed over a marked beacon. The marker beacon equipment is normally turned on when the radio compass is put into operation. The position of the interphone jack box switch does not affect the operation of the marker beacon equipment.

(6) LIAISON SET SCR-287.

(a) The liaison equipment is to be used for long range communication. Limited control is available to the pilot. The type of receiver and transmitter desired must be forwarded to the radio operator, who will in turn put the radio equipment in operating condition.

(b) Set the interphone jack box switch in "LIAISON" position to receive or transmit with the liaison equipment.

(c) It is possible for all crew members to receive on this equipment, but only the pilot, copilot, and radio operator may transmit.

(7) RADIO SET SCR 535 (IFF). - The portable "OFF-ON" switch for this equipment is located on the top of the instrument panel hood. The two destroyer push-button switches are located to the left of the "OFF ON" switch. The destroyer switches should be used only when it is contemplated abandoning the airplane over enemy territory. When both destroyer push buttons are pressed simultaneously a detonator is set off in the receiver which is located in the radio compartment. The explosion of the detonator will destroy the receiver internally. No damage should be done to either the airplane or personnel at the time of destruction of the set, but bodily contact with the receiver at the time of detonation should be avoided.

NOTE

Regeneration adjustment of the IFF set must be made on the ground prior to flight in order to insure correct operation of the equipment.



3. FLIGHT INSTRUCTIONS.

a. BEFORE ENTERING PILOTS' COMPARTMENT.

(1) Check weight and balance data, form F, AN 01-1-40.

(2) Check forms 1 and 1A and sign exceptional release if necessary.

(3) Check flight engineer's report of preflight inspection.

b. ON ENTERING PILOTS' COMPARTMENT. - Check for all flights:

PILOT	COPILOT
(1) Emergency ignition switch "ON."	
(2) Check each battery switch separately with either inverter on.	
(3) Master battery switches "ON."	
(4) Turn hydraulic pump switch "ON." If it is momentary "AUTO-MANUAL" type, it should remain in "AUTO" unless the pump fails to operate.	
(5) Landing gear control switch in neutral.	
(6) Flap control switch in neutral.	
(7) Have copilot set parking brake.	(7) Set parking brake at command of pilot.
(8) Ascertain free movement of flight control column, wheel and rudder pedals to the extremities of their operating range.	

c. SPECIAL CHECK FOR NIGHT FLIGHTS.

(1) Master battery switches "ON."

(2) Turn control panel lights "ON."

(3) Turn side control panel lights "ON."

(4) Test operate the instrument panel lights.

(5) Test operate the landing lights.

WARNING

Do not permit lights to burn more than 5 seconds during test.

(6) Test operate the identification lights.

(7) Test operate the passing lights.

(8) Test operate the position lights.



d. STARTING ENGINES.

PILOT	COPILOT
(1) If the engines have stood for over 2 hours, have the propellers turned over three complete revolutions by hand. Be sure ignition switches are "OFF."	
	(2) Order flight engineer to open manual shut-off valve and set selective check valve to "SERVICING" position.
	(3) Check hydraulic pressure, both gages (600 to 800 pounds per square inch). Order flight engineer to close manual shut-off valve. Set selective check valve to "NORMAL" position.
(4) Cabin heat control in "OFF" or "COLD" position.	(4) Open cowl flaps and return valves to "LOCKED" position.
(5) Move turbo controls to "OFF."	(5) Fuel transfer valves and pump switch should be "OFF." Have flight engineer check them.
(6) Post fire guard.	(6) Set fire extinguisher selector valve (if installed) to engine being started.
(7) Open all fuel shut-off valves.	(7) Move intercooler controls to "COLD."
(8) Crack throttles (approximately 1000 rpm).	(8) Turn carburetor air filters "ON" when directed by pilot.
(9) Direct copilot to open carburetor air filters.	(9) Move mixture controls to "ENGINE OFF."
(10) Set propeller controls for high rpm.	(10) Set primer to "OFF" position.
(11) Turn magneto switch for engine affected to "BOTH."	(11) Start No. 3 fuel booster pump for primer pressure. It should be 6 to 8 pounds per square inch.
	(12) Start fuel booster pump for engine affected.
(13) Direct copilot to start engines. Recommended starting order is 1-2-3-4.	(13) Start engines when directed by pilot.
	(a) OLD-TYPE STARTER.
	1. Move starter switch of engine affected to "START" position and hold for approximately 30 seconds.
	2. While starter switch is in "START" position, unlock primer, set to engine affected, and expel air from line by pumping until a solid charge of fuel is obtained.
	3. When directed by pilot, move starter switch to "MESH" position.
	(b) NEW-TYPE STARTER.
	1. Throw "START" switch to engine affected and energize for 12 seconds.

PILOT	COPILOT
	2. Throw "MESH" switch while "START" switch is held on.
(14) When the engine fires, move the mixture control to "AUTOMATIC RICH."	(14) When the starter is meshed, prime with quick strokes (to atomize the primer charge) until the engine fires.
CAUTION Do not advance the throttles as lean mixture and backfire hazard will result.	
	(15) If necessary to prevent engine from quitting due to lack of fuel, pump primer with several slow strokes.
	CAUTION Return primer to "OFF" position.
	(16) Shut off booster pump if fuel pressure from engine pump remains steady.
	(17) If engine stops, return mixture control to "ENGINE OFF" immediately, cut ignition switch and repeat the starting procedure.
(18) If no oil pressure is indicated within 1/2 minute after starting, direct copilot to stop engine with mixture control. Cut ignition and investigate.	(18) After engine starts, check for indication of oil pressure. If no pressure is indicated within 1/2 minute, notify pilot; move mixture control to "ENGINE OFF" when directed by pilot.
(19) In case of fire in the exhaust system, run up the engine in an attempt to blow out the fire. If this fails, direct copilot to stop the engine.	(19) When directed by pilot, stop engine by moving mixture control to "ENGINE OFF."
(20) Close cowl flaps if the fire is in nacelle 1 or 2.	(20) Close cowl flaps if the fire is in nacelle 3 or 4.
(21) If fire is not smothered by closing the cowl flaps, close fuel shut-off valve, stop booster pump, and direct copilot to pull fire extinguisher, both charges if necessary.	(21) Pull fire extinguisher charges (if available) at command from pilot.
	NOTE If engine accessory cowling is not installed, it is unlikely that the fire can be extinguished by the CO_2 system. External fire extinguishers must, therefore, be used.
(22) Before resuming operations after fire, be sure that CO_2 cylinders are replaced.	

★

e. ENGINE WARM-UP.

PILOT

(1) When oil temperature begins to rise and oil pressure is 50 pounds per square inch, open throttles 1000 to 1250 rpm.

(2) When engines are thoroughly warmed, the rpm may be increased for instrument check.

COPILOT

(1) Notify pilot when oil temperature begins to rise and oil pressure is 50 pounds per square inch.

(2) Notify pilot when maximum temperature and pressure values are reached.

CAUTION

2500 rpm must not be maintained for more than 1/2 minute and the following values must not be exceeded:

Fuel pressure	16 lb/sq in.
Oil pressure	80 lb/ sq in.
Oil temperature	88°C (190.4°F)
Cylinder temperature	205°C (401°F)

f. EMERGENCY TAKE-OFF.

(1) If the airplane has been on the "alert," the engines will have been started, and will be warm and ready for take-off by the time the flight crew gets within the airplane. The pilot will proceed with a routine take-off, being careful not to exceed 46 inches Hg manifold pressure.

(2) If an emergency take-off is necessary with cold engines, due to the lack of a ground crew, the following procedure should be followed:

(a) Start engines, using oil dilution as soon as engines fire in order to get minimum oil pressure of 70 pounds per square inch.

(b) Fuel pressure should be at least 12 pounds per square inch.

(c) Set wing flaps for take-off, leave cowl flaps less than 1/3 open to expedite warm-up. Proceed with take-off. Do not exceed 46 inches Hg manifold pressure.

g. ENGINE AND ACCESSORIES GROUND TEST.

PILOT

(1) Direct gunner to secure lower turret with guns pointing rearward.

(2) Set altimeter.

(3) A.F.C.E. switches "OFF," all knobs on control panel, "POINTERS-UP," turn control, "CENTERED."

(4) Set propeller controls for high rpm and lock.

COPILOT

(1) See that all doors and hatches are closed.

(2) Hydraulic pressure should be 600 to 800 pounds per square inch on each gage.

(3) With ignition and battery switches "ON," hydraulic switch in "AUTO," warning and indicator lights should be:

Tail wheel unlocked - On (red)
Landing gear - On (green)
Hydraulic pressure: Service - Off.
Emergency - Off.
Vacuum - Off.

(4) Check all fuel quantities.

PILOT	COPILOT
(5) Turn command radio on.	(5) Set intercooler controls to "COLD" unless icing conditions exist.
(6) Flight controls unlocked. Move them to the limits of their ranges to insure free operation.	(6) Cowl flaps should be open. Check visually.
	(7) Wing flaps up. Switch in neutral.
	(8) Tail wheel unlocked. Locking handle should be in up position.
(9) Contact control tower for clearance.	
(10) Signal ground crew to remove wheel chocks.	
(11) With mixture controls in the "AUTOMATIC RICH," check ignition at 1900 to 2000 rpm.	(11) Check the following during ignition check:
NOTE The rpm drop should not exceed 100 when switching from two magnetos to one.	Fuel Pressure: Desired - 12 to 16 lb/sq in. Maximum - 16 lb/sq in. Minimum - 12 lb/sq in. Oil Pressure: Desired - 75 lb/sq in. 80 lb/sq in. 70 lb/sq in. Oil Temperature: Desired - 70°C (158°F) Maximum - 88°C (190°F) Minimum - 60°C (140°F) Cylinder Temperature: 205°C (401°F) Maximum
(12) Check propeller governor at 1500 rpm by moving control to low rpm. When rpm decreases to approximately 1100, return control to high rpm position and lock.	
(13) Run up each engine individually and adjust supercharger regulator control stops for 46 inches Hg manifold pressure at full throttle and 2500 rpm.	(13) Notify pilot if any temperature or pressure reading is not satisfactory.
IMPORTANT This adjustment must be made as quickly as possible and must not exceed 1/2 minute for each engine.	
(14) Set trim tabs in neutral.	
(15) Check flight controls.	(15) Turn all fuel boost pumps "ON."
WARNING Operate to full extent of their ranges to insure free and proper movement.	
(16) Close window.	(16) Close window.

h. TAXYING.

PILOT	COPILOT
(1) Inboard throttles may be locked for taxying with outboard engines.	(1) Notify pilot if: Cylinder temperature exceeds 205°C (401°F). Oil pressure exceeds 75 pounds per square inch or is less than 15 pounds per square inch for idling engines. Oil inlet temperature exceeds 70°C (158°F). Fuel pressure is over 16 pounds per square inch or under 12 pounds per square inch.
	(2) Lock tail wheel (warning lamps off) after airplane has taxied to take-off position.

i. TAKE-OFF.

PILOT	COPILOT
(1) Refer to the Take-Off Chart, Appendix II.	
(2) Turn generator switches "ON."	
(3) Open throttles slowly to FULL THROTTLE (3 to 5 seconds). Hold three-point position until airplane leaves ground.	
(4) With a runaway turbo or propeller, follow the following instructions:	
(a) THROTTLE BACK FIRST.	
(b) Move turbo control to "OFF."	
(c) If necessary, set propeller controls (figure 40-3) in "LOW RPM." There is small likelihood of a runaway turbo, but the danger is great if it occurs during a take-off. The pilot MUST be alert during the take-off to note immediately and correct any excessive manifold pressure.	
(5) When airplane is clear of the ground, direct copilot to retract the landing gear.	(5) Retract landing gear at command from pilot.
(6) Accelerate to speed for cruising climb.	(6) Cylinder head temperatures must not exceed 260°C (500°F) (5 minutes maximum). Oil pressure - desired - 80 lb/sq in. Oil Temp - desired - 70°C (158°F) Fuel Pressure - 12 to 16 lb/sq in.
	(7) Adjust intercooler control to "COLD" unless icing conditions prevail.

j. ENGINE FAILURE DURING TAKE-OFF.

PILOT

(1) Failure of an engine during take-off may not be noticeable immediately except for a resultant swing. If, therefore, a swing develops, and there is room to close the throttles and pull up, this should be done.

(2) If it is necessary to continue with the take-off, even though one engine has failed, hold the airplane straight by immediate application of rudder. Gain speed as rapidly as possible. See that the landing gear is up, or coming up, and feather the propeller of the dead engine. Retrim as necessary.

COPILOT

(1) Press proper propeller feathering switch when ordered by pilot.

k. CLIMB. (Refer to climb chart, Appendix II.)

PILOT

(1) Reduce manifold pressure with supercharger controls.

(2) Reduce rpm as required for climb.

(3) Make a visual check of engines 1 and 2.

(4) Adjust trim tabs as required.

(5) Order copilot to set carburetor air filter switch to "FILTER OFF" at 8000 feet unless dust conditions are found above that altitude.

COPILOT

(2) Adjust cowl flaps as required to maintain proper cylinder head temperature.

(3) Make a visual check of engines 3 and 4.

(5) When ordered by pilot, move switch to "FILTER OFF."

WARNING

Switch must never be left in the "FILTER ON" position above 15,000 feet.

l. LEVEL FLIGHT.

PILOT

(1) Refer to Cruising Control Charts, Appendix II.

(2) Use full throttle and set power with turbo regulators at all altitudes.

COPILOT

(2) Set mixture controls to "AUTOMATIC LEAN," below 2100 rpm, 30 inches Hg manifold pressure.

CAUTION

Do not exceed 30 inches Hg manifold pressure below 2100 rpm.

CAUTION

Instantaneous load factors above the allowable can be reached very easily with rough elevator control movements. In turbulent air or in combat maneuvering, corrections should be made very smoothly.

PILOT	COPILOT
	(3) Adjust cowl flaps as required to maintain proper cylinder head temperatures.
	(4) Stop booster pumps until needed (which will be above 15,000 feet).
	(5) Begin flight performance log and made entries in Form I as required.

m. PROPELLER FEATHERING.

PILOT	COPILOT
(1) TO FEATHER A PROPELLER.	
(a) Notify copilot to stop engine affected.	(a) Move mixture control of affected engine to "ENGINE OFF."
(b) Turn automatic flight control equipment switches "OFF."	(b) Stop the booster pump if running.
(c) Notify copilot to press proper feathering switch.	(c) Press proper feathering switch.
(d) When propeller stops, turn proper ignition switch to "ENGINE OFF."	(d) Close cowl flaps of engine affected.
(e) Close throttle.	
(f) Adjust trim tabs as required.	
(g) Turn automatic flight control equipment switches "ON."	
(h) If the engine is not to be restarted, order engine fuel transferred to other tanks as required.	(h) Assist aerial engineer to transfer fuel from the dead engine tank.
(i) When No. 2 engine is affected:	
1. The glycol pump is inoperative. If cold air is not desired in the cabins, shut off heating and ventilating system by moving control handle fully aft.	
2. When one vacuum pump is inoperative, (engine No. 2 or 3): Set vacuum pump selector ("GYRO INSTR.") valve to the other vacuum pump. (De-icer pressure will thus be reduced and de-icer vacuum will not be available. De-icer system will, therefore, operate inefficiently.)	
(2) TO UNFEATHER A PROPELLER.	
PILOT	COPILOT
(a) Notify copilot which engine is to be restarted.	(a) Set propeller control to "LOW" rpm.
(b) Turn automatic flight control equipment switches "OFF."	(b) Set intercooler control to "HOT" position.

PILOT	COPILOT
	(c) Close cowl flaps.
(d) Crack proper throttle to 1000 rpm approximately.	(d) Start proper booster pump (if above 15,000 feet).
(e) Turn ignition switch to "BOTH."	(e) Check fuel quantity in proper tank.
(f) Press proper feathering switch and hold it closed until engine speed reaches 1000 rpm.	(f) When engine speed reaches 1000 rpm, move mixture control from "ENGINE OFF" to "AUTOMATIC RICH."
(g) Open throttle slowly to 1200 rpm.	
(h) Adjust trim tabs as desired.	
(i) Maintain 1200 rpm until notified by copilot that oil temperature is 70°C (158°F).	(i) Notify pilot when oil temperature reaches 70°C (158°F).
	(j) When cylinder head temperature reaches 205°C (401°F), open cowl flaps as required for continuous operation.
(k) Synchronize manifold pressure and rpm with other engines.	(k) Adjust intercooler control as required.

CAUTION

Above 15,000 feet, power must be adjusted with turbo control - full throttles.

(l) Adjust trim tabs as required.

(m) Turn automatic flight control equipment switches "ON."

NOTE

When No. 2 propeller is unfeathered, the pilot may turn on the heating and ventilating system by moving the control to any position between one-half and fully forward.

n. GENERAL FLYING CHARACTERISTICS.

(1) GENERAL STABILITY.

(a) Increasing the power on the inboard engines causes the airplane to become slightly tail heavy, while a change of power on the outboard engines has no appreciable effect upon the trim.

(b) Closing the cowl flaps on the inboard engines causes a similar tail heaviness, but cowl flaps on the outboard engines have a negligible effect upon the trim.

(c) With the airplane properly trimmed for a landing with power off and flaps down, the pilot may apply power, throw the flap switch into the up position and go around with no change in trim tab setting if a second approach is necessary. The flaps retract at a satisfactorily slow rate.

(2) TAKE-OFF. - During the take-off run, directional control should be maintained with rudder movement and throttles, differential throttling being done with the outboard engines as much as possible.

(3) CLIMB. - The airplane will require very little elevator trim and the elevator control pressure will build up rapidly as the climbing speed is reduced below normal.

(4) LEVEL FLIGHT. - In normal flight, turns can be made very smoothly with aileron control only. In instrument flight, the pilot should pay special attention

to holding the wing level, because the directional stability produces a noticeable turning tendency with one wing down.

WARNING

Care should be taken to avoid excessive use of the ailerons.

(5) ROUGH AIR OPERATION.

(a) The ailerons and rudder can be used without concern regarding excessive loads. It is almost impossible to damage the system without a deliberate attempt to do so. The forces required are small enough and the resultant responses large enough to maintain ample control of the airplane.

(b) In the case of the elevators, however, care must be exercised to assure smooth operation. In thunderstorms, squalls, and in or near extremely turbulent cumulous clouds, it is possible to develop excessive load factors with the elevators unless proper care is exercised.

(c) Operation in rough air should be made on the basis of holding constant the air speed with the elevator. Corrections for changes in altitude must be done with power, and for very rapidly rising air currents, it may be necessary to lower the landing gear.

(d) The airplane should not be dived through a cloud layer or through rough air at the maximum diving speed, nor should high-speed flight be attempted in rough air.

(6) OBTAINING MAXIMUM PERFORMANCE.

(a) The ceiling and climb at 35,000 feet are as great or greater than that of many fighter airplanes, but the high speed is not as great as most fighters at normal altitudes; therefore, in order to outperform any enemy at 35,000 feet it will be necessary to outclimb him rather than to outdistance him.

(b) The increase of speed obtained by nosing the airplane down below the horizontal at rated power and at any high power condition is smaller than that obtained by fighters.

(c) In order to obtain maximum climb, the following technique should be used:

1. Maintain the proper climbing air speed (135 mph indicated).

2. In any emergency whatever, such as being pursued by the enemy, engine speed should be increased to 2500 rpm. The increase in rpm has a very appreciable effect on increasing propeller efficiency and rate of climb under conditions of climbing speed and high altitude, and, in addition, is not detrimental to the engine. The pilot should avoid the use of less than 2500 rpm when primarily interested in a high rate of climb at high altitudes.

3. 21,300 rpm has been determined to be the maximum operating turbo speed with a 5 percent overspeed allowance in emergencies. This would provide an emergency rating of 22,400 rpm. At any altitude greater than 30,000 feet and at any power obtained in automatic rich (with 2300 rpm or 2500 rpm, full throttle and turbos set for manifold pressures indicated in the following table), the exhaust gas temperatures are dropping rapidly and it is very unlikely that critical temperatures will be approached. The following tentatively determined manifold pressures will permit safe operation of the turbo under the given conditions:

Altitude	Manifold Pressures giving rated power at 2300 engine rpm and 21,300 turbo rpm		Manifold Pressures giving military power at 2500 engine rpm and 21,300 turbo rpm	
S.L.	Rated Power	39.0	Military Power to 28,000 ft	47 in.
10,000		38.0		46 in.
20,000		37.5		45 in.
30,000		37.0		41.5 in.
31,000		37.0		40.0 in.
32,000	Decreasing Power	36.5		38.5 in.
33,000		35.0		37.0 in.
34,000		33.5		35.0 in.
35,000		32.0		33.0 in.
	These manifold pressures not allowable below 2300 rpm		These manifold pressures not allowable below 2500 rpm	

NOTE

This table is based on the best present available information for maximum performance at 55,000-pound gross weight with carburetor air filters closed. All four turbo installations are not identical and hence, operation according to the above table will not result in identical turbo rpm for all engines.

4. The outboard engines have higher critical altitudes than the inboards by approximately 2000 to 3000 feet, and the inboard engine without boilers in the stack has a 1500-foot higher critical altitude than the engine with the boilers in the stack. The critical altitude of the outboard engines as far as limiting turbo rpm is concerned is 31,000 feet.

5. The above table actually applies only to the outboard engines. However, the differences between the inboard and outboard engines are covered by the margin of safety incorporated in the design of the turbo itself. Even though 22,400 rpm are allowable for military power operation, the right-hand column of the above table, is made for only 21,300 rpm.

(7) LANDING. - During the approach for landing very little change in elevator trim will be required. As the flaps are lowered the airplane becomes slightly tail heavy, but if it is trimmed slightly nose heavy at 147 mph with flaps up, it will be properly trimmed at 120 mph with flaps down. This is a satisfactory approach speed for gross weights below 50,000 pounds.

o. STALLS.

(1) Stalling characteristics are very satisfactory. Under no condition is there any sharp tendency to roll. Yawing is sufficiently suppressed to make any rolling at the stall of a very mild nature. Under all conditions a stall warning of several miles per hour is indicated by buffeting of the elevators.

(2) A pitching motion started by the elevators should be damped slowly. It will easily reduce the air speed well below the stall unless it is deliberately stopped.

(3) Full flap reduces the stalling speed about 15 mph for gross weights between 40,000 and 45,000 pounds, but full military power for the same loading conditions may reduce the stalling speed another 15 mph. Accidental or deliberate yawing will increase the stalling speed and increase any tendency to roll at the stall.

(4) The ailerons have a tendency to overbalance and reverse effectiveness at the stall. For example, if the left wing tends to drop at the stall and right aileron control is applied in an attempt to raise the left wing, the aileron operating forces will tend to decrease and cause full aileron deflection, but the response will be an increase in the roll to the left. THE PROCEDURE IN RECOVERING FROM A STALL IS TO HOLD THE AILERONS NEUTRAL AND REFRAIN ENTIRELY FROM THEIR USE.

(5) Procedure for recovering from a stall is normal. The air speed for normal flight must first be regained by smooth operation of the elevators. This may put the airplane into a dive of 30 degrees or less. During the process of regaining air speed the rudder may be used to maintain laterally level flight for lateral control, but not until the air speed is regained. RECOVERY FROM THE DIVE MUST BE DONE IN A SMOOTH MANNER. Failure to make a smooth recovery may be a restalling of the airplane or a structural failure, both due to excessive load factors.

(6) Air-speed increase necessary to regain normal flight need not generally be more than 20 mph, and possibly, after practice, even less.

p. SPINS. - Inadvertent spinning is very unlikely, as stability and damping are very high. The airplane is not designed for spinning, and this maneuver should never be attempted.

q. DIVES. - Airplanes having modified elevators are limited to a maximum diving speed of 270 mph. Those airplanes whose elevators have not been modified are restricted to 220 mph maximum diving speed. See Warning Placard!

When diving, it is essential that the sensitivity of the elevator trim tab be kept constantly in mind. In making dives the elevator trim tabs must be set during the dive to maintain zero elevator force and must be used with great care during recovery.

r. PRECAUTIONS.

(1) MAXIMUM LOAD.

(a) B-17F airplanes, with modified landing gear and added chord-wise wing tip tanks, can be flown up to and including a gross weight of 64,500 pounds, with the following restrictions:

(b) At 64,500 pounds, the extra wing tip tanks must be full to obtain the effect of a relieving load on the wings in flight. Care must be exercised in taxying avoiding rough ground. Take-offs, above a gross weight of 56,000 pounds may be made only on smooth fields or prepared runways. All pivot turns on one wheel, while taxying, will be avoided.

(c) All B-17 type airplanes, equipped with extra wing tip chord-wise tanks, must be operated in accordance with (b) preceding, whenever the wing tip tanks are more than half full. Maximum permissible indicated air speed of B-17F airplanes, with extra wing tip tanks full, must be limited to 230 mph, when loaded to 64,500 pounds. Maximum maneuver permissible at 64,500 pounds; positive, 2.056; negative, 1.22; landing gear, 2.1.

(2) 1600-POUND BOMBS. - Some B-17F airplanes do not have a complete set of B-10 bomb shackles. 1600-pound bombs may be carried on the B-7 bomb shackle with these restrictions: If an airplane returns to base with 1600-pound bombs remaining on the racks, they shall be released, in the safe condition, over water or the safest available area. The maximum permissible gross weight of the airplane will not be exceeded when carrying 1600-pound bombs. The pilot will guard against any severe maneuvering of airplane.

s. APPROACH AND LANDING.

PILOT	COPILOT
(1) Check center of gravity location for landing by means of the load adjuster.	(1) SELECTIVE CHECK VALVE MUST BE IN "NORMAL" position.
(2) Set altimeter to airport pressure altitude.	(2) Set mixture controls in "AUTOMATIC RICH."
(3) Notify radio operator to retract trailing antenna.	(3) Set intercooler controls in "COLD," unless icing conditions exist.
(4) Turn automatic flight control equipment switches "OFF."	(4) Radio control tower or landing clearance.
(5) Direct copilot to adjust carburetor air to "FILTERS ON."	(5) When directed by pilot, throw carburetor air filter switch to "FILTER ON."
(6) Move supercharger controls to full "ON," and propeller controls to "MAX. CRUISE." (2100 rpm).	
(7) Shut off de-icer system, if operating.	(7) Check instruments.
(8) Order copilot to extend landing gear.	(8) Extend landing gear when directed by pilot (green signal light on).
(9) Check position of ball turret. Guns should be horizontal and pointing rearward.	(9) Tail wheel should be locked (warning light off), locking lever flush with floor.
(10) Check hydraulic pressure; it should be 600 to 800 pounds per square inch on both gages.	
(11) Operate brakes. Hydraulic pressure should remain above 600 pounds per square inch. If main brakes are inoperative, prepare for emergency landing.	
	(12) Check cowl flap valves. They must be in "LOCKED" position to guard against loss of oil supply through leaks in cowl flap actuating mechanisms.
(13) After speed has dropped below 147 mph, order copilot to lower wing flaps.	(13) Lower wing flaps when directed by pilot.
(14) Adjust trim tabs as required.	
(15) Order copilot to call off air speed as required.	(15) Call off air speeds when directed by pilot.

t. EMERGENCY TAKE-OFF IF LANDING IS NOT COMPLETED.

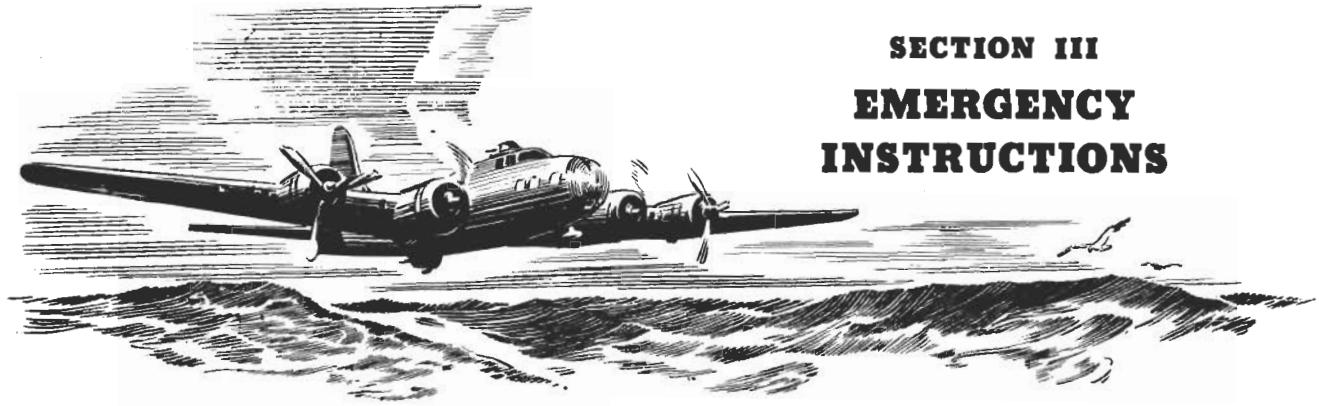
(1) Open throttle wide.

CAUTION

Do not exceed 46 inches Hg manifold pressure.

PILOT	COPILOT
(2) Increase propeller speed to 2500 rpm.	
(3) Order copilot to raise landing gear and proceed with a normal take-off.	(3) Raise landing gear when directed by pilot.
(4) Order copilot to raise wing flaps after 500 feet altitude has been reached.	(4) Raise wing flaps when directed by pilot.
u. AFTER LANDING.	
(1) Move supercharger controls to "OFF" position.	(1) Raise wing flaps.
(2) Generator switches "OFF."	(2) Check cowl flaps "OPEN."
(3) Order tail wheel unlocked after taxi speed has dropped below 30 mph.	(3) Unlock tail wheel when directed by pilot (lever as nearly vertical as possible).
v. STOPPING OF ENGINES.	
(1) If parking brakes are set, do not permit them to remain so for very long if the brake drums are hot.	
(2) Idle engines at approximately 800 rpm until cylinder temperature gages show temperatures are 170°C (338°F).	
(3) If the airplane is to remain outside overnight, or if an engine start is anticipated in temperatures below 0° C (32°F), order copilot to dilute oil for 4 minutes maximum: During oil dilution period, operate supercharger controls continuously full open to fully closed in cycles of approximately 10 seconds, to dilute oil in supercharger regulator system.	(3) Close oil dilution switches when ordered by pilot.
(4) Set propeller controls in "HIGH RPM."	
(5) Before stopping engines, run at 1200 rpm for 30 seconds. Direct copilot to stop engines with mixture control.	(5) When directed by pilot, stop engines by moving mixture controls to "ENGINE OFF."
w. BEFORE LEAVING THE PILOT'S COMPARTMENT.	
Cut off all radio, de-icer, compartment, central control panel, and pilot's side control panel switches.	Complete Form 1.

Moor the airplane with the nose into the wind, set the parking brakes and lock the rudder and elevators. When attaching the mooring lines at the rope wells in the wings, allow approximately 16 inches slack in the line. This will prevent damage to the structure or loss of mooring control in case a tire goes flat with result and elevation of the opposite wing. Rudder and elevator locks will withstand gust loads from any direction up to 60 mph velocity.



SECTION III
EMERGENCY INSTRUCTIONS

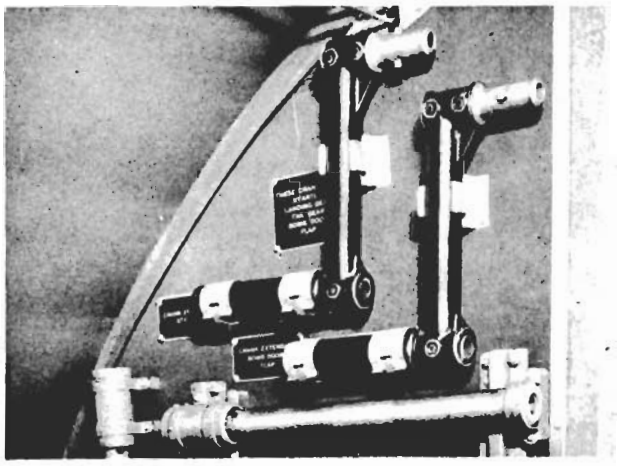


Figure 30 - Hand Cranks Stowed

1. HAND CRANKS.

Cranks for manual operation of landing gear, wing flaps, and bomb bay doors, and for hand starting of engines, are stowed on the aft bulkhead of the radio compartment. Crank extensions for use when operating engine starters, bomb doors, and wing flaps are stowed adjacent to the cranks.

2. EMERGENCY OPERATION OF LANDING GEAR.

Each main landing gear may be operated separately by means of a hand crank connection in the bomb bay, one to the left of the door in the forward bulkhead, and one to the right. To raise one of the landing wheels, insert the crank into the connection and rotate clockwise. Turn the crank counterclockwise to lower the wheel.

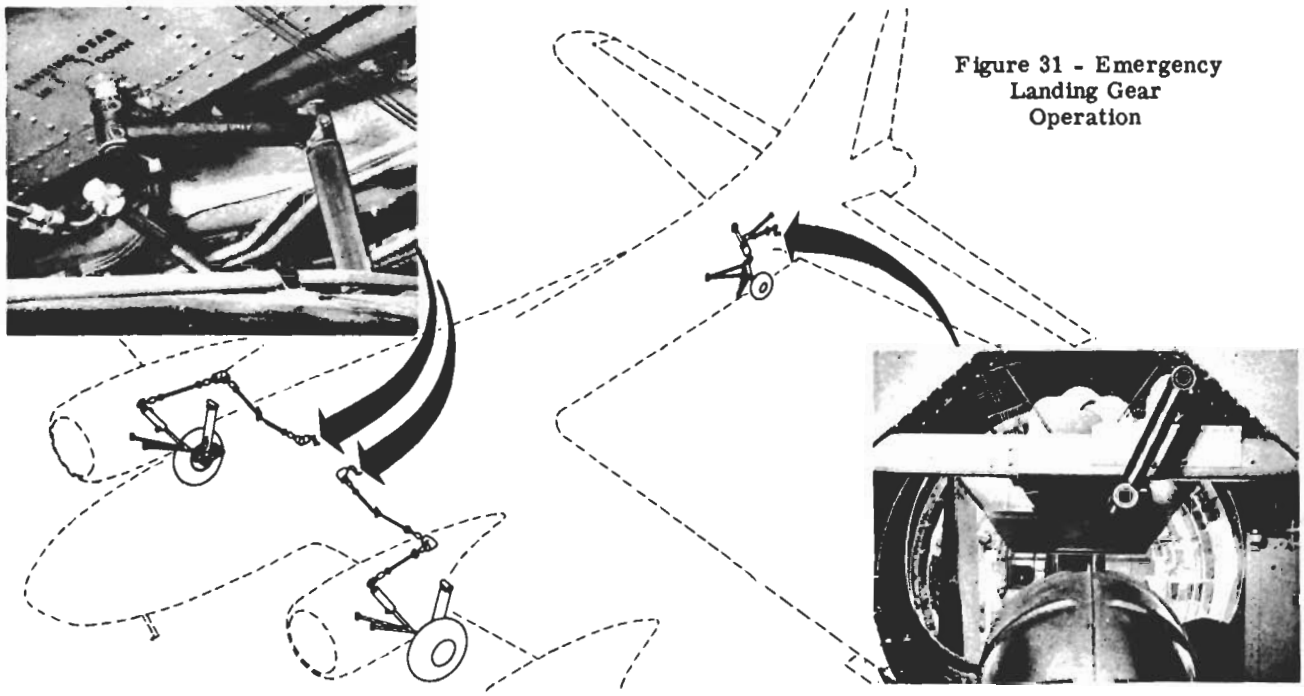


Figure 31 - Emergency Landing Gear Operation

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DANGER

Be sure the landing gear electric switch is "OFF" before you attempt hand cranking.

3. EMERGENCY OPERATION OF THE TAIL WHEEL.

The crank used for manual operation of the landing wheels is also used for manual operation of the tail wheel. Insert the crank into the connection in the tail wheel compartment and rotate as desired.

4. EMERGENCY OPERATION OF WING FLAPS.

Lift the camera pit door in the floor of the radio compartment and insert the hand crank into the torque connection at the forward end of the pit. Rotate the crank clockwise to lower the flaps and counterclockwise to raise them.

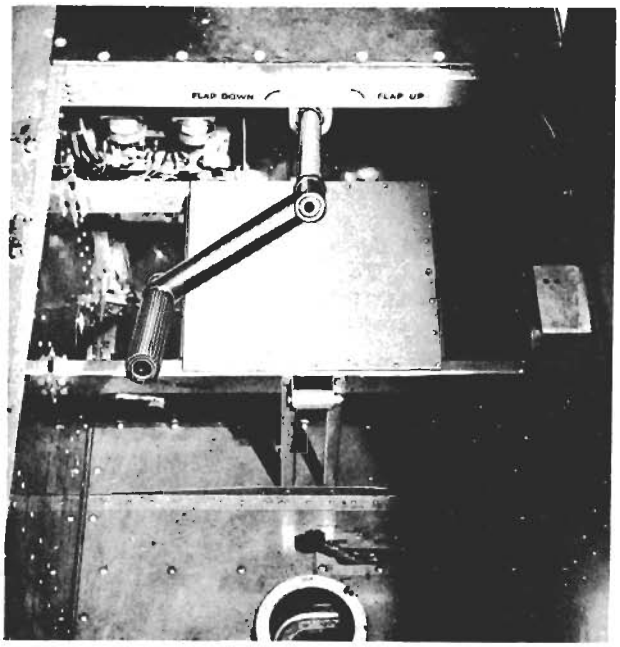


Figure 32 - Emergency Wing Flap Operation

5. EMERGENCY OPERATION OF BOMB BAY DOORS.

Insert the hand crank into the torque connection in the step at the forward end of the catwalk in the bomb bay and rotate clockwise to close the doors and counterclockwise to open them.

6. EMERGENCY BOMB RELEASE.

a. An emergency release handle is located at the pilot's left and another at the forward end of the catwalk in the bomb bay. Pull either handle through its full travel. The first portion of the stroke releases

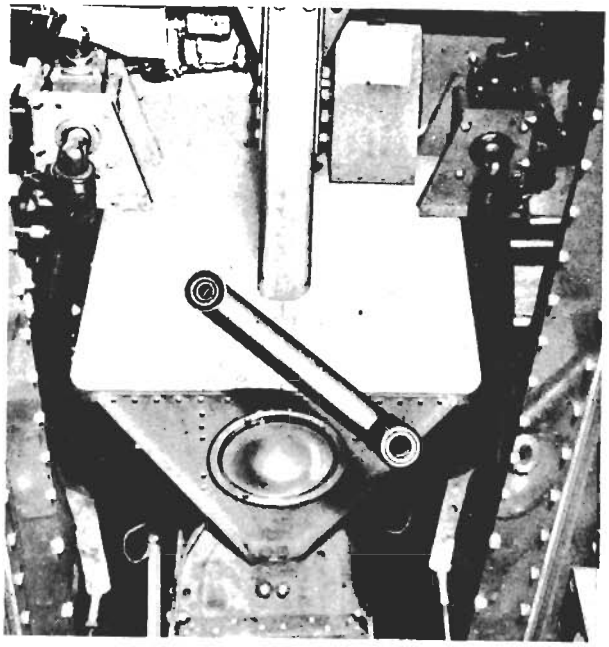


Figure 33 - Emergency Bomb Bay Door Operation

the bomb door latches, permitting the doors to open independently of the retracting screw, as shown in figure A. The latter portion of the stroke releases all external and internal bombs salvo and unarmed.

b. DOOR RETRACTION AFTER EMERGENCY RELEASE. - If the spring in the emergency release mechanism under the hinged door beneath the pilot's compartment floor has not entirely retrieved the linkage as shown in B, reset by pushing at the hinge of the link as shown in C. Operate the retracting screws electrically (or manually) to the fully extended position. This will engage the latches between the screws and door fittings as shown in D. The doors may now be retracted in the normal manner.

AT PILOT'S LEFT IN BOMB BAY

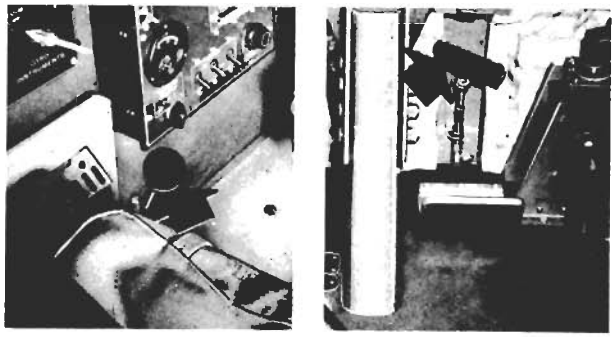


Figure 34 - Emergency Bomb Release Handles

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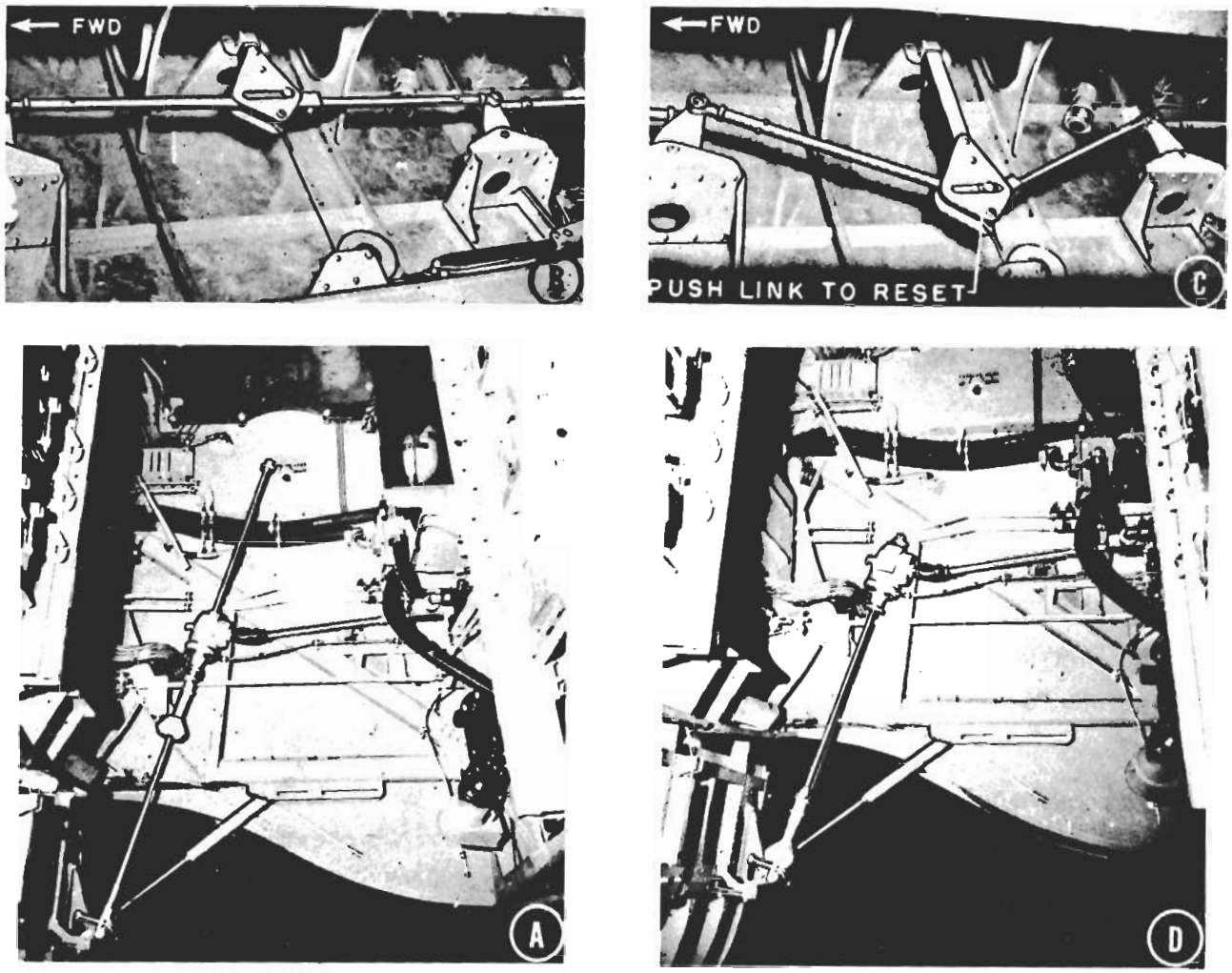


Figure 35 - Emergency Bomb Release Procedure

7. FIRE IN FLIGHT.

In case of engine or wing fires, open the emergency exits; signal stand by to abandon: one long ring (approximately 6 seconds). In case of a cabin fire, exits should NOT be open; signal stand by to abandon, exits closed: one long ring (approximately 6 seconds), and one short ring (approximately 2 seconds).

a. FUSELAGE FIRES.

(1) Three carbon dioxide fire extinguishers are located, one on the aft bulkhead of the navigator's compartment, one on the right rear bulkhead of the pilots' compartment, and one on the forward face of bulkhead of the radio compartment.

(a) To use; stand close to fire, raise horn, and direct gas to base of fire, holding on to rubber-insulated tubing.

WARNING

Do not grasp metal horn on top of cylinder. White discharge is "dry ice"; avoid frost bite.

(b) To shut off flow of gas, return horn to clip on side of cylinder. Extinguisher must be recharged after each use.

(2) Two carbon tetrachloride fire extinguishers are located one at the copilot's left, and one aft of the main entrance door.

(a) Stand as far as possible from the fire when using a carbon tetrachloride extinguisher; effective range is 20 to 30 feet.

(b) To operate, turn handle and pump plunger. Keep stream full and steady. To shut off, push handle in and turn until sealing plunger is depressed.

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WARNING

When sprayed on a fire, carbon tetrachloride produces phosgene, an extremely poisonous gas, which can be harmful even in small amounts; and if inhaled in excessive quantities may prove fatal. Do not use in a confined area and do not stand near fire. OPEN WINDOWS AND VENTILATORS immediately after fire is extinguished.

b. ENGINE FIRES DURING FLIGHT.

(1) If caused by fuel or oil leakage:

(a) Close fuel shut-off valve of engine affected.

(b) Feather propeller immediately. This stops the pumping of oil to the flames, and should be done before so much oil is lost that the propeller cannot be feathered and additional damage is caused by windmilling.

(c) Slow the air speed as much as possible.

(d) Close the cowl flaps.

(e) Pull CO_2 charge (if available).

CAUTION

Leave propeller feathered. Do not attempt to restart engine while hot.

(2) Fire in exhaust due to overrich mixture:

(a) Move mixture control to lean.

(b) Attempt to blow out fire by engine run-up.

(c) Close cowl flaps.

(d) Close fuel shut-off valve to engine affected.

(e) Pull CO_2 charge (if available).

8. EMERGENCY BRAKE OPERATION.

The emergency system operates the brake only. Pressure is applied through two hand-operated metering valves on the pilots' compartment ceiling; the left lever controls the left wheel, and the right lever controls the right wheel. If it is impossible to rebuild the pressure in the service system, use of the following procedure is recommended:

a. Manual shut-off valve "CLOSED."

b. Selective check valve "NORMAL."

c. Check pressure in emergency accumulator: 650 to 800 pounds.

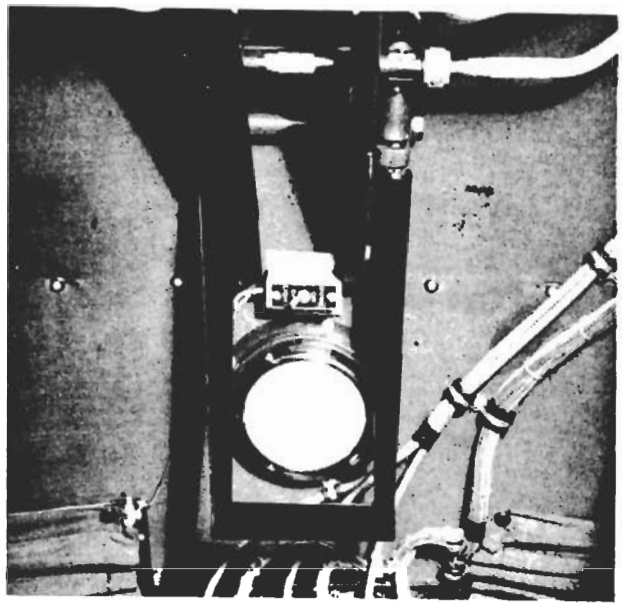


Figure 36 - Emergency Brake Handles

CAUTION

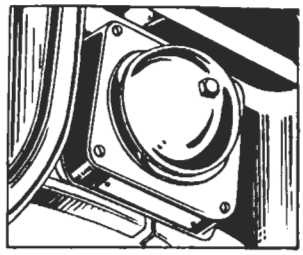
Do not attempt to raise the accumulator pressure with the hand pump.

d. Pilot: Operate throttle and rudder.

e. Copilot: Operate emergency brake control.

WARNING

DO NOT "PUMP" EMERGENCY BRAKES. The pressure supply is limited and repeated applications may result in complete loss of emergency braking control.



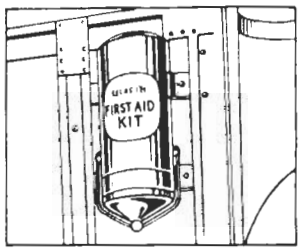
9. WARNING SIGNALS.

The pilot can communicate with the crew by means of the interphone system, phone call lamps, and the alarm bell system. For emergency purposes, the alarm bell should be used according to prearranged signals which are thoroughly understood by the crew. A toggle switch on the pilot's electrical control panel operates three bells located, one under the navigator's table, one on the wall above the radio operator's table, and one in the tail compartment above the tail wheel boot.

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10. FIRST-AID KITS.

First-aid kits are located on the bomb-sight storage box in the navigator's compartment, on the wiring diagram box on the back of the copilot's seat, and on the bulkhead forward of the lower turret.



11. ABANDONING AIRPLANE IN FLIGHT.

a. ESCAPE DOORS AND HATCHES. - All doors and hatches are quickly releasable. The side gunner's windows slide forward to open. Bomb doors may be opened by either of two emergency release handles, one at the left of the pilot and the other at the forward end of the catwalk in the bomb bay.

b. SIGNAL.

(1) Stand by to abandon: one long ring (approximately 6 seconds).

(2) Abandon airplane: three short rings (approximately 2 seconds each).

c. SWITCHES. - The situation will determine whether fuel and electrical systems should be turned off prior to abandoning the airplane. Under normal conditions outside of combat zones, the master ignition switch battery switches and fuel shut-off valve switches should be turned off.

12. CRASH LANDING.

a. SIGNAL.

(1) Stand by for crash landing; by interphone.

(2) Abandon: four short rings (approximately 1/2 second each).

(3) Pilot should:

(a) Cut engines.

(b) Turn master switch "OFF."

(c) Turn battery switches "OFF."

(d) Turn fuel shut-off valve switches "OFF."

b. EGRESS.

(1) All crew members will take proper stations, remove parachutes, and fasten safety belts upon receiving interphone warning.

(2) At the signal to abandon, all crew members will leave the plane through the most practicable exit. (See figure 37.)

(3) In addition to the seven standard exits, the two side windows in the pilot's compartment are possible exits.

(4) In case some of the exits are blocked by fire, damage, or congestion, it may be best to make exit through a rupture in the fuselage, if any have occurred. Caution is required in this process to avoid fatal cuts from metal or broken glass.

(5) If there is imminent danger of fire, all personnel should disperse at least 50 feet from the airplane.

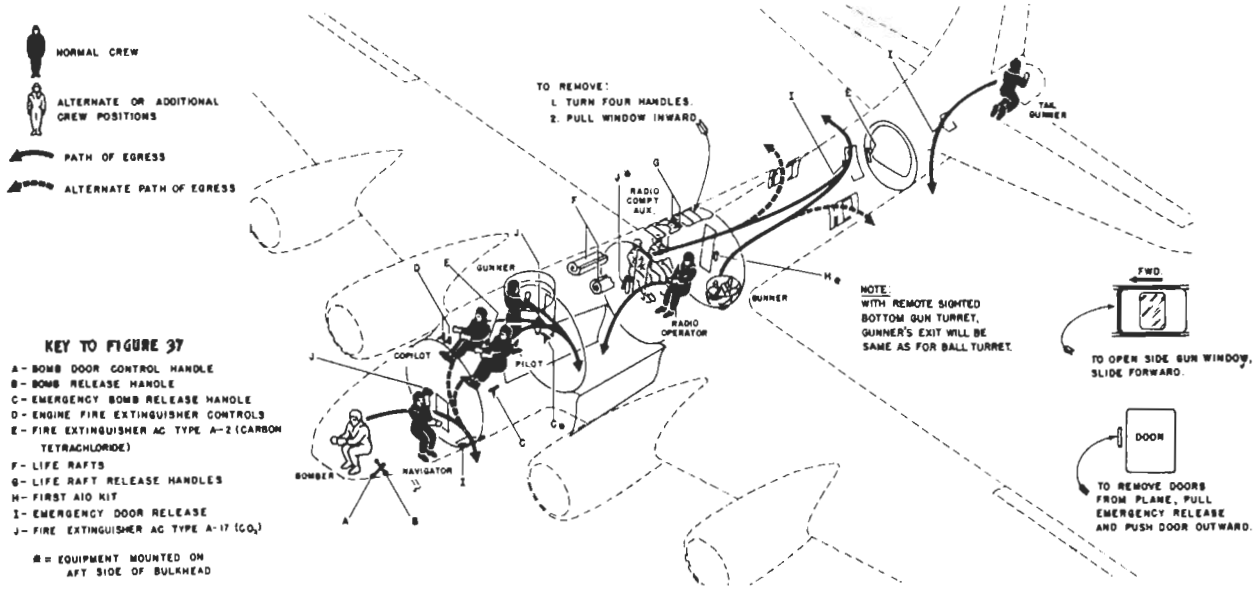
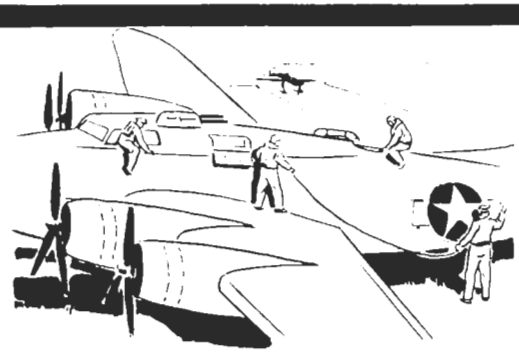


Figure 37 - Emergency Escape Routes

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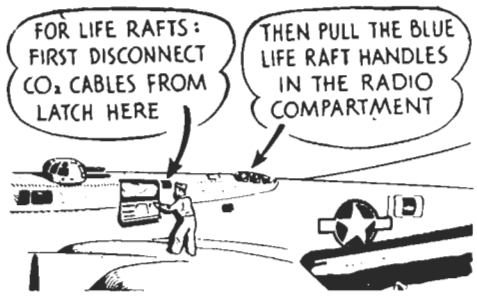
13. FORCED DESCENT AT SEA

1



As complete evacuation of the airplane should not take over 30 seconds, preflight practice drills should be participated in by all crews who are to make a flight over water, or whose operations are generally over water.

2



A complete and careful inspection of emergency equipment should be *made* before each long over water flight. Check life rafts, emergency kit bags (provisions), and emergency radio equipment. The kit bags and radio are stored aft of the radio compartment.

3

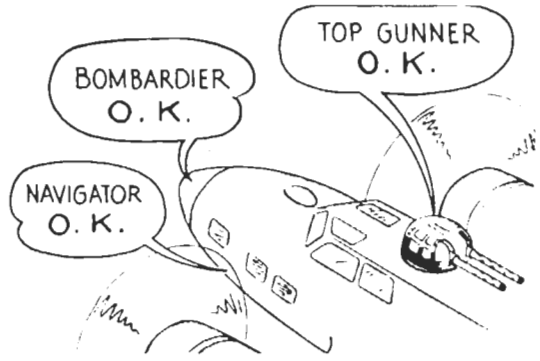


When it becomes evident that the airplane is to be forced down at sea due to lack of fuel, or that an altitude of at least 1,000 feet cannot be maintained, the pilot gives warning over the interphone.

WARNING!

This command *must*, if possible, be given while the fuel supply is still sufficient for 15 minutes of flight. The chances for a successful landing are much greater, if power is used.

4



Each crew member will acknowledge the command over the interphone.

5



The *bombardier* after acknowledging the command, will jettison bombs, or bomb bay tanks if more than half full, and close the bomb bay doors. If there is not sufficient time to release the bombs and close the bomb bay doors, ascertain that the bombs are "SAFE" and leave the doors closed.

6

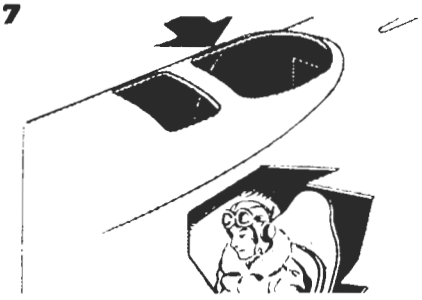


The *navigator* will determine the position and inform both the pilot and the radio operator. He will take with him the instruments necessary to make simple computation while on life rafts.

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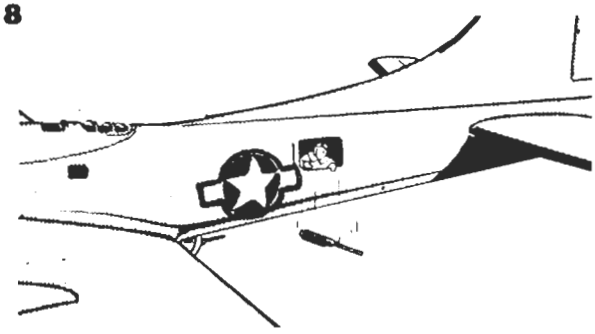
FORCED DESCENT AT SEA

7



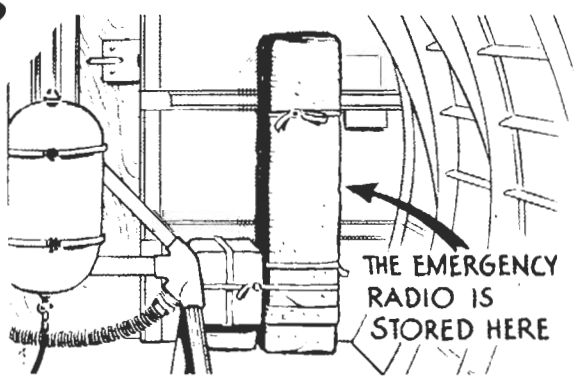
The *radio operator* will jettison the hatch cover. Then, when directed by the pilot, he will send an appropriate distress signal and position. After completing this duty, he will bring the emergency radio set into the radio compartment.

8



The *side gunners* will jettison the side guns as they make very dangerous battering rams. If there are no side gunners, this duty should be given to other crew members before flight.

9



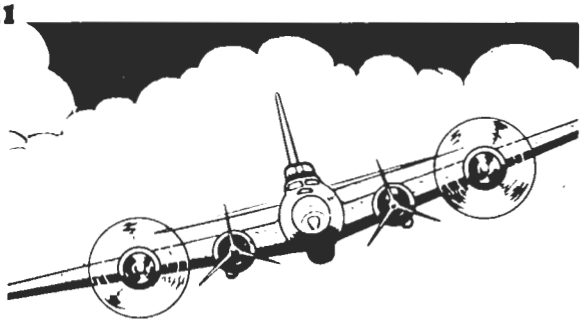
A crew member appointed before flight will take the emergency kit bags to the radio compartment.

10



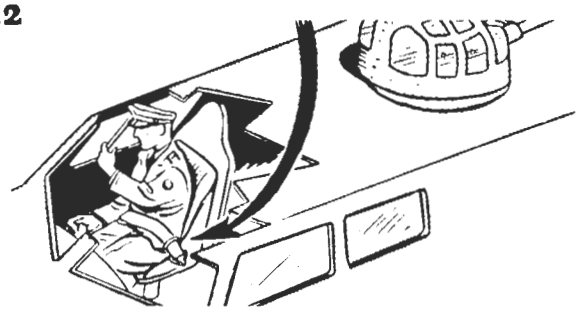
After completing his individual duties, each member goes to the radio compartment which is the crash station for all but the pilot and copilot.

11



The pilot will direct the copilot to cut the two inboard engines, if the two outboard engines are functioning satisfactorily, and to feather their propellers.

12

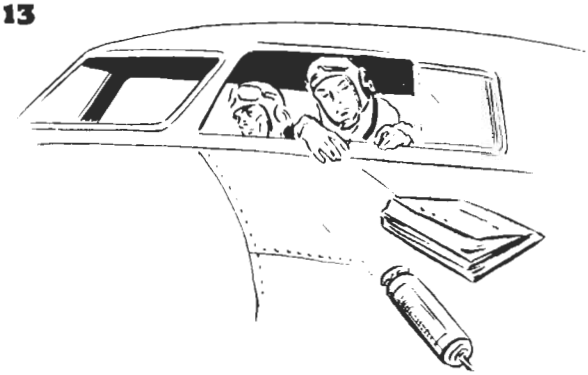


Both the pilot and the copilot will strap themselves in their seats. If the side windows are to be used as exits, slide windows open, then close, insuring freedom of operation. Leave them closed until after the impact. CAUTION! Place axe handy in event of jamming.

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FORCED DESCENT AT SEA

13



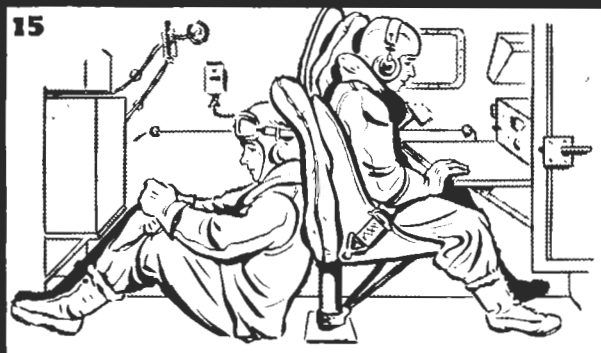
Be sure all emergency equipment is in the radio compartment. Throw overboard any equipment that might come loose.

14



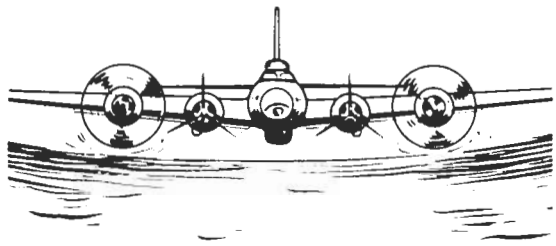
Remove cushions from seats for head protection and take crash positions. Do not take a position in the center of the compartment as ball turret upper structure makes this unsafe. Brace head against solid structure, if possible. Do not leave these positions until plane has come to rest as there will probably be more than one shock.

15



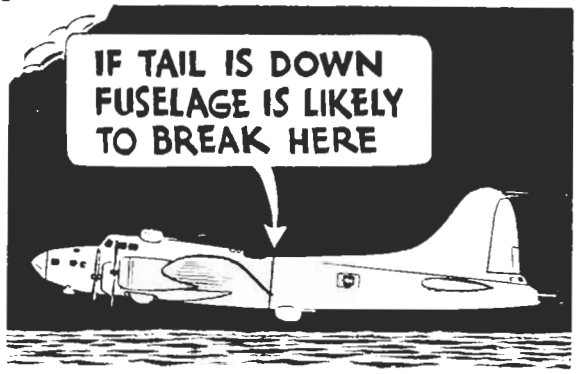
All members should have life vests on, parachutes removed, and should have on all extra clothing to be worn on rafts. At night, turn off all bright internal lights and use only the amber lamps.

16



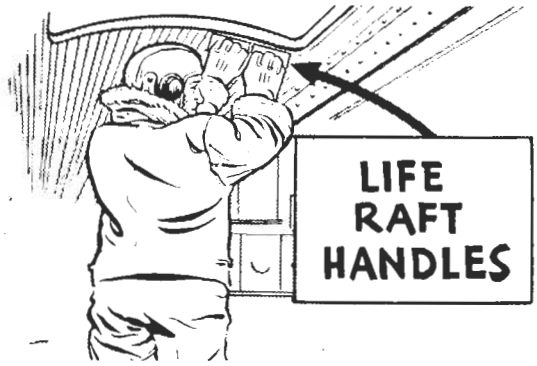
The pilot should attempt to set the airplane down in a trough, which is usually cross wind. The two outboard engines are used for control and to flatten the approach. The landing gear should be up, the flaps lowered medium, and the ignition switches cut a foot or so above the water.

17



The water should be touched at about 90 mph. Come in as level as possible.

18



As soon as the airplane has come to rest the predesignated member will pull the life raft handles.

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FORCED DESCENT AT SEA

19



During preflight drill, men should be assigned to evacuation duties. Each man should be familiar with these so that in case of accident alternate men can carry on. Each man should know his order.

20

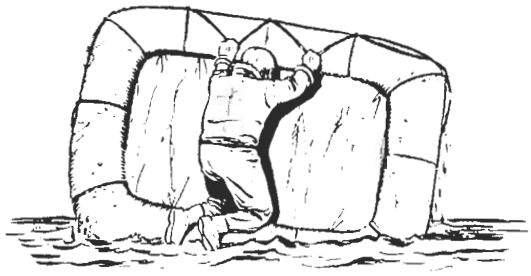


Pilot and copilot will exit through their side windows or through the radio compartment hatch. Decide which before flight.

CAUTION!

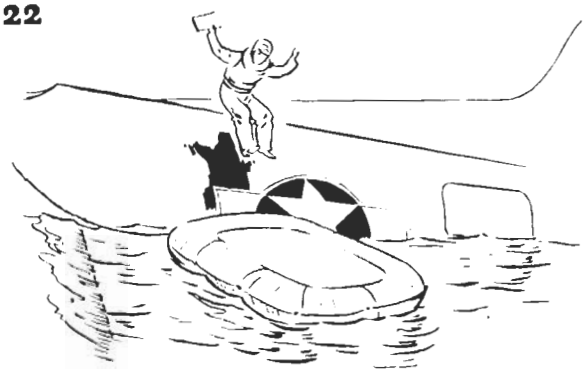
No crew member should inflate his life vest until he has emerged from the airplane.

21



If the life raft is inflated upside down, one man should jump into the water and right it. If there are handling patches on bottom of raft, grasp them with both hands, and with knees on bouyancy chamber, lean back and prepare to be submerged for a moment. Even the largest raft will turn over.

22



WARNING!

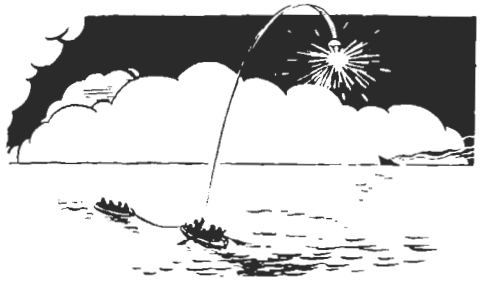
Do not jump on an inverted raft, as this will expel the air trapped under it and righting becomes more difficult.

23



The rafts should be fastened together so they will not drift apart. Once aboard the rafts a check should be made to locate leaks. Repair them with the kit provided in the raft. Keep away from the airplane, if it floats but stay in the vicinity if possible. Do not remove wet clothing. Do not talk more than necessary; it dries the mouth. Do not move more than necessary; it takes energy.

24



A signal kit containing a pistol and flares is in a waterproof sealed pocket of the life raft. It may be advisable to leave the kit sealed in the pocket until a ship or a plane is sighted so as to have dry signal equipment.

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14. EMERGENCY OPERATION OF RADIO EQUIPMENT.

a. PORTABLE EMERGENCY RADIO TRANSMITTER (Type SCR-578-A).

(1) GENERAL.

(a) A complete self-contained portable emergency transmitter is stowed on the right rear side of bulkhead 6, and is provided for operation anywhere away from the airplane. It is primarily designed for use in a small boat or life raft, but it may be placed in operation anywhere a kite can be flown or where water may be found.

(b) When operated, the transmitter emits an MCW signal and is pretuned to the international distress frequency of 500 kilocycles. Automatic transmission of a predetermined signal is provided. Any searching party can "home" on the signal with the aid of a radio compass.

(c) No receiver is provided.

(2) REMOVAL FROM AIRPLANE.

(a) If the airplane has made an emergency landing on water, the emergency set should be removed at the same time that the life raft is removed. The set is waterproof and will float, and it is not necessary to take any precautions in keeping the equipment out of the water; however, be sure that it does not float out of reach.

(b) The emergency set may be dropped from the airplane by use of the parachute attached. The altitude of the airplane when dropping the equipment should be between 300 and 500 feet. To drop the equipment, the following steps should be observed:

1. Tie the loose end of the parachute static line to any solid metal structure of the airplane.

CAUTION

Be sure that the static line is in the clear and will not foul.

2. Throw the emergency set out through a convenient opening in the airplane. Parachute will be opened by the static line.

CAUTION

Do not attach static line to any part of one's clothing or body when throwing the equipment through the opening.

(3) OPERATION. - Complete operating instructions are contained in one of the bags which contain the equipment. Complete instructions for the use of the transmitter are also located on the transmitter itself.

b. INTERPHONE EQUIPMENT FAILURE. - In the event of interphone equipment failure, the audio frequency section of the command transmitter may be substituted for the regular interphone amplifier. To make this connection, the pilot should place his command transmitter control box channel selector switch in either channel No. 3 or 4 position. Set the interphone jack-box selector switch on the "COMMAND" to place the interphone equipment in operation.

NOTE

When the command transmitter control box channel selector switch is set in either the No. 3 or 4 position for emergency operation of the interphone equipment, it is not possible to establish communication with any station or any other airplane. It is possible at all times to resume normal command set operation by placing the channel selector switch of the command transmitter control box in either the No. 1 or 2 position.

c. SUBSTITUTION OF RADIO COMPASS RECEIVER FOR LOW FREQUENCY COMMAND SET RECEIVER. - If the low frequency receiver of the command set fails, the radio compass receiver may be substituted, with the pilot having direct control over the compass receiver. To complete this emergency hook-up, the pilot must set his interphone jack-box selector switch in the "COMP" position and then place the radio compass selector switch in the " ANT" position. The radio compass can then be tuned as desired.

d. SUBSTITUTION OF LIAISON RECEIVER FOR LOW, MEDIUM, AND/OR HIGH FREQUENCY COMMAND RECEIVER. - In case of the failure of the low, medium, and/or high frequency receiver of the command radio equipment, the liaison receiver may be substituted, but the pilot will have only limited control over it. The pilot should first call the radio operator on the interphone system and tell him what frequency he desires to receive, that he is switching the interphone selector switch to the "LIAISON" position, and for him (the radio operator) to tune in this frequency and maintain the setting until further advised.

e. COMMAND SET TRANSMITTER FAILURE. - In case of failure of the command set transmitter, the liaison transmitter may be substituted. The pilot should first call the radio operator on the interphone and have him adjust the liaison transmitter to the frequency he desires to use. He should then set his interphone selector switch to the "LIAISON" position and operate his microphone button in the same manner that he did when the command set was in operation. When he is through using the liaison transmitter, the pilot should place the interphone selector switch in the "INTER" position and tell the radio operator to cut the liaison transmitter off, so as to reduce the load on the electrical system.

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NOTE

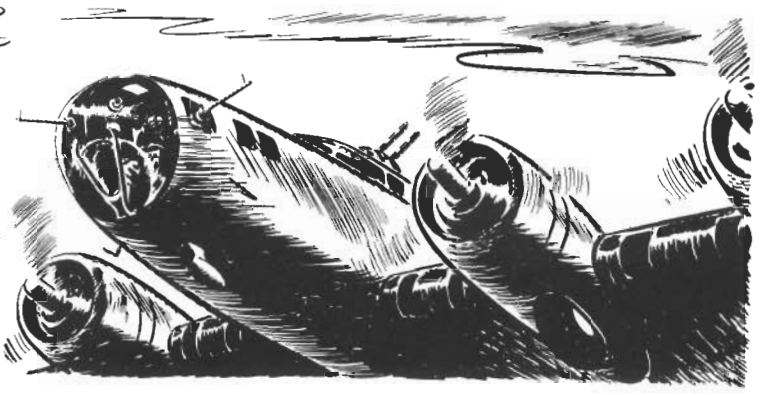
When substituting one receiver for another, such as the compass receiver for the command receiver, the pilot must move his interphone selector switch to the "COMMAND" or "LIAISON" position, as the case may be, in order to transmit. At the end of the transmission, he must switch back to the position of the receiver being used. This will have to be done every time that the pilot desires to hold a two-way conversation.

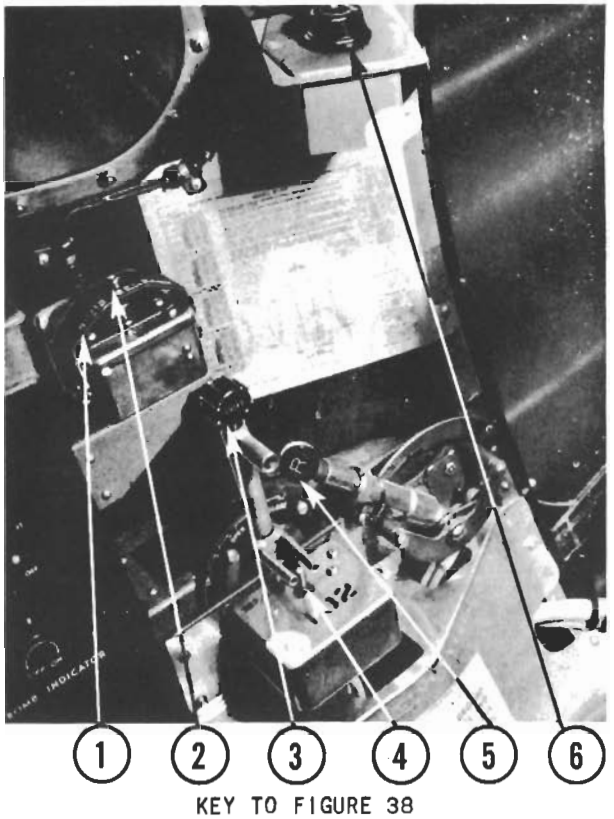
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SECTION IV
BOMBARDIER'S COMPARTMENT





KEY TO FIGURE 38

1. BOMB RELEASE SWITCH GUARD
2. BOMB RELEASE SWITCH
3. BOMB DOOR CONTROL HANDLE
4. BOMB DOOR SWITCH
5. BOMB RELEASE HANDLE
6. BOMBARDIER'S LIGHT SWITCH

Figure 38 - Bomb Controls

1. BOMB CONTROLS.

a. Bombs are normally released electrically, but can be released mechanically in an emergency. Electrical control provides for individual release of bombs either singly (selective) or continuously at predetermined intervals (train). Mechanical control is always in "SALVO," by operation of the bombardier's release handle or by operation of the emergency release handles. The bomb release handle has three positions.

(1) In the "LOCK" position the bomb racks are locked against any release of bombs except by means of the emergency release handles.

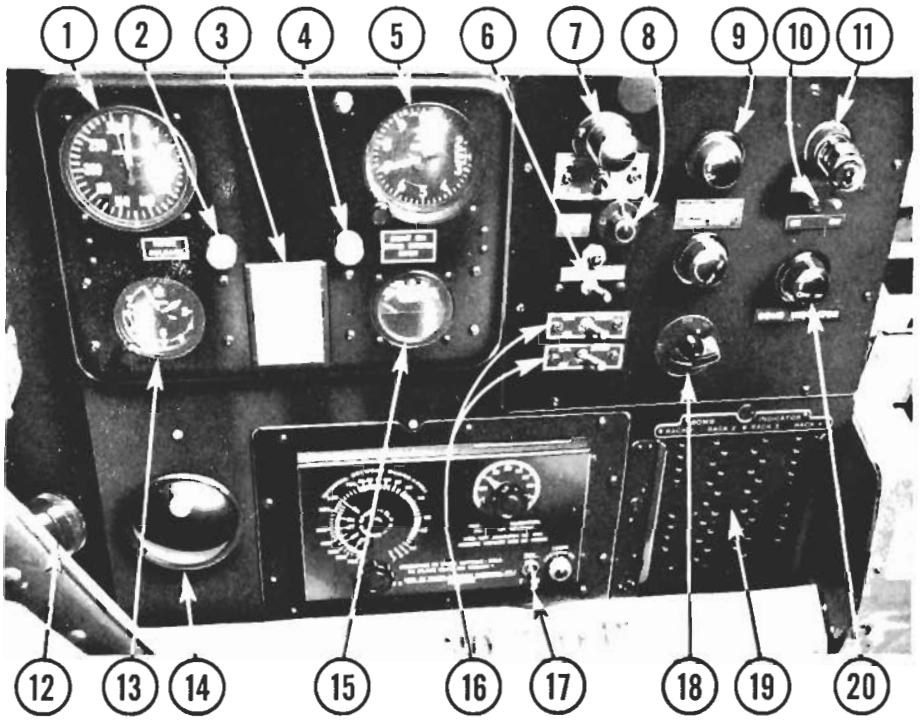
(2) In the "SELECTIVE" position the bomb racks are prepared for electrical release by manual operation of the release switch, or by automatic operation through the bomb sight.

(3) The "SALVO" position, when the bomb doors are open, mechanically releases all bombs simultaneously and unarmed.

b. The bombardier's release switch, mounted on the forward end of the control panel, operates in either direction to energize the release unit solenoids through the interval release control mechanism. A hinged guard prevents accidental operation of this switch.

c. The interval release control unit is mounted at the bottom of the bombardier's control panel and may be set to provide either "SELECT" or "TRAIN" release. On airplanes serial Nos. 42-5050 and on, four switches on the bombardier's control panel permit selection of any external or internal rack for electrical release. Two indicator lamps beside the rack selector switches correspond to the external racks. Two additional rack selector switches in the bomb bay permit elemination of either right or left bomb bay from the release circuit if bomb bay fuel tanks are carried. Bomb release sequence is given in figure 40. Any rack or combination of racks may be eliminated from the release sequence by turning off

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KEY TO FIGURE 39

1. AIR SPEED INDICATOR
2. BOMB RELEASE WARNING LAMP
3. ALTIMETER SCALE ERROR CARD
4. BOMB DOOR WARNING LAMP
5. ALTIMETER
6. PILOT CALL SWITCH
7. PANEL LIGHT
8. PHONE CALL LAMP
9. WARNING LAMP RHEOSTAT
10. EXTENSION LIGHT SWITCH
11. EXTENSION LIGHT
12. ULTRA-VIOLET SPOT LIGHT
13. CLOCK
14. ASH RECEIVER
15. FREE AIR THERMOMETER
16. BOMB RACK SELECTOR SWITCHES
17. BOMB INTERVAL SWITCH
18. ULTRA-VIOLET SPOTLIGHT CONTROL SWITCH
19. BOMB INDICATOR
20. BOMB INDICATOR CONTROL KNOB

Figure 39 - Bombardier's Control Panel

the respective selector switch on the bombardier's control panel.

d. A bomb arming solenoid in each external rack is controlled by a switch on the bombardier's panel. A red indicator lamp beside the switch is on when the bombs are armed.

NOTE

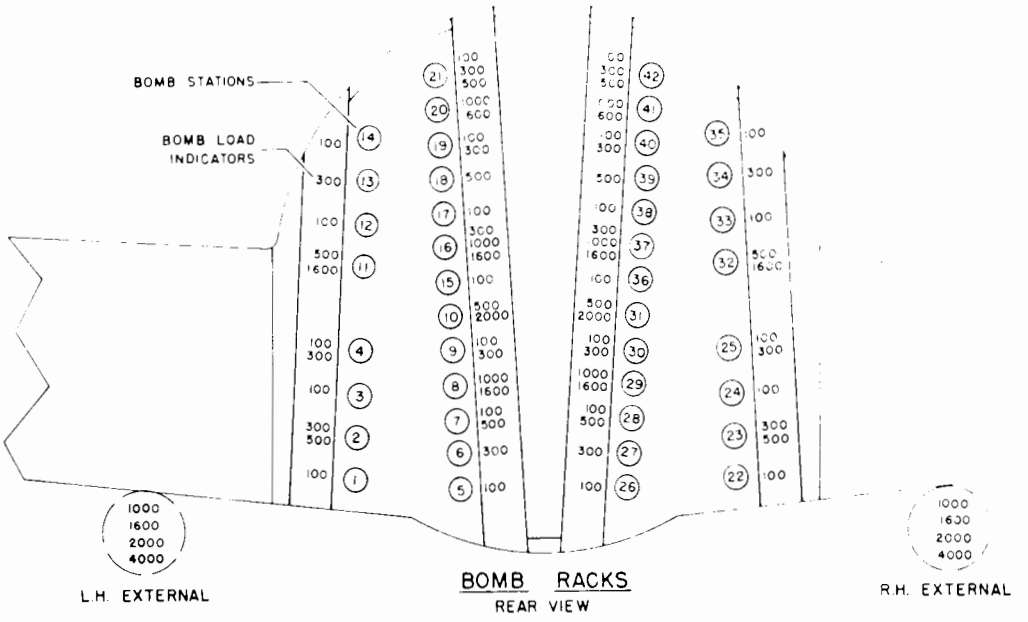
Some B-17F airplanes not equipped for external racks have only two rack selector switches and no bomb arming switch on the bombardier's panel. A few airplanes have no rack selector switches on the bombardier's panel but have a three-position switch in the bomb bay to turn off either internal rack.

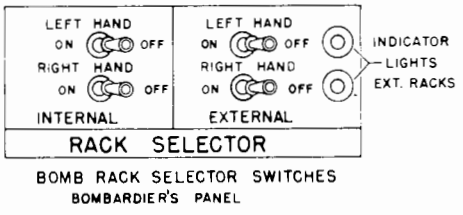
e. The bomb door control handle is at the left of the bombardier, forward of the control panel, and operates a double-throw toggle switch controlling the solenoid switches for the bomb door retracting motor. A lug on the side of the handle is located so that when the door handle is in the "CLOSED" position, the bomb release lever cannot be moved out of the "LOCK" position.

CAUTION

If bombs are carried above the 2000-pound bomb, they MUST NOT be released until the D-6 shackle and adapter have been removed. This definitely requires "SELECTIVE" release control for the 2000-pound bomb.

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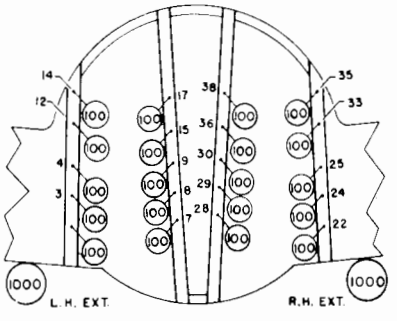
BOMB RACK SELECTOR SWITCHES
BOMBARDIER'S PANEL

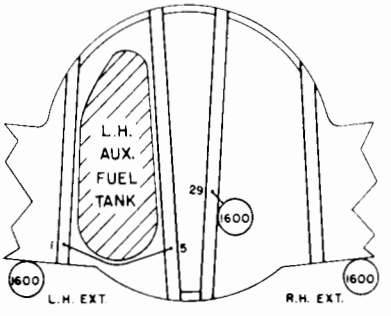
NOTE: ANY UNLOADED OR UNCOCKED STATION WILL BE AUTOMATICALLY SKIPPED IN THE RELEASE SEQUENCE AND THE REMAINING STATIONS RELEASED ACCORDING TO THE NORMAL SEQUENCE.

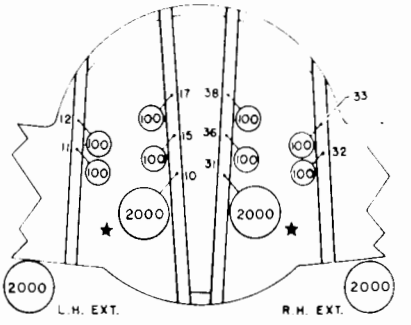
BOMB BAY DOORS MUST BE OPEN TO DROP ANY BOMBS

CAUTION: CLOSING THE BOMB DOORS, RETURNING THE BOMB RELEASE LEVER TO LOCK, OR TURNING OFF ANY RACK SELECTOR SWITCH MAY CAUSE THE RELEASE SEQUENCE TO RESUME AT SOME STATION OTHER THAN THE NEXT IN THE NORMAL SEQUENCE DUE TO THE ACTION OF THE RACK SELECTOR RELAYS.

EXAMPLES







NO. I

INTERNAL RACKS ON EXTERNAL RACKS ON			
SEQ.	STA.	SEQ.	STA.
1	1	12	29
2	R.H. EXT.	13	9
3	22	14	30
4	L.H. EXT.	15	12
5	3	16	33
6	24	17	14
7	4	18	35
8	25	19	15
9	7	20	36
10	28	21	17
11	8	22	38

NO. 4

L.H. INT. OFF R.H. INT. ON EXT. RACKS ON	
SEQ.	STA.
1	R.H. EXT.
2	L.H. EXT.
3	29

I NO. II

INT. RACKS OFF EXT. RACKS ON	
SEQ.	STA.
1	R.H. EXT.
2	L.H. EXT.

II RELEASE TWO 2000 LB BOMBS SELECT OR TRAIN OF 2.

★ LARGE TYPE D-6 SHACKLE MUST BE REMOVED BEFORE RELEASING BOMBS ABOVE.

III NO 2

INT. RACKS ON EXT. RACKS OFF	
SEQ.	STA
1	11
2	32
3	12
4	33
5	15
6	36
7	17
8	38

Figure 40 - Bomb Release Sequence Diagram (Sheet 1)

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ANY BOMB LOAD WILL BE RELEASED ACCORDING TO ONE OF THESE SEQUENCES. COMBINATIONS OF RELEASE SEQUENCES FOR A PARTICULAR BOMB LOAD ARE POSSIBLE BY OPERATION OF THE RACK SELECTOR SWITCHES BETWEEN "STICKS." (SEE CAUTION ON SHEET NO.1)

NO.1

INTERNAL RACKS ON
EXTERNAL RACKS ON

Sequence	Bomb Sta.	Sequence	Bomb Sta.
1	1	23	11
2	R.H. Ext.	24	32
3	22	25	12
4	L.H. Ext.	26	33
5	2	27	13
6	23	28	34
7	3	29	14
8	24	30	35
9	4	31	15
10	25	32	36
11	5	33	16
12	26	34	37
13	6	35	17
14	27	36	38
15	7	37	18
16	28	38	39
17	8	39	19
18	29	40	40
19	9	41	20
20	30	42	41
21	10	43	21
22	31	44	42

NO.2

INTERNAL RACKS ON
EXTERNAL RACKS OFF

Sequence	Bomb Sta.	Sequence	Bomb Sta.
1	1	22	32
2	22	23	12
3	2	24	33
4	23	25	13
5	3	26	34
6	24	27	14
7	4	28	35
8	25	29	15
9	5	30	36
10	26	31	16
11	6	32	37
12	27	33	17
13	7	34	38
14	28	35	18
15	8	36	39
16	29	37	19
17	9	38	40
18	30	39	20
19	10	40	41
20	31	41	21
21	11	42	42

NO.3

L.H. INT. ON
R.H. INT. OFF
EXT. RACKS ON

Sequence	Bomb Sta.
1	1
2	R.H. Ext.
3	L.H. Ext.
4	2
5	3
6	4
7	5
8	6
9	7
10	8
11	9
12	10
13	11
14	12
15	13
16	14
17	15
18	16
19	17
20	18
21	19
22	20
23	21

NO.4

L.H. INT. OFF
R.H. INT. ON
EXT. RACKS ON

Sequence	Bomb Sta.
1	R.H. Ext.
2	22
3	L.H. Ext.
4	23
5	24
6	25
7	26
8	27
9	28
10	29
11	30
12	31
13	32
14	33
15	34
16	35
17	36
18	37
19	38
20	39
21	40
22	41
23	42

NO.5

BOTH INTERNAL RACKS ON
L.H. EXTERNAL RACK OFF
R.H. EXTERNAL RACK ON

Sequence	Bomb Sta.	Sequence	Bomb Sta.
1	1	23	32
2	R.H. Ext.	24	12
3	22	25	33
4	2	26	13
5	23	27	34
6	3	28	14
7	24	29	35
8	4	30	15
9	25	31	36
10	5	32	16
11	26	33	37
12	6	34	17
13	27	35	38
14	7	36	18
15	28	37	39
16	8	38	19
17	29	39	40
18	9	40	20
19	30	41	41
20	10	42	21
21	31	43	42
22	11		

NO.6

BOTH INTERNAL RACKS ON
L.H. EXTERNAL RACK ON
R.H. EXTERNAL RACK OFF

Sequence	Bomb Sta.	Sequence	Bomb Sta.
1	1	23	32
2	22	24	12
3	L.H. Ext.	25	33
4	2	26	13
5	23	27	34
6	3	28	14
7	24	29	35
8	4	30	15
9	25	31	36
10	5	32	16
11	26	33	37
12	6	34	17
13	27	35	38
14	7	36	18
15	28	37	39
16	8	38	19
17	29	39	40
18	9	40	20
19	30	41	41
20	10	42	21
21	31	43	42
22	11		

NO.7

L.H. INT. ON
R.H. INT. OFF
EXT. RACKS OFF

Sequence	Bomb Sta.
1	1
2	2
3	3
4	4
5	5
6	6
7	7
8	8
9	9
10	10
11	11
12	12
13	13
14	14
15	15
16	16
17	17
18	18
19	19
20	20
21	21

NO.8

L.H. INT. OFF
R.H. INT. ON
EXT. RACKS OFF

Sequence	Bomb Sta.
1	22
2	23
3	24
4	25
5	26
6	27
7	28
8	29
9	30
10	31
11	32
12	33
13	34
14	35
15	36
16	37
17	38
18	39
19	40
20	41
21	42

NO.9

L.H. INT. ON
L.H. EXT. ON
OTHERS OFF

Sequence	Bomb Sta.
1	1
2	L.H. Ext.
3	2
4	3
5	4
6	5
7	6
8	7
9	8
10	9
11	10
12	11
13	12
14	13
15	14
16	15
17	15
18	17
19	18
20	19
21	20
22	21

NO.10

R.H. INT. ON
R.H. EXT. ON
OTHERS OFF

Sequence	Bomb Sta.
1	R.H. Ext.
2	22
3	23
4	24
5	25
6	26
7	27
8	28
9	29
10	30
11	31
12	32
13	33
14	34
15	35
16	36
17	37
18	38
19	39
20	40
21	41
22	42

NO.11

INT. RACKS OFF
EXT. RACKS ON

Sequence	Bomb Sta.
1	R.H. Ext.
2	L.H. Ext.

NO.12

INT. RACKS OFF
L.H. EXT. ON
R.H. EXT. OFF

Sequence	Bomb Sta.
1	L.H. Ext.

NO.13

INT. RACKS OFF
L.H. EXT. OFF
R.H. EXT. ON

Sequence	Bomb Sta.
1	R.H. Ext.

NO.14

L.H. INT. ON
R.H. EXT. ON
OTHERS OFF

Sequence	Bomb Sta.
1	1
2	R.H. Ext.
3	2
4	3
5	4
6	5
7	6
8	7
9	8
10	9
11	10
12	11
13	12
14	13
15	14
16	15
17	16
18	17
19	18
20	19
21	20
22	21

NO.15

R.H. INT. ON
L.H. EXT. ON
OTHERS OFF

Sequence	Bomb Sta.
1	22
2	L.H. Ext.
3	23
4	24
5	25
6	26
7	27
8	28
9	29
10	30
11	31
12	32
13	33
14	34
15	35
16	36
17	37
18	38
19	39
20	40
21	41
22	42

Figure 40 - Bomb Release Sequence Diagram (Sheet 2)

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MAXIMUM AIRPLANE GLIDE & CLIMB ANGLES FOR BOMB RELEASE

WITH WHEELS AND FLAPS UP: MAXIMUM ALLOWABLE INDICATED AIR SPEED I° 270 MPH SAFE GLIDE ANGLE IS 15-1/4°.

WITH WHEELS AND FLAPS DOWN: MAXIMUM ALLOWABLE INDICATED AIR SPEED IS 147 MPH SAFE GLIDE ANGLE IS 13-1/2°.

NOTE: THE SAFE GLIDE ANGLES ARE BASED ON AN AIRPLANE GROSS WEIGHT OF 40,000 LBS WITH POWER OFF AND WINDMILLING PROPELLERS.

WHILE THE MAJORITY OF BOMB STATIONS WILL PERMIT RELEASE OF BOMBS AT AN ANGLE WHICH WILL PRODUCE AN INDICATED AIR SPEED GREATER THAN THAT DESIGNATED FOR THE SAFE GLIDE ANGLE OF THE AIRPLANE, UNDER NO CONDITIONS SHALL THE MAXIMUM ALLOWABLE INDICATED AIR SPEED BE EXCEEDED.

ANGLES SHOWN ALLOW 10° FOR SAFETY. HOWEVER, UNDER PERFECTLY SMOOTH FLYING CONDITIONS, IF IN THE AIRPLANE COMMANDER'S OPINION CONDITIONS WARRANT IT, THESE GIVEN ANGLES MAY BE EXCEEDED BY NOT MORE THAN 5°.

THE GLIDE OR CLIMB ANGLE IS THE ANGLE INCLUDED BETWEEN THE EARTH'S SURFACE AND THE FUSELAGE CENTERLINE.

THE ANGLES LISTED IN THE TABULATION ARE THE MAXIMUM AT WHICH BOMBS MAY BE RELEASED WITH A 10° CLEARANCE ANGLE MAINTAINED IN THE BOMB BAY.

1100 LB. M-33

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
2 & 3	29 & 8	26	15
	37 & 16	11	6 1/2
	41 & 20	5	2

300 LB. MK.I - MK.IMI

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	28 & 23	37	33 3/4
	48 & 25	23 3/4	22
	13 & 34	14 3/4	15
2 & 3	27 & 6	44 1/2	40
	30 & 9	27	25
	37 & 16	17 1/4	16 1/4
	40 & 19	11 1/2	11 1/4
	42 & 21	8	8

100 LB. M-38A2

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	1 & 22	49 3/4	44 1/2
	3 & 24	40	32
	4 & 25	29 1/2	26 3/4
	12 & 33	23	20 3/4
	14 & 35	20	15
2 & 3	26 & 5	57 1/2	52
	28 & 7	44 1/2	39 3/4
	30 & 9	33	29 1/2
	36 & 15	25	22 1/2
	38 & 17	19 3/4	18
	40 & 19	15 1/2	14 1/4
	42 & 21	11 1/2	10 1/2

100 LB. M-30

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	1 & 22	47 3/4	51
	3 & 24	36 1/2	41
	4 & 25	28 1/4	33 1/2
	12 & 33	22	27 1/2
	14 & 35	17 1/2	22 3/4
2 & 3	26 & 5	56	57 1/2
	28 & 7	42 1/2	46 1/2
	30 & 9	31 1/2	36 1/2
	36 & 15	23 3/4	29 1/4
	38 & 17	19	24
	40 & 19	15	20
	42 & 21	11 1/4	15 3/4

2000 LB. M-34

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
2 & 3	31 & 10	0	0

600 LB. M-32

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	28 & 23	32 1/2	29
2 & 3	28 & 7	34 1/2	29 1/2
	31 & 10	18	17 1/2
	39 & 18	10	10
	42 & 21	5 1/2	6

600 LB. MK.IMI - MK.IMII

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
2 & 3	28 & 7	33	23
	31 & 10	18	12 1/2
	39 & 18	9 1/2	6 1/2
	42 & 21	5	2 1/2

300 LB. M-31

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	28 & 23	38	38 1/2
	48 & 25	24	26 1/2
	13 & 34	16	18 3/4
2 & 3	27 & 6	45	44 3/4
	30 & 9	27 1/4	29 1/2
	37 & 16	17 1/2	20
	40 & 19	11 3/4	14 1/2
	42 & 21	8 1/4	10 1/2

100 LB. MK.I-MK.IMI MK.IMII

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	1 & 22	46 1/4	45
	3 & 24	34 1/2	34 3/4
	4 & 25	26 1/4	27
	12 & 33	20 1/2	21 1/2
	14 & 35	16	16 3/4
2 & 3	26 & 5	54 1/2	52 1/2
	28 & 7	40 3/4	40 1/4
	30 & 9	29 3/4	30
	36 & 15	22	23
	38 & 17	17 1/4	19 1/4
	40 & 19	13 1/2	14 1/2
	42 & 21	9 3/4	10 3/4

500 LB. M-43

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	28 & 23	33	33 1/4
	11 & 32	17	19 1/4
2 & 3	28 & 7	34 1/4	34
	31 & 10	18 3/4	21
	39 & 18	10	12 1/2
	42 & 21	5 1/2	8

1100 LB. MK. III

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
2 & 3	29 & 8	23 1/2	9
	37 & 16	10	1 1/2
	41 & 20	4	0

1600 LB. AN-MKI

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	11 & 32	7	1 1/2
2 & 3	8 & 29	16 1/2	6 1/2
	16 & 37	4 1/2	0
	20 & 41	0	0

1000 LB. M-44

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
2 & 3	29 & 8	25	17
	37 & 16	11	8
	41 & 20	5	3

100 LB. M-39

RACK NO.	STA.	GLIDE ANGLE	CLIMB ANGLE
1 & 4	1 & 22	46 1/4	45
	3 & 24	34 1/2	34 3/4
	4 & 25	26 1/4	27
	12 & 33	20 1/2	21 1/2
	14 & 35	16	16 3/4
2 & 3	26 & 5	54 1/2	52 1/2
	28 & 7	40 3/4	40 1/4
	30 & 9	29 3/4	30
	36 & 15	22	23
	38 & 17	17 1/4	19 1/4
	40 & 19	13 1/2	14 1/2
	42 & 21	10	10 3/4

Figure 41 - Bomb Release Angles Chart

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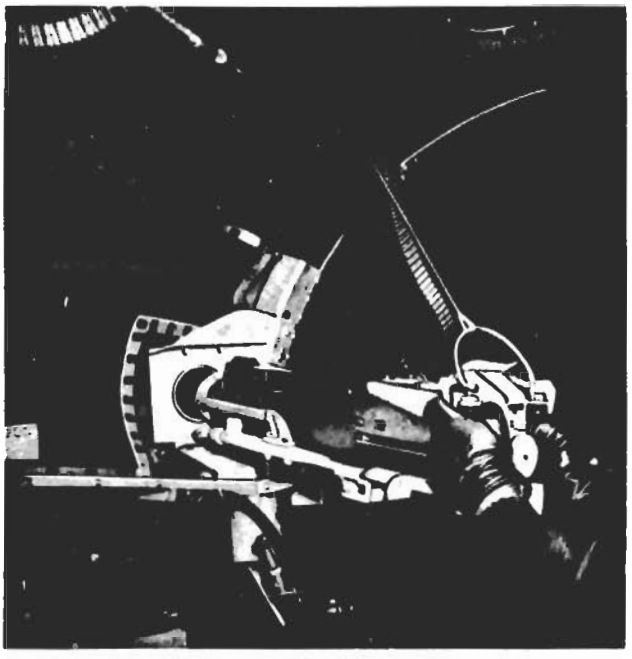
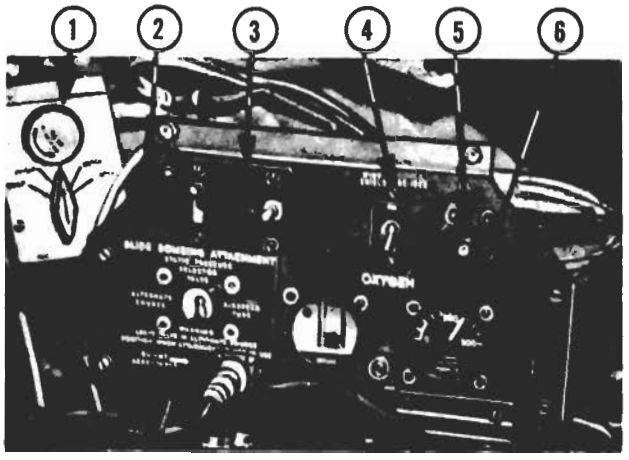


Figure 42 - Bombardier's Gun - Left Side



KEY TO FIGURE 43

1. INTERPHONE JACKBOX
2. GLIDE BOMBING ATTACHMENT STATIC PRESSURE SELECTOR SWITCH
3. WINDSHIELD WIPER CONTROLS
4. WINDSHIELD ANTI-ICER PUMP SWITCH
5. ANTI-ICER ALCOHOL FLOW VALVE
6. OXYGEN INDICATORS

Figure 43 - Bombardier's Compartment - Right Side

2. BOMBARDIER'S GUNS.

a. Most airplanes have two .50-caliber machine gun installations, one mounted through a window on either side of the bombardier's compartment. A .50-caliber gun is also mounted in the center Plexiglas nose of some airplanes. In some airplanes ball and socket mounts are incorporated in the nose, side, and top windows for insertion of a .30-caliber machine gun.

b. On B-17G airplanes a type A-16 chin turret with two .50 calibre machine guns is mounted below, and is remotely controlled from, the bombardier's compartment.

3. INTERPHONE.

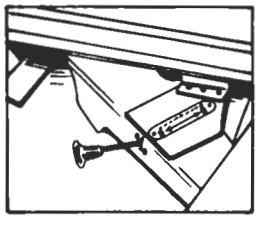
Two interphone jack boxes are on the right side of the compartment. Operating instructions are given in section I, paragraph 10.

4. OXYGEN.

The oxygen regulator and indicator panel are on the right wall of the compartment. Operating instructions are given in section I, paragraph 9.

5. BOMB-SIGHT WINDOW DEFROSTER.

A control knob in the floor in front of the bombardier's seat controls the flow of air to the bomb-sight window. Push forward to shut off the flow of air; pull aft to allow air to reach the bomb-sight window. Selection of hot and cold air is made by the pilot.



6. WINDSHIELD WIPER AND ANTI-ICER.

Anti-icer and wiper controls for the bomb-sight window are on a panel at the bombardier's right.

a. A toggle switch regulates the wiper motor "OFF," "SLOW," or "FAST." A circuit breaker protects the circuit in case of an overload.

b. An "ON-OFF" switch controls the alcohol and flow is regulated by a needle valve.

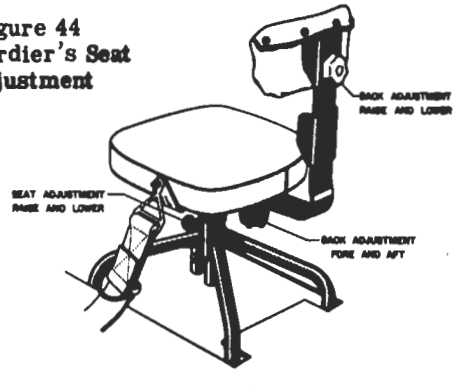
CAUTION

Do not operate the wiper on dry glass.

7. BOMB-SIGHT HEATING PAD.

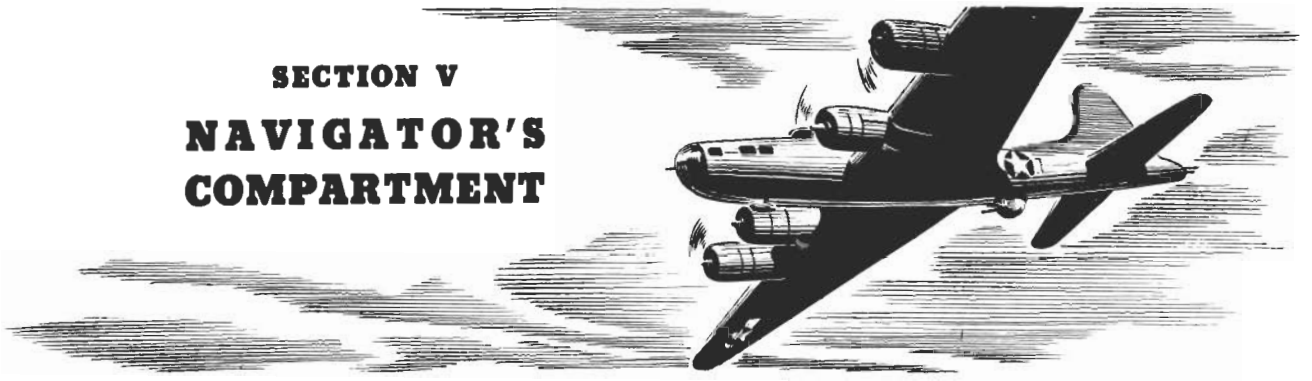
Most airplanes are equipped with an electrical bomb-sight heating pad which may be plugged into the bombardier's suit heater receptacle.

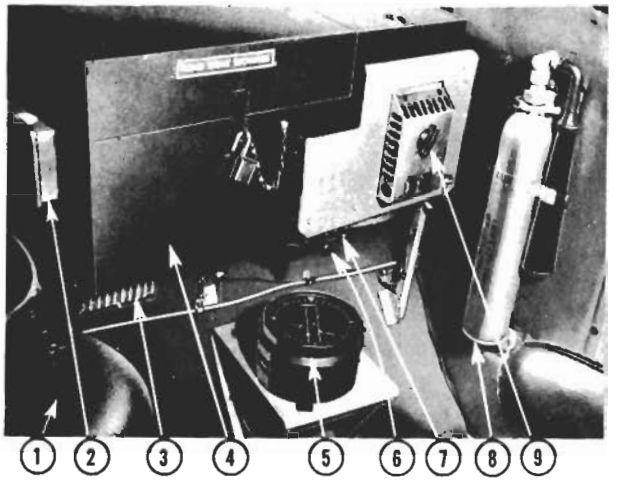
Figure 44
Bombardier's Seat Adjustment



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SECTION V
NAVIGATOR'S COMPARTMENT





KEY TO FIGURE 45

1. DRIFT METER
2. FUSE BOX
3. HEATING AND VENTILATING OUTLET
4. BOMB SIGHT STOWAGE BOX
5. APERIODIC COMPASS
6. PANEL LIGHT
7. PANEL LIGHT SWITCH
8. FIRE EXTINGUISHER
9. SUIT HEATER OUTLET

Figure 45 - Navigator's Compartment Right Rear Corner

1. LIGHTING.

A dome light and switch are in the ceiling of the compartment. A panel light and switch are above the navigator's table on the aft wall. The navigator's light is on the wall directly over his table; the switch is on the base of the lamp.

2. FIRE EXTINGUISHER.

A hand CO_2 fire extinguisher is clipped to the aft wall of the compartment to the right of the door.

3. INTERPHONE.

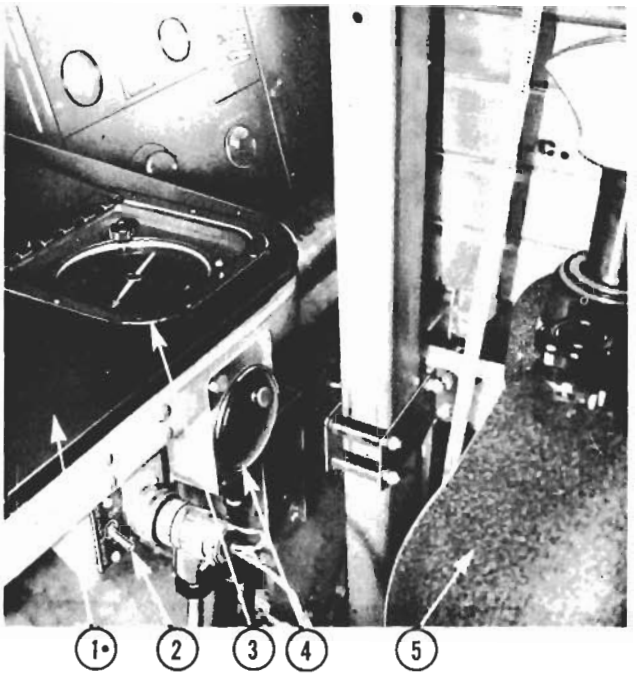
The interphone jack box is between the radio compass control box and the map case. Operating instructions are given in section I, paragraph 10.

4. OXYGEN.

The oxygen regulator is on the wall above the navigator's table. Refer to section I, paragraph 9.

5. HEATING AND VENTILATING INLET.

The inlet beneath the bomb-sight storage box is equipped with a push-pull knob for regulating the flow

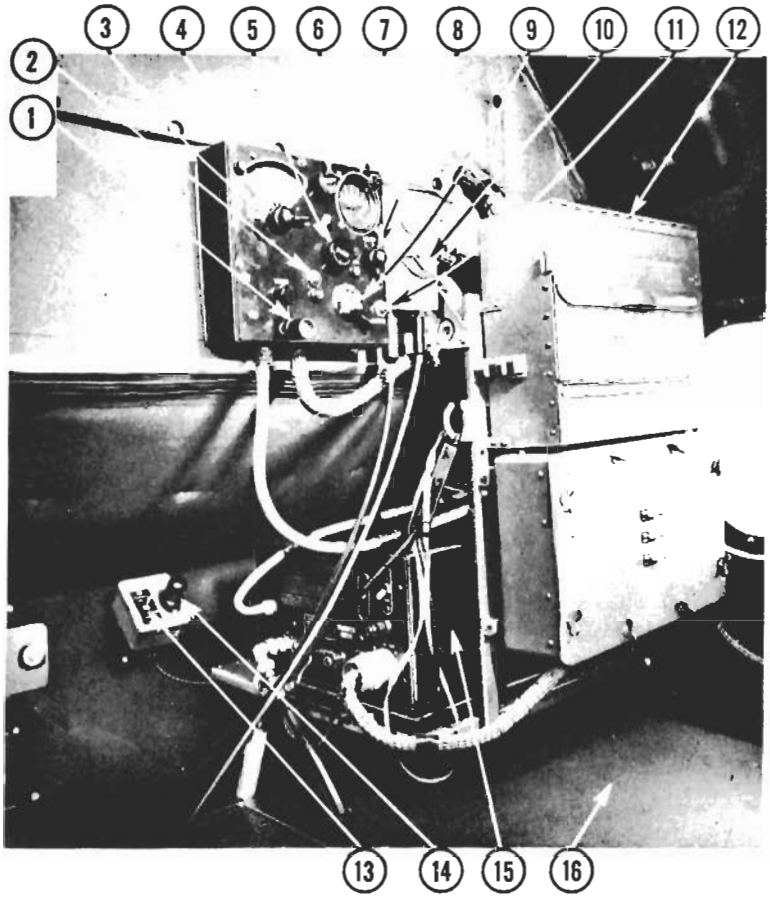


KEY TO FIGURE 46

1. NAVIGATOR'S TABLE
2. DRIFT METER MASTER SWITCH
3. RADIO COMPASS INDICATOR
4. ASH RECEIVER
5. DRIFT METER

Figure 46 - Navigator's Equipment

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KEY TO
FIGURE 47

1. TUNING CRANK
2. CONTROL INDICATOR LAMP
3. BAND SELECTOR SWITCH
4. RADIO COMPASS CONTROL UNIT
5. VOLUME CONTROL
6. LIGHT CONTROL SWITCH
7. TUNING METER
8. LOOP CONTROL SWITCH
9. RADIO COMPASS POWER SWITCH
10. INTERPHONE JACKBOX
11. CONTROL PUSH BUTTON
12. MAP CASE
13. PANEL LIGHT SWITCH
14. PANEL LIGHT
15. RADIO COMPASS RECEIVER
16. NAVIGATOR'S TABLE

Figure 47 - Navigator's Communications Controls

of air. Push to open and pull to close. The selection of hot or cold air is made by the pilot.

6. DRIFT METER MASTER SWITCH.

A master switch for the drift meter is below the edge of the navigator's table near the ash receiver on the front forward corner.

7. RADIO COMPASS RECEIVER.

a. The radio compass receiver is above the navigator's table and may be remotely controlled either from the pilot's compartment ceiling or from the control unit on the navigator's table. Operation of the radio compass receiver is the same for the navigator as for the pilot. Refer to section II, paragraph 2.

b. The bearing indicator is mounted beneath the forward inboard corner of the navigator's table and its dial may be seen by lifting the cover on the table. The loop antenna is remotely controlled from the radio compass receiver.

8. APERIODIC COMPASS.

The navigation compass is on the right side of the compartment, below the bomb-sight storage box.

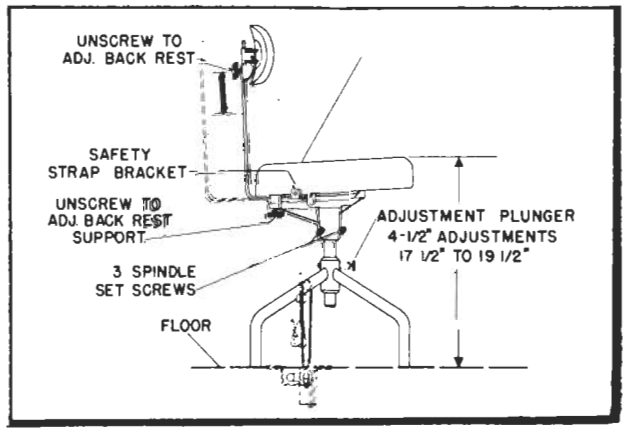


Figure 48 - Navigator's Seat Adjustment

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SECTION VI
UPPER TURRET

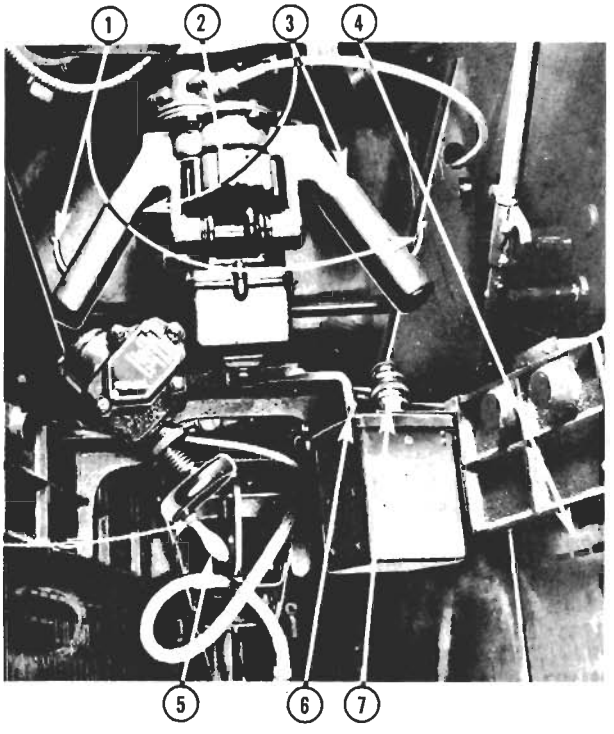
1. GENERAL.

a. Elevation of the guns is controlled by lifting or depressing the hand control grips, the direction corresponding to the direction of the handgrip motion about the horizontal axis.

b. Rotation of the turret is obtained by turning the handgrips about the vertical axis. The range knob is mounted between the grips, so that the gunner rests both thumbs on this knob while holding the grips in the palms of his hands. This knob sets the range in the computing sight.

c. The hydraulic power unit furnishes the mechanical power for rotating the turret and elevating the guns.

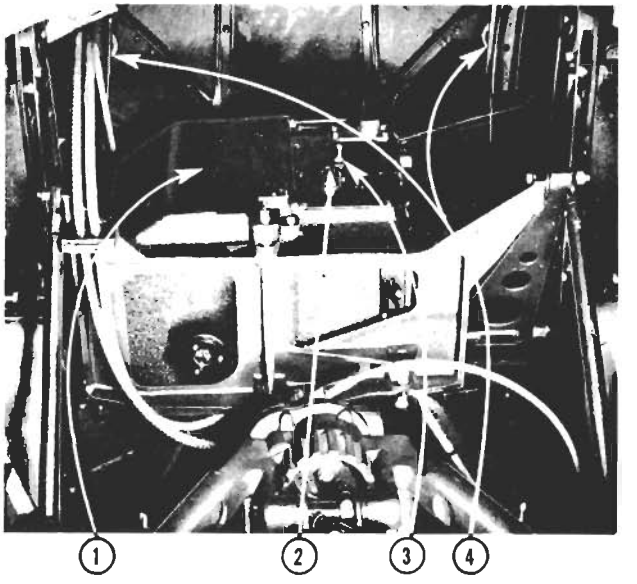
d. A gun firing switch is mounted to the rear and at the upper end of each handgrip. The two firing



KEY TO FIGURE 49

1. DEADMAN SWITCH
2. RANGE KNOB
3. HAND GRIP
4. AMMUNITION BOX
5. AZIMUTH HANDCRANK
6. TROUBLE LIGHT SWITCH
7. TROUBLE LIGHT

Figure 49 - Upper Turret Controls



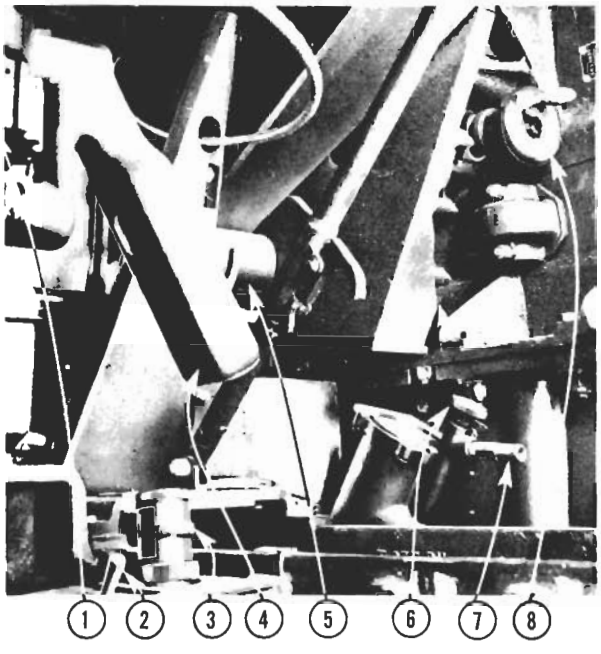
KEY TO FIGURE 50

1. GUN SIGHT
2. SIGHT LIGHT RHEOSTAT CONTROL
3. SIGHT SWITCH
4. GUN CHARGING HANDLES

Figure 50 - Inside Upper Turret

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switches are connected in parallel so that either switch can be used to fire the guns. Deadman switches, one on each grip, are connected in parallel so that the gunner can operate the turret when either hand rests on a grip. The deadman switch is provided so that the power circuits of the turret will be opened and all turret motion and firing of guns will be stopped when the gunner's hands are removed from the grips.



KEY TO FIGURE 51

1. RANGE KNOB	5. DEADMAN SWITCH
2. TROUBLE LIGHT SWITCH	6. OXYGEN FLOW CONTROL
3. TROUBLE LIGHT	7. OXYGEN MASK FITTING
4. HAND GRIP	8. ELEVATION HANDCRANK

Figure 51 - Upper Turret Interior

2. PREFLIGHT CHECK.

a. Allow hydraulic units and sight to warm up at least 5 minutes before take-off.

b. Engage power clutches.

c. See that hand cranks are disengaged. (Do not disengage until after power clutches have been engaged.)

d. Feed ammunition just up to the guns.

e. Move main gun switch to "ON" position.

f. Place sight switch in "ON" position.

g. Close deadman switches on handgrips.

h. Check response of azimuth and elevation mechanisms by manipulating the handgrips.

i. Turn range knob and observe that reticles move in response.

j. Adjust reticle light to approximately the desired brilliance.

3. TURRET OPERATION.

a. Charge guns by pulling each handle twice.

b. Turn on gun selector switches.

c. When target is sighted, set in target dimension on sight.

d. Turn hand controls so that reticles frame the target.

e. Adjust range knob until reticles frame the target.

f. Press either firing switch.

g. After ammunition has been used, charge guns at least twice to clear out live shells.

h. When the turret is not being used, turn it so that the guns point aft and are parallel to the center line of the airplane.

i. In event of power failure, the turret may be controlled by the azimuth and elevation hand cranks. It is not possible to track a target with the hand cranks, but they may be used for approximate positioning of the turret and guns.

j. To use the hand cranks:

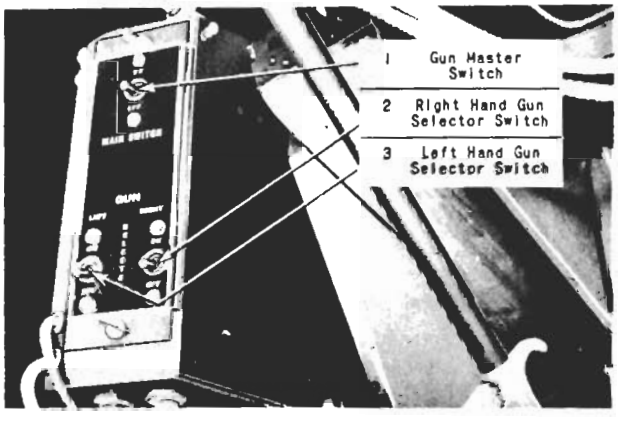


Figure 52 - Upper Turret Switches

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(1) Engage azimuth and elevation hand cranks.

(2) Disengage power clutches.

(3) Move turret and guns into desired position.

(4) When finished, reengage power clutches.

(5) Be sure to disengage hand cranks before operating power motor again.

4. ADJACENT EQUIPMENT.

a. LIGHTING. - A panel light and switch are on the wall of the compartment to the left of the turret. A trouble light and switch are inside of the turret; on the right side looking aft.

b. INTERPHONE. - An interphone jack box is on the wall of the compartment to the left of the turret. Operating instructions are given in section I, paragraph 10.

c. OXYGEN.

(1) An A-12 demand oxygen regulator on the right wall of the compartment is part of the main oxygen system and is operated as instructed in section I, paragraph 9. A continuous flow regulator, type A-9 is inside the turret, on the right side looking aft, and is connected to a separate supply cylinder attached to the turret.

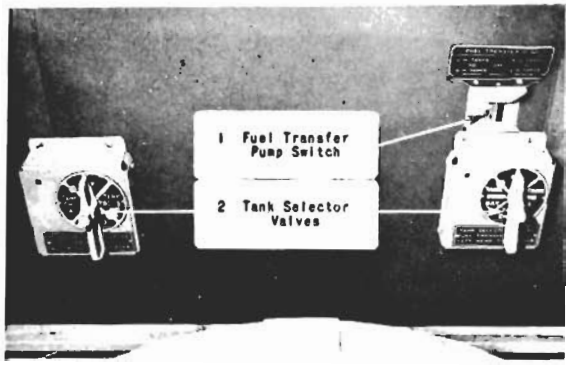


Figure 53 - Fuel Transfer Controls

(2) To use A-9A regulator, attach mask hose to regulator and open the manually operated valve until indicator points to altitude at which airplane is flying. If valve vibrates off setting, tighten packing nut.

(3) The turret supply cylinder can be refilled from the main supply system.

d. FUEL TRANSFER CONTROLS. - Two fuel transfer valves and the transfer pump switch are below the door leading to the bomb bay. Refer to section I, paragraph 4., for operating instructions.

e. HYDRAULIC EQUIPMENT. - The hydraulic pump panel, accumulators, fluid tank, and servicing valves are at the right side of the compartment. Refer to section I, paragraph 3.

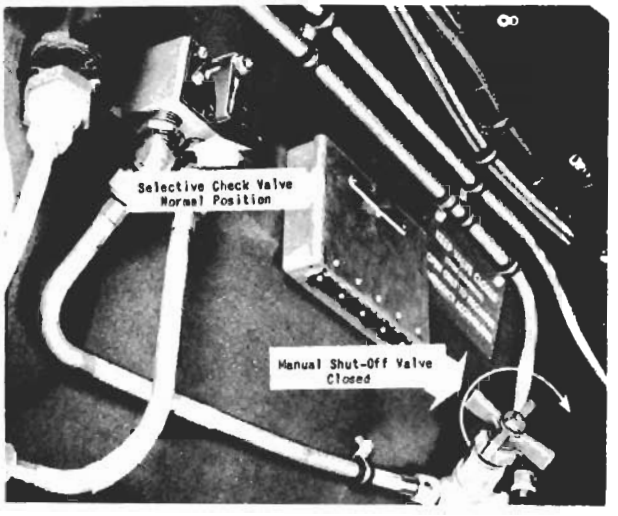


Figure 54 - Hydraulic Servicing Valves

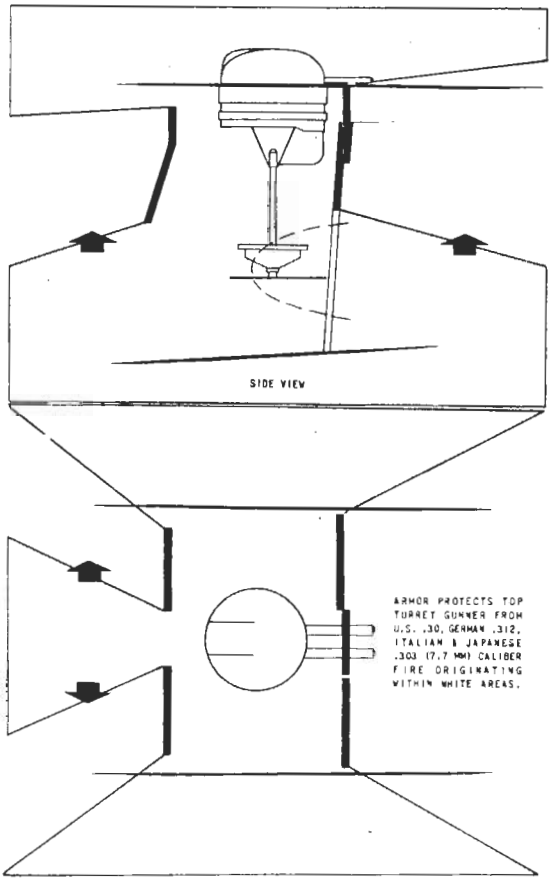


Figure 55 - Top Gunner's Armor Protection

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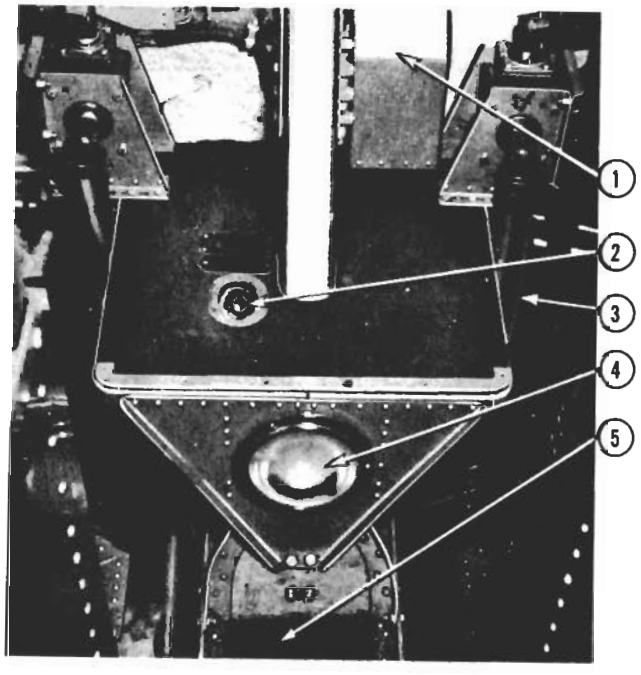


SECTION VII
BOMB BAY

1. LIGHTING.

a. The step light at the forward end of the catwalk is operated by a switch on the forward wall of the radio compartment, to the right of the door.

b. Two dome lights, one on either side of aft end of the bay, are operated by switches on the aft bulkhead to the right of the door.



KEY TO FIGURE 56

1. EMERGENCY BOMB RELEASE 2. BOMB DOOR HAND CRANK CONNECTION 3. HOSE TO FUEL TRANSFER PUMP 4. STEP LIGHT 5. CATWALK

Figure 56 - Forward End of Catwalk - Bomb Bay

2. OXYGEN.

The oxygen regulator is on the aft wall of the bomb bay to the left of the door.

3. EMERGENCY EQUIPMENT.

a. A hand crank connection for manual operation of each main landing wheel is on the forward wall of the bomb bay.

b. A hand crank connection for manual operation of the bomb bay doors is on the step at the forward end of the catwalk.

c. An emergency bomb release handle is also on the step at the forward end of the catwalk and is protected by a hinged guard.

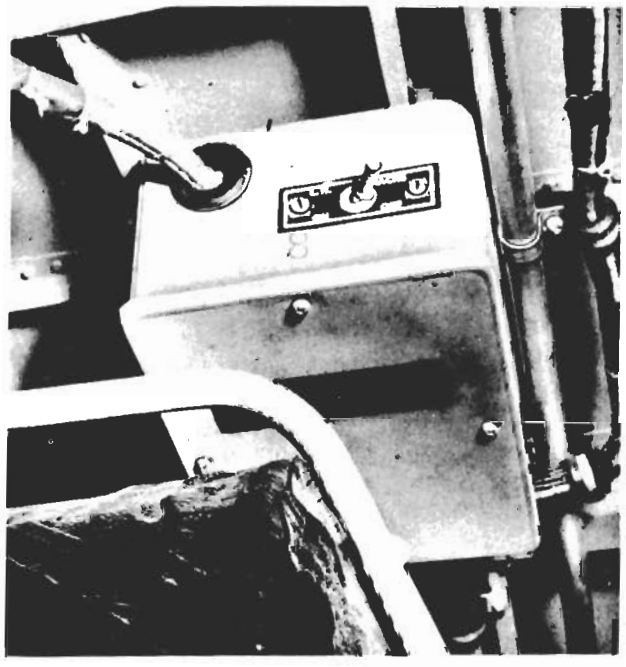


Figure 57 - Bomb Rack Selector Switch - Left Side

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d. For use of emergency equipment, refer to section III.

4. BOMB RACK SELECTOR SWITCHES.

Two switches, one on each side of the bomb bay, are used in conjunction with the rack selector switches on the bombardier's control panel. When either switch is "OFF," electrical release of bombs or fuel tanks from that rack is impossible.

5. HAND TRANSFER OR REFUELING PUMP.

A hand pump mounted on the aft bulkhead of the bomb bay may be used to transfer fuel in case of electrical power failure or may be attached to a main landing gear shock strut and used as a refueling pump. (See figure 60.)

6. AUXILIARY WING FUEL CELL SHUT-OFF VALVES.

Remote control handles, operating shut-off valves in the lines from each group of outer wing fuel cells, are mounted below the door at the aft end of the bomb bay. Refer to section I, paragraph 4., for operating instructions.

NOTE

In some installations these valve controls are in the radio compartment.

7. RELIEF TUBE.

A relief tube is located behind the dome light in the left bomb bay.

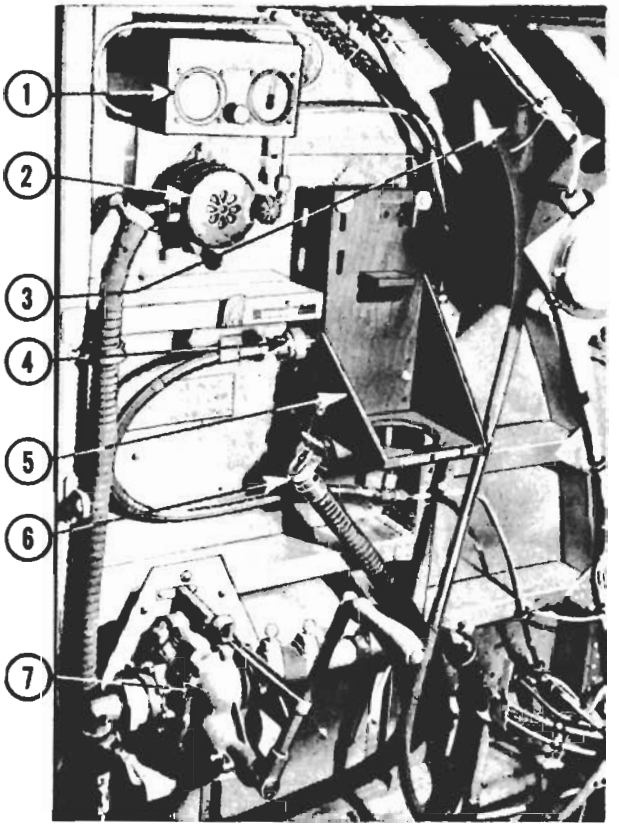


Figure 58 - Bomb Bay - Left Side, Aft

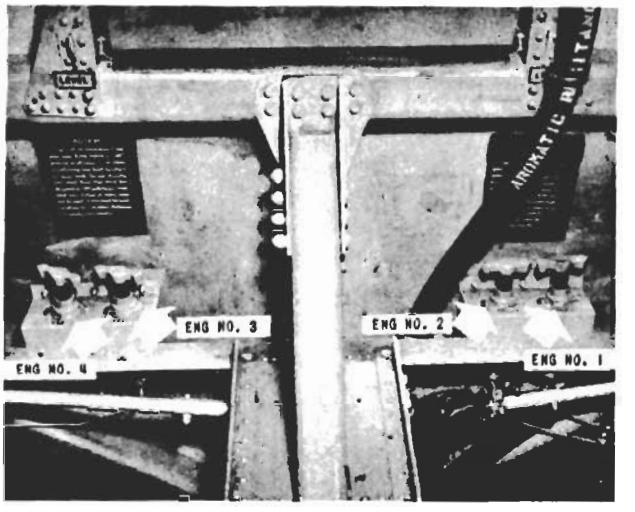


Figure 59 - Auxiliary Fuel Tank Shut-Off Valves

KEY TO FIGURE 58

1. OXYGEN INDICATOR PANEL
2. OXYGEN REGULATOR
3. RELIEF TUBE
4. PORTABLE OXYGEN UNIT RECHARGER
5. PORTABLE OXYGEN UNIT STORAGE BRACKET
6. OXYGEN MASK CONNECTION
7. HAND FUEL PUMP

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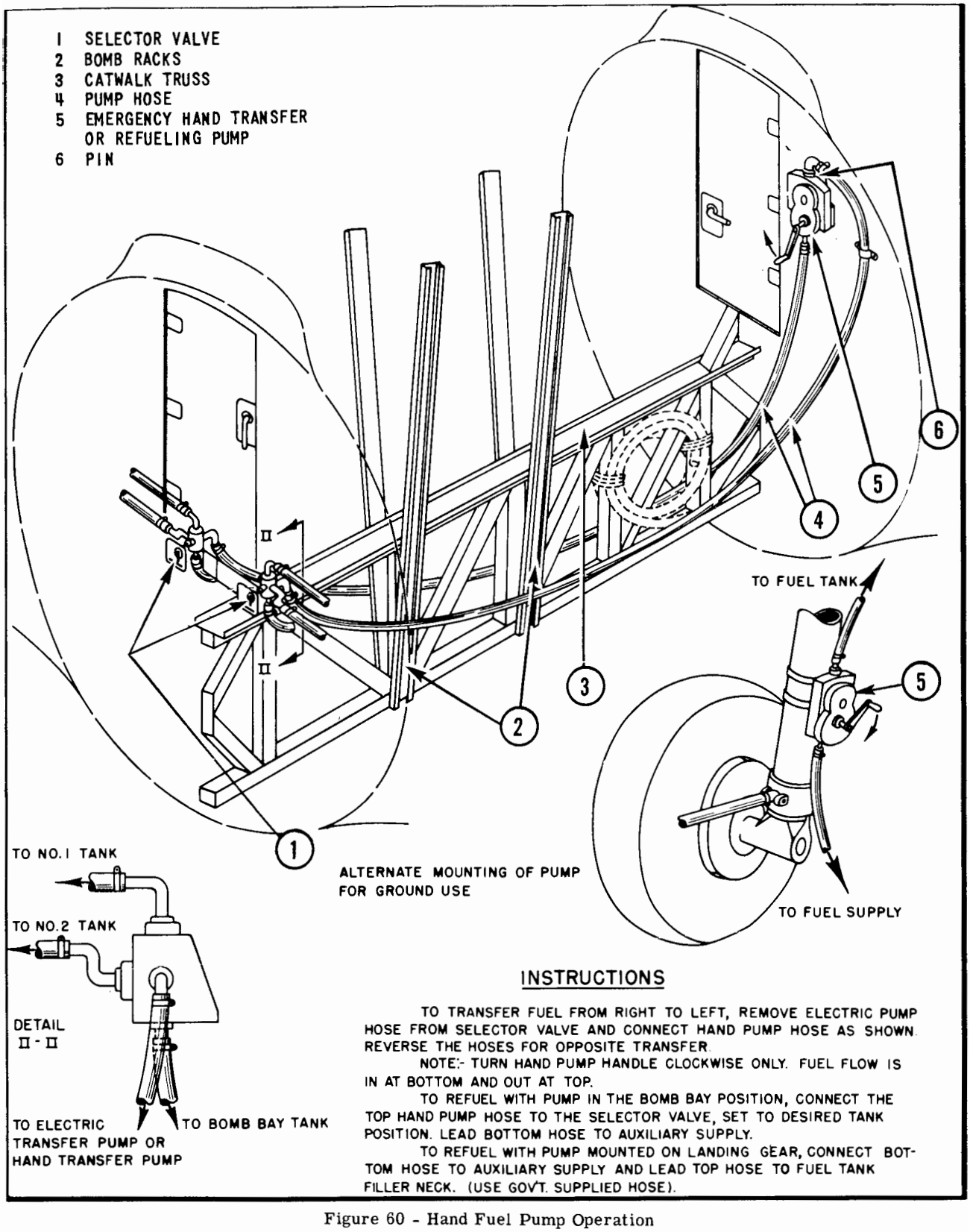
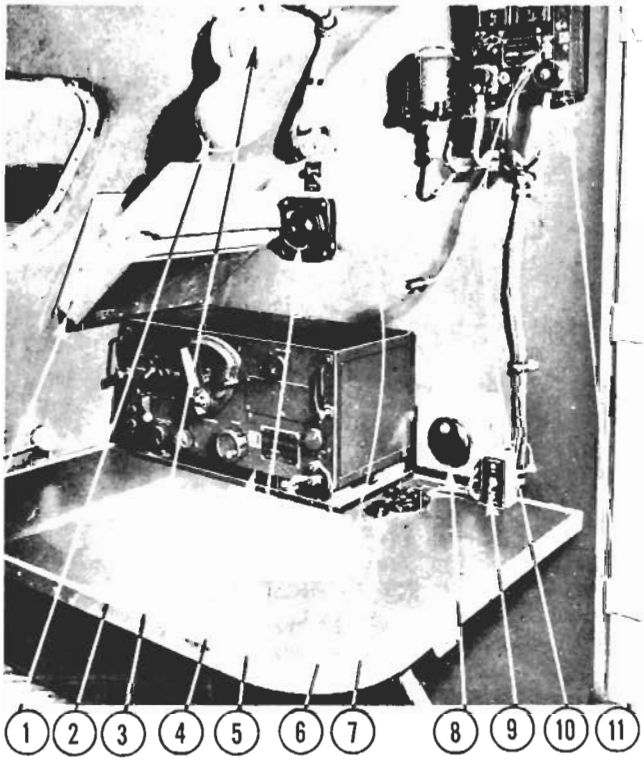


Figure 60 - Hand Fuel Pump Operation

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SECTION VIII
RADIO COMPARTMENT



KEY TO FIGURE 61

1. RADIO OPERATOR'S LIGHT
2. RADIO OPERATOR'S TABLE
3. LIGHT SWITCH
4. LIAISON SET RECEIVER
5. ALARM BELL
6. PHONE CALL LAMP
7. TRANSMITTING KEY
8. ASH RECEIVER
9. LIAISON TRANSMITTER MASTER SWITCH
10. LOCAL "OFF-ON" SWITCH SCR-535
11. RADIO SET SCR-535 CONTROL BOX

Figure 61 - Radio Operator's Table and Controls

1. LIGHTING.

A lamp above the radio operator's table is operated by an adjacent switch. A similar lamp and switch are in the aft end of the compartment above the liaison transmitter. Another lamp and switch are on the side wall to the left of the radio operator's seat.

2. EMERGENCY EQUIPMENT.

a. A fire extinguisher is on the forward wall of the compartment to the right of the door.

b. Two life raft release handles are on the ceiling of the compartment, just aft of the top hatch on the right side.

c. Four red emergency release handles are located along the edge of the top hatch.

d. An alarm bell is on the forward wall of the compartment above the radio operator's table.

e. Two hand cranks and two crank extensions for manual operation of the wing flaps, bomb bay doors, landing gear, tail gear, and engine starters are clipped to the aft wall of the compartment, above the transmitter tuning units. For use of hand cranks refer to section III.

3. OXYGEN CONTROLS.

Oxygen outlets are provided for the radio operator and for each of the two auxiliary crew members. Refer to section I, paragraph 9., for instructions.

4. HEATING AND VENTILATING INLET.

The inlet is on the floor of the compartment, to the left and aft of the radio operator's seat. Push the knob to close; pull, to open. Selection of hot or cold air is controlled by the pilot.

5. INTERPHONE CONTROLS.

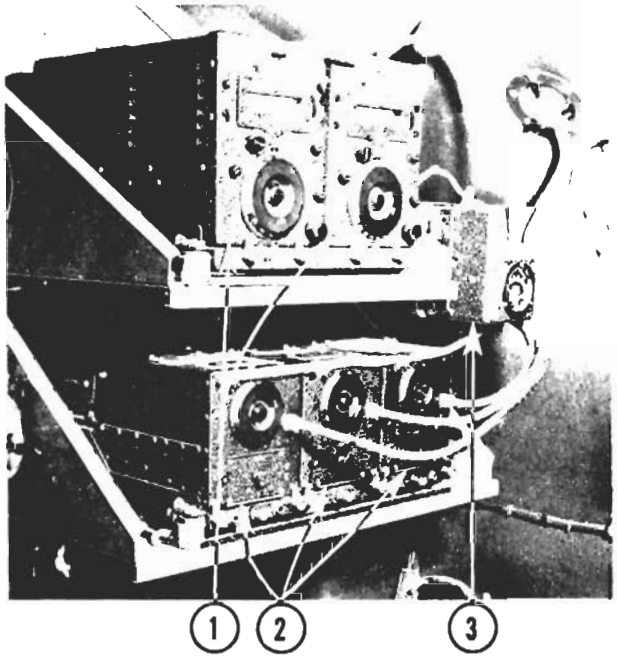
The radio operator's interphone jack box is on the left side wall. Two additional jack boxes are provided in the compartment for other crew members. Refer to section I, paragraph 10., for instructions.

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6. COMMUNICATIONS EQUIPMENT.

a. The communications equipment consists of the following:

Command set	SCR-274-N
Liaison set	SCR-287-A
Radio compass set	SCR-269-G
Interphone equipment	RC-36
Marker beacon equipment	RC-43
Radio altimeter	SCR-518-A
IFF radio set	SCR-535-A

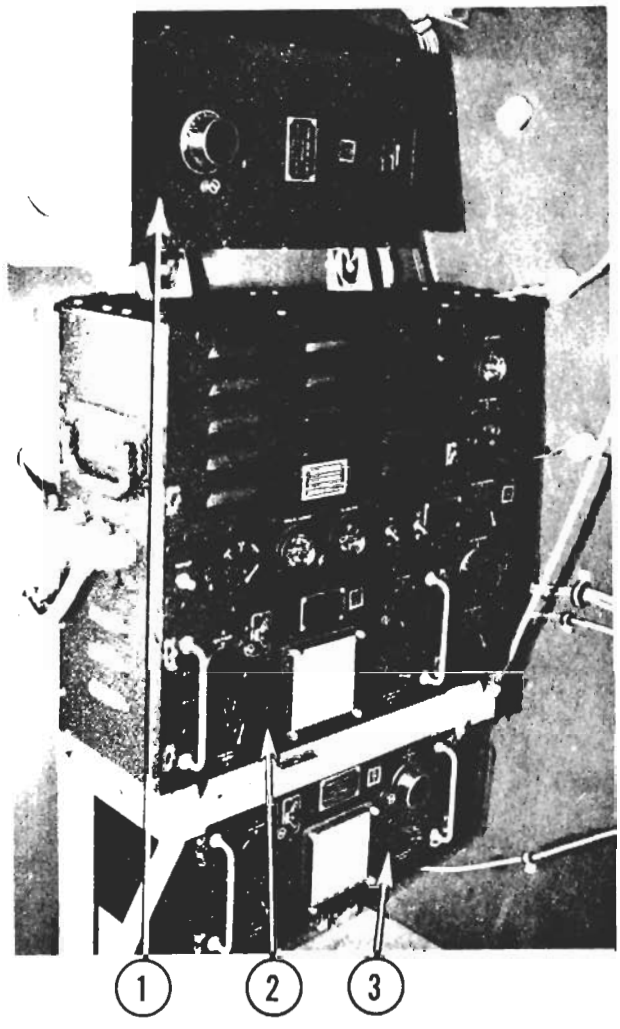


KEY TO FIGURE 62

1. COMMAND TRANSMITTERS
2. COMMAND RECEIVERS
3. ANTENNA RELAY CONTROL BOX

Figure 62 - Command Radio Installation

b. COMMAND RADIO. - Two command radio transmitters and three receivers are mounted on the right side of the compartment on the forward bulkhead. They are controlled by remote control units on the ceiling of the pilot's compartment. The transmitters' dynamotor and modulator are on the floor in the forward right corner of the compartment. The receiver's dynamotors are mounted on supports behind the receivers.



KEY TO FIGURE 63

1. LIAISON ANTENNA TUNING UNIT
2. LIAISON TRANSMITTER
3. TRANSMITTER TUNING UNIT

Figure 63 - Liaison Radio Installation

c. LIAISON RADIO. - The liason transmitter is installed on the left side of the aft bulkhead. The receiver is on the radio operator's table. The dynamotor is on the left rear side of the aft bulkhead, in the ball turret compartment. Two antennas are available for use with the liaison set. One employs the skin of the airplane, with the lead-in attached to the change-over switch on the left side wall. The other is the trailing antenna which is also attached to the change-over switch. The trailing antenna reel is operated electrically from a control box to the right of the change-over switch.

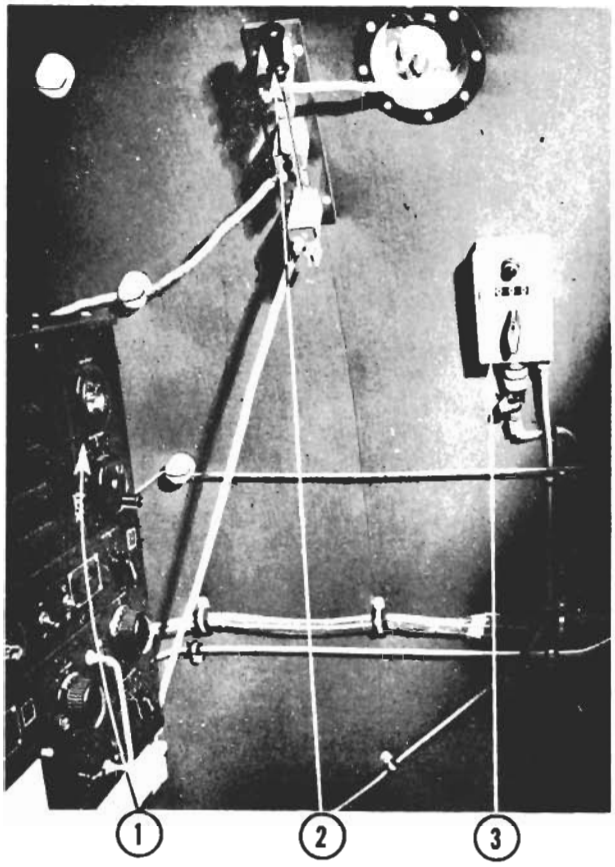
d. RADIO SET, SCR-518-A (HIGH-ALTITUDE ALTIMETER). - Radio set SCR-518-A consists of a

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complete set of apparatus for determining the height of the airplane above the ground. It is operative over an altitude range of 0 to 20,000 feet, and it will work satisfactorily up to 30,000 feet, before the indications become erroneous. Operation of the set does not depend upon barometric pressure. It indicates altitude of the aircraft above the terrain below the airplane, and has no reference to sea level. If the aircraft is flying over broken country, more than one peak will appear on the indicator, the highest one representing the object closest to the airplane.

(1) Place the power switch in the "ON" position. This energizes all parts of the set except the automatic volume control which is controlled by a separate switch. A pilot lamp at the lower center of the control panel should light, indicating that the power is on.

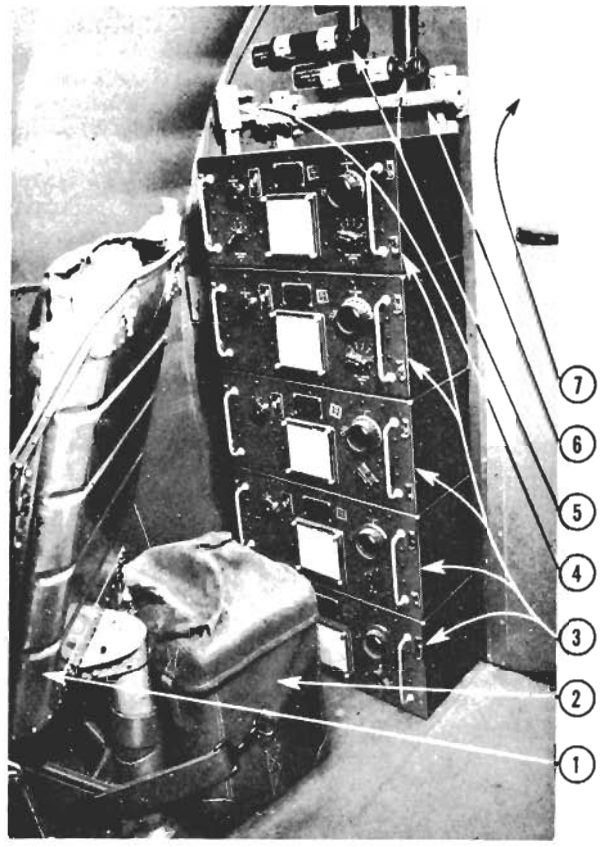
(2) As the tubes reach their operating conditions, the circle traces, and indicating lobes appear on the screen of the indicator. During the first few minutes of operations the indications will be unsteady.



KEY TO FIGURE 64

1. LIAISON TRANSMITTER
2. ANTENNA CHANGE-OVER SWITCH
3. TRAILING ANTENNA REEL CONTROL

Figure 64 - Radio Compartment - Left Side



KEY TO FIGURE 65

1. SEAT FOR AUXILIARY CREW
2. FREQUENCY METER
3. TRANSMITTER TUNING UNITS
4. STARTER CRANK EXTENSION
5. HAND CRANKS
6. CRANK EXTENSION FOR BOMB DOORS AND FLAPS
7. DOOR TO BALL TURRET COMPARTMENT

Figure 65 - Transmitter Tuning Units

(3) Turn the "CIRCLE SIZE" control knob until the two circle traces on the indicator screen are adjusted to the required diameter for readings. The proper size occurs when each circle is just visible as a luminous green ring on the gray background, just beyond the outer circumference of its dark calibrated scale ring.

(4) Turn the "RECEIVER GAIN" control to adjust the lobe readings for clearest legibility on the indicator screen. Maximum receiver sensitivity may be used at the higher altitudes and less than maximum sensitivity may be required at the lower altitudes. The receiver gain control must be adjusted in conjunction with the automatic volume control switch for maximum lobe legibility on the altimeter scale in accordance with the following paragraphs.

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(5) USE OF AUTOMATIC VOLUME CONTROL AT LOWER ALTITUDES.

(a) The automatic volume control improves the performance of the radio set at altitudes below 2000 feet and should only be used for reading up to 2000 feet. With the AVC switch on, receiver sensitivity is reduced but is automatically increased with altitude up to about 2000 feet. Overloading of the receiver is thus prevented at the lower altitudes.

(b) For operation when descending below 2000 feet:

1. At any altitude above 1000 feet, throw AVC switch on.

2. Adjust "RECEIVER GAIN" control until the initial lobe appearing at zero on the 2000-foot scale is the proper height.

3. The reception lobe giving the altitude reading on the 2000-foot scale should now remain approximately constant in size as the ground is approached.

(6) USE OF AVC AT HIGHER ALTITUDES. - The AVC switch must be turned off, when the equipment is operating at altitudes above 2000 feet, as the AVC would otherwise impair the receiver sensitivity in certain sections of the higher-altitude ranges.

(7) Starting from zero and reading in a clockwise direction, read the counterclockwise edge of each lobe on each circle trace. (If the lobe is on the top of the dial, read to the left edge, and if it is at the bottom of the dial, read the right edge.) The first lobe (or index lobe) appears at the zero calibration on each scale. The second lobe (reflection lobe) indicates the altitude above terrain.

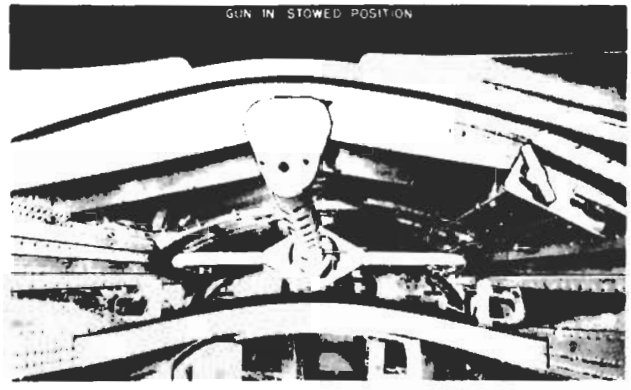
(a) On each scale (inner and outer), the index lobe will appear at the zero calibration. The second (reflection lobe) on each scale indicates the absolute altitude of the aircraft.

(b) The inner circle is merely a vernier on the outer circle. On the outer circle, it is possible to read to within 250 feet. If greater accuracy is required, the inner scale reading must be taken into consideration, as follows: Read the outer scale to the next lower even thousand (4000, for instance). Read the inner scale. If the reading of the inner scale should be 750 feet, the actual altitude of the aircraft is then obtained by adding the readings of the two scales: 4750 feet. The inner scale can, with practice, be read to within 25 feet.

(c) If the zero lobes have shifted away from zero, correct readings may be obtained by adding the amount of zero shift, if the shift is to the left of zero, and by subtracting the amount of zero shift, if the shift is to the right, from the reading of altitude which was obtained by following the procedure outlined in the preceding paragraph.

7. FREQUENCY METER.

A portable frequency meter for use with any radio is carried in each airplane. No provision is made for stowage, so the unit is usually strapped to the support of the rear auxiliary seat in the radio compartment.



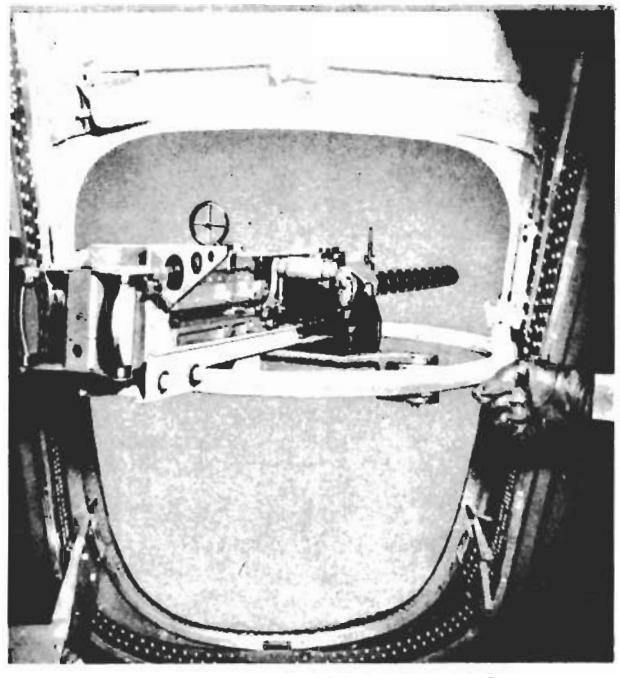


Figure 66 - Radio Compartment Gun

8. RADIO COMPARTMENT GUN.

In some airplanes a single .50-caliber flexible machine gun is mounted on a yoke in top of the radio compartment to fire through the top hatch opening. The yoke slides on rails from stowed to firing position.

9. CAMERA PIT.

a. Camera equipment is installed in the pit under the floor of the radio compartment accessible door.

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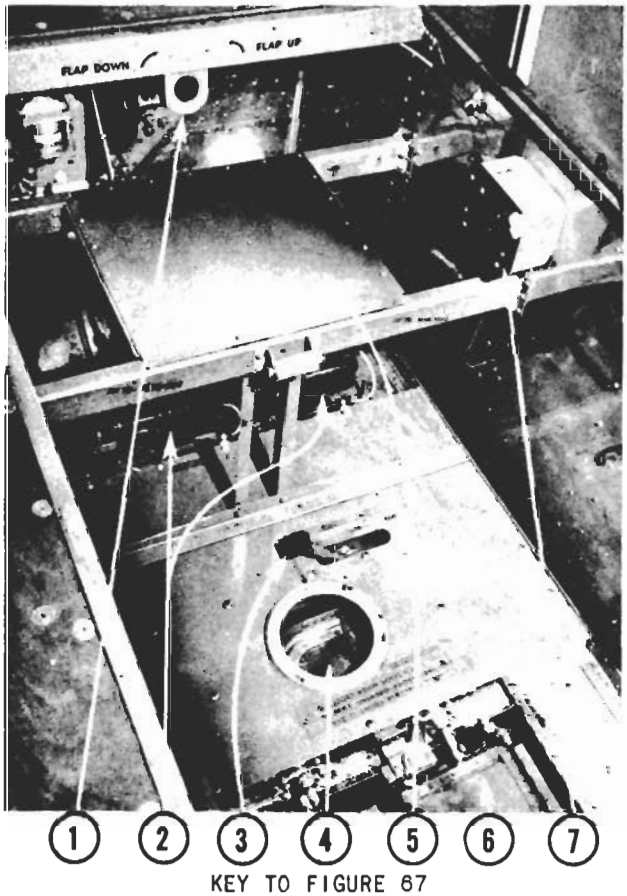
Provision is made for three alternate installations as follows:

Type T-3A Installation:

Camera	Type T-3A
Camera mount	A-5A
View finder	A-2
Filter	A-3
Shutter induction coil	

Type K-3B Installation:

Camera	Type K-3B
Camera mount	A-8
View finder	A-2
Intervalometer	
Magazine	A-1A
Filter	A-2A



KEY TO FIGURE 67

1. WING FLAP HAND CRANK CONNECTION
2. PROPELLER ANTI-ICER PUMPS
3. CAMERA DOOR CONTROL HANDLE
4. VIEWFINDER APERTURE
5. CAMERA OPERATOR'S SEAT
6. CAMERA DOOR
7. INTERVALLMETER POWER RECEPTACLE

Figure 67 - Camera Pit

Type K-7C Installation:

Camera	Type K-7C
Camera mount	A-8
View finder	A-2
Filter	A-4

b. The type A-2 view finder may be installed forward of the camera. The bracket assembly used to support the intervalometer is stowed on the right side of the camera pit. The intervalometer is stowed on the right side. A direct current power receptacle for the intervalometer is installed on the right side of the pit and a connection to the vacuum system is provided on the left side.

c. The double camera doors (figure 67) and the view finder door are hinged in the bottom of the fuselage and are operated by a lever located on the floor at the operator's seat.

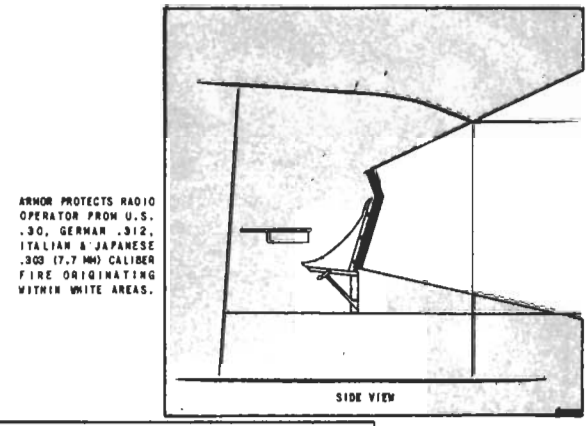


Figure 68 - Radio Operator's Armor Protection

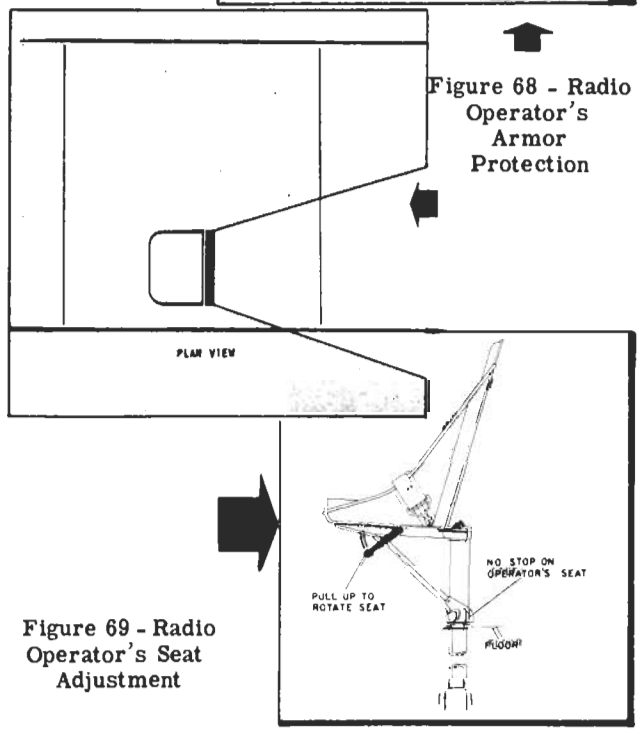
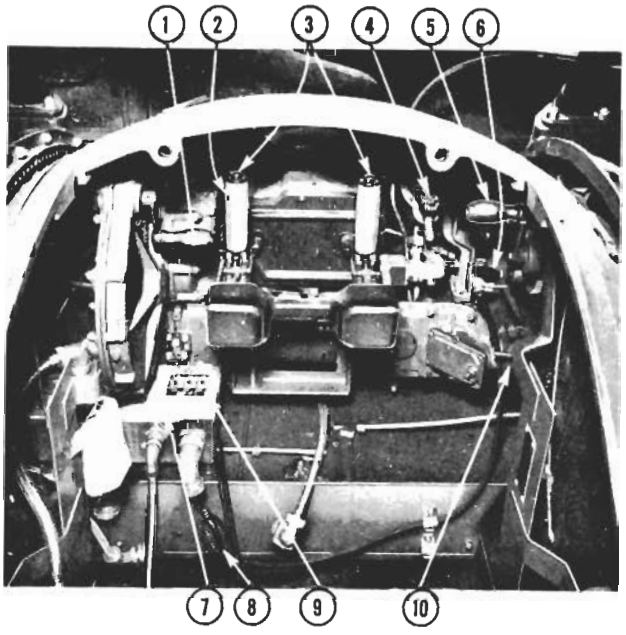


Figure 69 - Radio Operator's Seat Adjustment

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SECTION IX
BALL TURRET



KEY TO FIGURE 70

1.	ELEVATION HANDCRANK	6.	SPOT LIGHT
2.	HAND CONTROL GRIP	7.	ELECTRICAL SWITCH BOX
3.	FIRING SWITCHES	8.	SPOT LIGHT CONTROL SWITCH
4.	OXYGEN REGULATOR	9.	GUN SELECTOR SWITCHES
5.	AZIMUTH HANDCRANK	10.	ELEVATION POWER CLUTCH

Figure 70 - Interior of Ball Turret

1. GENERAL.

a. A Sperry ball-type power turret, equipped with twin .50-caliber machine guns, is installed in the bottom of the fuselage aft of the radio compartment.

b. A hydraulic unit provides power for driving the turret in azimuth and elevation.

c. The hand control and limit unit controls the outputs of the azimuth and elevation hydraulic systems. A pair of handgrips controls the motion of the turret in azimuth and elevation. Each handgrip has a firing switch on the top end.

d. The switch box controls distribution of the electric power to the various units in the turret. The terminal block in the top left end of the box has convenient posts for connecting the leads of the gunner's head set and microphone.

2. ENTERING THE TURRET.

CAUTION

Do not attempt to rotate the turret in elevation while the airplane is on the ground. No crew member shall be in the turret during landing or take-off and the guns of the turret shall be in the horizontal position pointing aft.

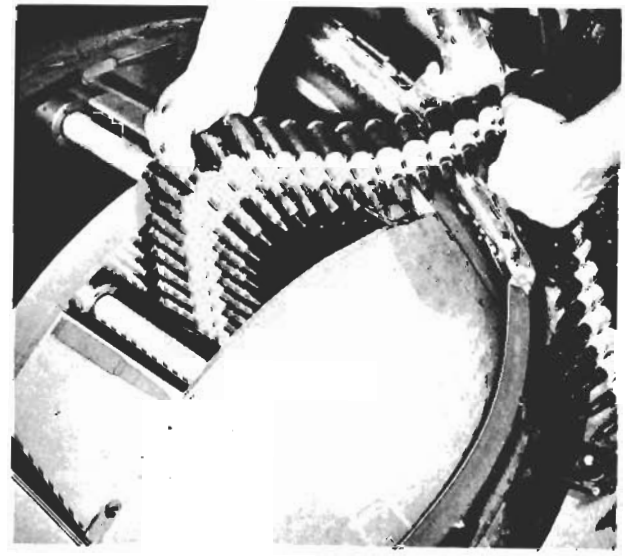
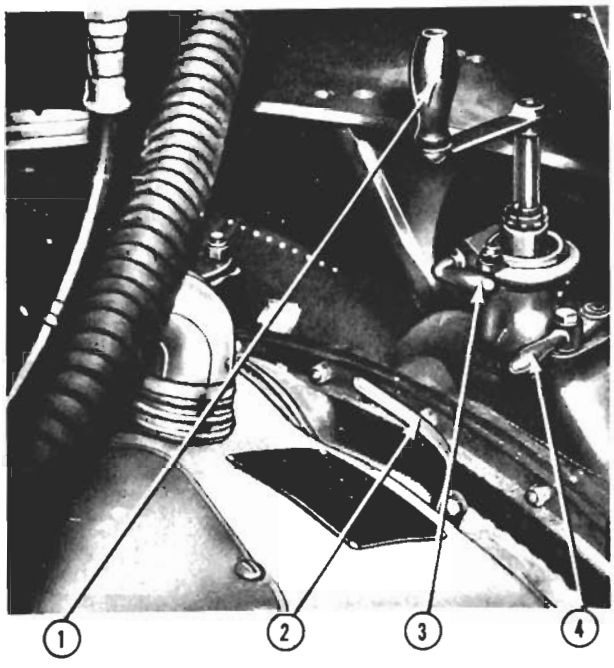


Figure 71 - Loading Ball Turret Ammunition Boxes

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KEY TO FIGURE 72

1 - ELEVATION HANDCRANK
2 - LUG WRENCH
3 - ELEVATION HANDBRAKE
4 - ELEVATION HANDCLUTCH

Figure 72 - External Manual Controls

a. Remove ammunition box cover and load. Push ammunition down to the guns.

b. Remove elevation hand crank from its clip and attach it to shaft. Be sure that the hand brake (figure 72) is locked.

c. Move elevation hand clutch to "IN" position. It may be necessary to loosen hand brake and rock hand crank back and forth before hand clutch can be moved to "IN" position.

d. Move elevation power clutch to "OUT" position using clutch handle; then, replace handle in its clip.

e. Loosen elevation brake slowly while holding elevation hand crank firmly.

f. Turn elevation hand crank in down direction until turret revolves to low limit of elevation (-90 degrees).

g. While holding elevation hand crank, open turret door, reach inside, and move elevation power clutch to "IN" position.

h. Move elevation hand clutch to "OUT" position, remove hand crank, and replace it in its clip.

i. Enter turret. Close door securely. Be sure door handles are pushed all the way up and that the

KEY TO FIGURE 73

1. ELECTRICAL SWITCH BOX
2. SPOT LIGHT SWITCH
3. GUNNER'S SEAT
4. RANGE FOOT PEDAL
5. HEADSET AND MICROPHONE LEADS
6. TURRET FRONT WINDOW
7. FOOT REST
8. CHARGING HANDLE
9. TURRET HAND CONTROL AND LIMIT UNIT
10. ELEVATOR POWER CLUTCH

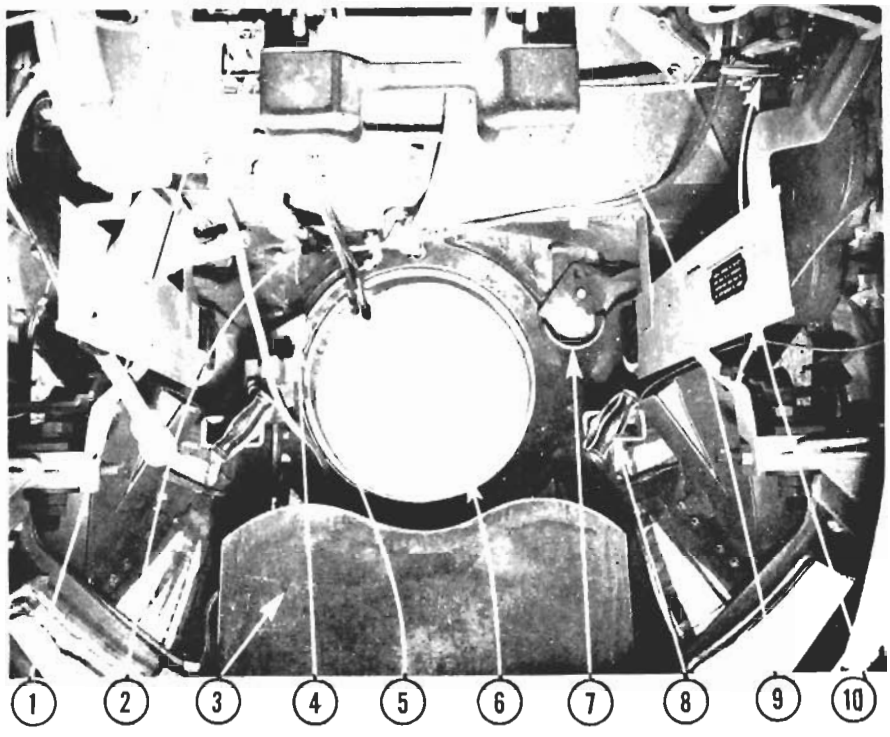


Figure 73
Ball Turret, Top View

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Figure 74 - Inside Ball Turret

turret door is locked before turning main power and sight switches "ON."

3. PREFLIGHT CHECK.

a. Turn power switch "ON."

b. Turn sight switch "ON."

c. Check response of azimuth and elevation mechanisms by manipulating the hand controls.

WARNING

Be sure that the guns are not driven down into the ground.

d. Adjust reticle light on sight to desired brilliance (approximately).

e. Work range foot pedal and observe if reticles move in response.

f. Lift each gun cover plate and pull ammunition down, feeding first shell by hand into magazine of gun; then, close gun cover plates.

4. OPERATION.

a. Load ammunition boxes. (See figure 71.) Enter turret.

b. Turn on power switch.

c. Turn on sight switch.

d. Charge guns by pulling charging handles twice.

e. Turn on fire selector switches.

f. By means of hand controls track the target.

g. Operate range foot pedal until reticles frame the target.

h. Close either firing key.

i. When ammunition is used up, charge guns at least twice to be sure that no live shells are left in the guns.

5. INTERPHONE.

A press-to-talk switch for inter-communication is located just in front of the gunner's right footrest.

6. SUIT HEATER.

A rheostat control is provided for use with the gunner's heated suit. It is located on the underneath side of the seat and is adjusted to obtain the desired temperature in the suit.

7. OXYGEN.

An oxygen regulator is provided on the inside of the ball turret on the right side. Refer to section VI, paragraph 4.c., for operation. Oxygen is supplied from the auxiliary cylinder above the turret. When the supply of this auxiliary cylinder is exhausted, it can be renewed from the airplane's main supply system.

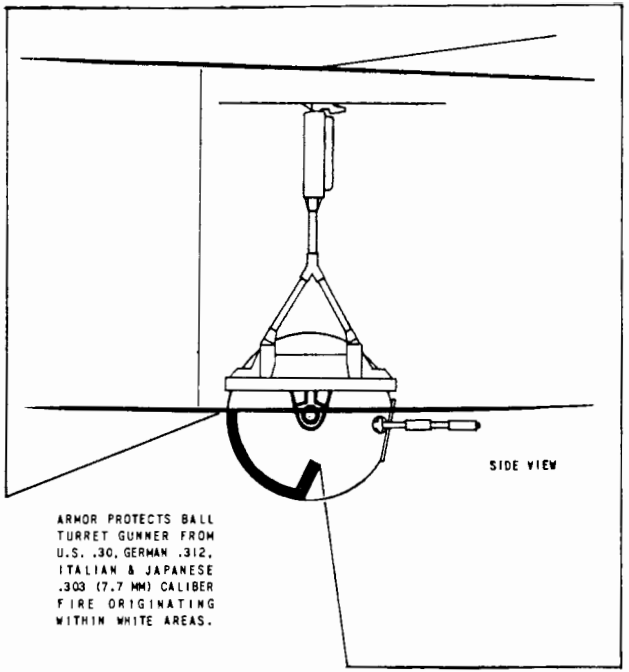
8. ADJACENT EQUIPMENT.

a. LIGHTING. - A dome light in the ceiling just aft of the turret support is operated by a switch to the right of the door to the radio compartment.

b. EMERGENCY RADIO - SCR 578. - Some airplanes are provided with a completely independent emergency radio which is carried on the right rear side of bulkhead 6 beside the ball turret. Refer to section III, paragraph 14., for further instructions.

c. FIRST-AID KIT. - A first-aid kit is clipped to the aft side of the bulkhead between the ball turret compartment and the radio compartment to the left of the door.

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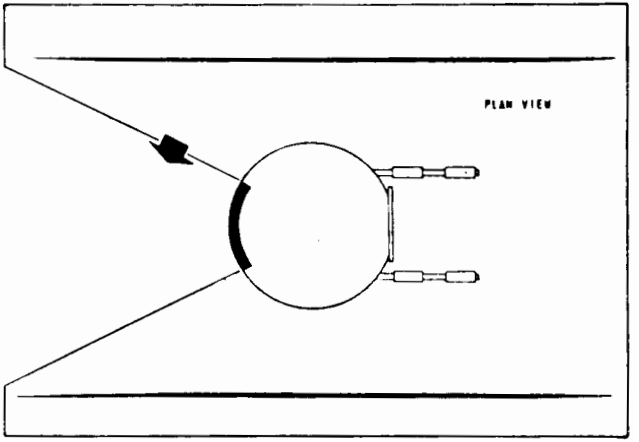


Figure 75
Ball Turret Gunner's
Armor Protection

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SECTION X
SIDE GUNNER'S COMPARTMENT

1. LIGHTING.

The dome light switch is aft of the entrance door.

2. INTERPHONE CONTROLS.

Interphone jack boxes are provided for both gunners. Refer to section I, paragraph 10., for operation.

3. SUIT HEATER OUTLET.

Rheostats control the temperature of the gunners' heated suits. They are adjusted to obtain the desired temperature in the suits.

4. OXYGEN.

Oxygen regulators and portable oxygen units are provided for each side gunner. Refer to section I, paragraph 9., for instructions.

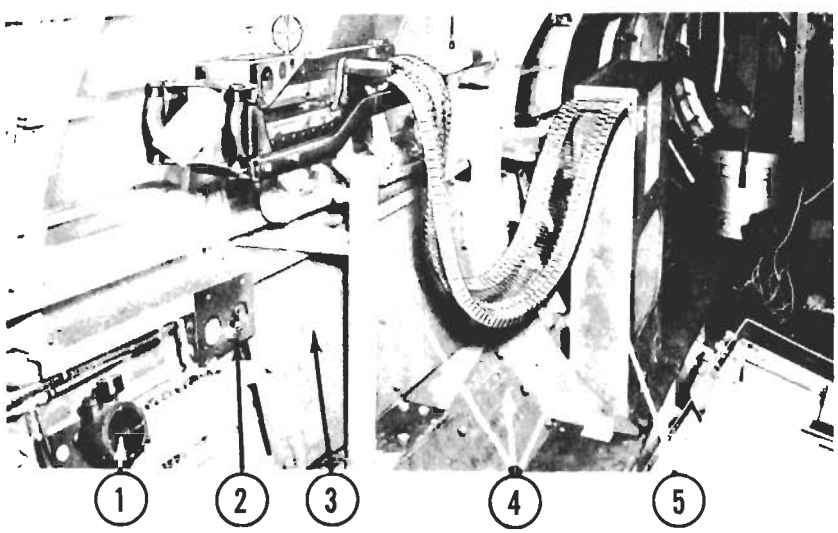
5. EMERGENCY EQUIPMENT.

a. FIRE EXTINGUISHER. - A carbon tetrachloride fire extinguisher is attached to the forward side of the bulkhead aft of the main entrance.

b. EMERGENCY RELEASES. - Each side window has an emergency release bar on the forward side of each window. To open the window, jerk the bar forward. There are no catches to be released. The main entrance door also has an emergency release handle.

6. GUN OPERATION.

To prepare the machine guns for action, remove the straps (figures 76 and 77) and swing the guns into position.



KEY TO FIGURE 76

1. PORTABLE OXYGEN UNIT 2. OXYGEN INDICATOR PANEL 3. MACHINE GUN, STOWED
4. ARMOR PLATE 5. AMMUNITION BOX

Figure 76 - Right Side Gun Stowed

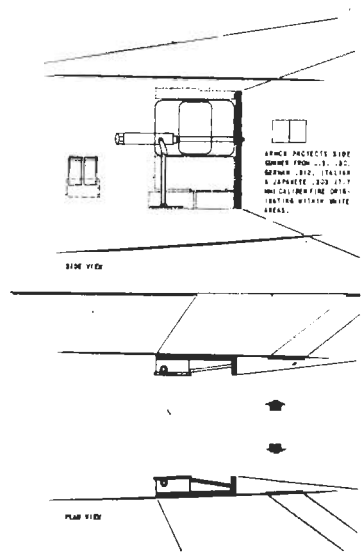
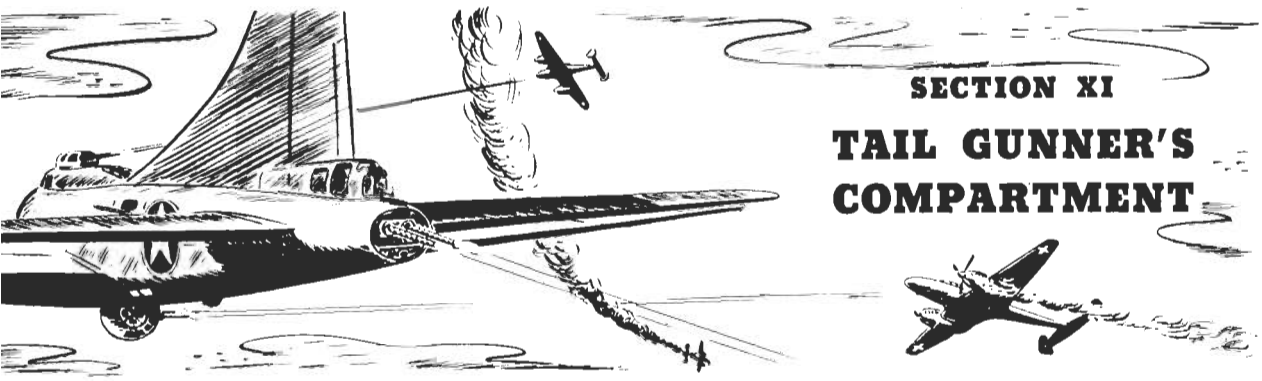


Figure 77 - Side Gunner's Armor Protection

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SECTION XI
TAIL GUNNER'S COMPARTMENT

ENTRANCE.

There are two ways of entering the tail gunner's compartment: one from the tail wheel compartment through a small door in the bulkhead, and one from the outside through a side door. The latter is used as emergency exit, and is equipped with an emergency release handle.

LIGHTING.

A dome light and switch are located above the gun cradles behind the armor plate.

3. INTERPHONE.

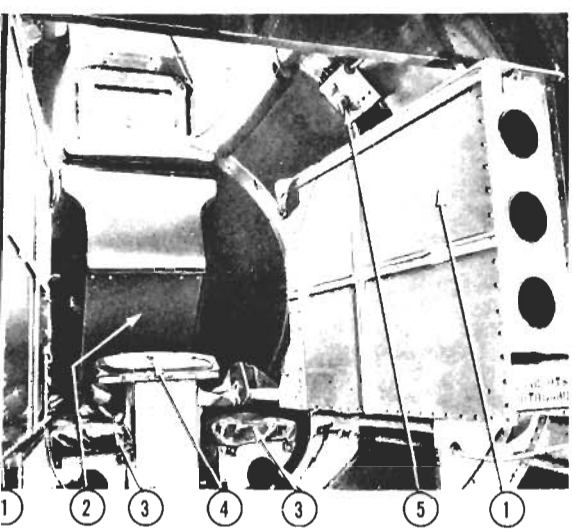
The jack box is on the right side of the compartment looking aft above the aft end of the ammunition box. Refer to section I, paragraph 10.

4. OXYGEN.

Two oxygen regulators are provided, one on each side wall. Refer to section I, paragraph 9.

5. SUIT HEATER OUTLET.

A rheostat control, provided for use with the gunner's heated suit is adjusted to obtain the desired temperature in the suit.



KEY TO FIGURE 78

1. AMMUNITION BOXES 2. ARMOR PLATE
3. KNEE PADS 4. TAIL GUNNER'S SEAT
5. INTERPHONE JACKBOX

Figure 78 - Tail Gunner's Compartment

Figure 79 - Tail Gunner's Compartment Door



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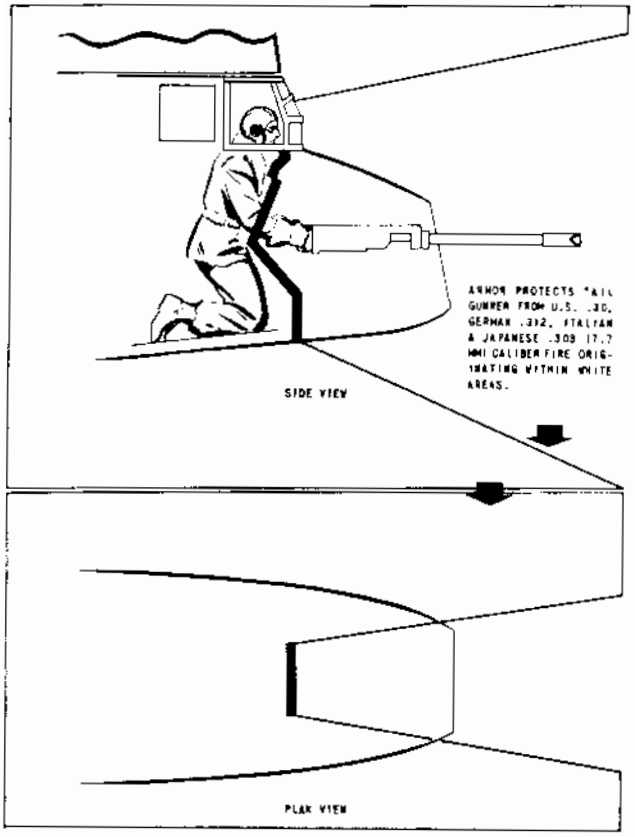
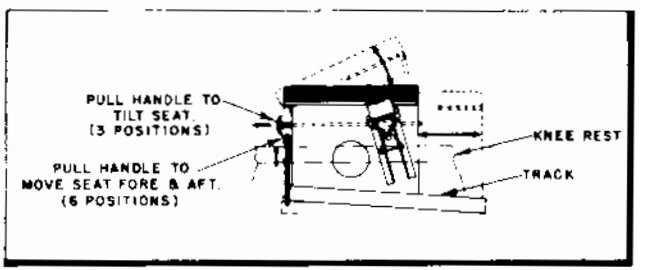


Figure 80 - Tail Gunner's Armor Protection

Figure 81 - Tail Gunner's Seat Adjustment



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APPENDIX I

U. S. A. - BRITISH GLOSSARY OF NOMENCLATURE

U. S. A	BRITISH
Accumulator (hydraulic)	Should not be confused with electrical accumulator or battery
Airfield	Aerodrome
Battery (electrical)	Electrical accumulator
Bombardier	Bomb aimer
Ceiling	Cloud height
Check valve (hydraulic)	Non-return valve
Copilot	Second pilot
Cylinder (hydraulic)	Jack
Dump valve	Jettison valve
Empennage	Tail Unit
Flight indicator	Artificial horizon
Gasoline (gas)	Petrol
Glass, bulletproof	Armour glass
Gross weight	All-up weight
Ground (electrical)	Earth
Gyro horizon	Artificial horizon
Gyro pilot	Automatic pilot
(to) Land	(to) Alight
Lean	Weak
Left	Port
(to) Level off	(to) Flatten out
Line, mooring	Mooring guy
Manifold pressure	Boost
Mast, radio	Rod aerial
Overload	Non-standard load
Panel, outboard	Outer plane
Reticle (gun sight)	Graticule
Screen	Filter
Set, command	Pilot controller set
Set, liaison	General purpose set
Airplane	Aircraft
Speed, indicated air (IAS)	Air-speed-indicator reading
Stabilizer, horizontal	Tail plane
Stabilizer, vertical	Fin
Stack	Manifold (inlet or exhaust)
Tachometer	Engine speed indicator
Tube (radio)	Valve
Turn indicator	Direction indicator
Valve (fuel or oil)	Cock
Weight empty	Tare
Windshield	Windscreen
Wing	Main plane



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APPENDIX II

FLIGHT OPERATION DATA

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CAUTION

POWER SETTINGS GIVEN IN THESE CHARTS ARE APPLICABLE ONLY WHEN USING 100 OCTANE FUEL. REFER TO APPENDIX III FOR RESTRICTIONS WITH USE OF 91 OCTANE FUEL.

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FORM ASC-512A

AIRPLANE MODELS B-17 F

SPECIFIC ENGINE FLIGHT CHART

ENGINE MODELS R-1820-97

CONDITION	FUEL PRESSURE (LB SQ. IN.)	OIL PRESSURE (LB SQ. IN.)	OIL TEMP. °C	OIL TEMP. °F	COOLANT TEMP. °C	COOLANT TEMP. °F
DESIRED	12-16	75	70	158		
MAXIMUM	16	80	88	190		
MINIMUM	12	70				
IDLING		15				

MAX. PERMISSIBLE DIVING RPM: 2760

CONDITION	ALLOWABLE OIL CONSUMPTION
NORMAL RATED (MAX. CONT.)	14.5 U.S.QT/HR 23 IMP.PT/HR
MAX. CRUISE	8.0 U.S.QT/HR 13 IMP.PT/HR
MIN. SPECIFIC	5 U.S.QT/HR 8 IMP.PT/HR

OIL GRADE: (S) 1120 (W) 1100-A

SUPERCHARGER TYPE: TURBO

FUEL GRADE: 100 OCTANE

OPERATING CONDITION	RPM	MANIFOLD PRESSURE (BOOST)	HORSE-POWER	CRITICAL ALTITUDE WITH RAM	CRITICAL ALTITUDE NO RAM	BLOWER	USE LOW BLOWER BELOW:	MIXTURE CONTROL POSITION	FUEL FLOW (GAL/HR/ENG.) U.S.	FUEL FLOW (GAL/HR/ENG.) IMP.	MAXIMUM CYL. TEMP. °C	MAXIMUM CYL. TEMP. °F	MAXIMUM DURATION (MINUTES)
TAKE-OFF	2500	46	1200	27,000		-		A.R.	152	127	260	500	5
WAR EMERGENCY													
MILITARY	2500	48*	1200	27,000		-		A.R.	152	127	260	500	5
NORMAL RATED (MAX. CONT.)	2300	41.5*	1000	30,000		-		A.R.	103	86	232 CLIMB 218	450 CLIMB 424	
MAXIMUM CRUISE	2000	35.2*	750	35,000				A.L.	62.5	52	205	401	
MINIMUM SPECIFIC CONSUMPTION	2000 1940 1780 1700 1600 1400	34.7 33.0 36.0 35.5 34.8 32.5	670 600 650 600 550 450	SEE ENGINE CALIBRATION CURVE				A.L.	52 44 45.5 41.5 37.8 31.2	43 37 38 34.6 31.4 26	205	401	

REMARKS: AIR INTAKE FILTER MUST BE OFF ABOVE 8000 FEET OR DANGEROUS TURBO OVERSPEEDING WILL RESULT. FULL THROTTLE MUST BE USED ABOVE 15,000 FEET OR DANGEROUS TURBO OVERSPEEDING WILL RESULT. DO NOT MANUALLY LEAN. AUTO LEAN GIVES MAXIMUM RANGE. *DECREASE MANIFOLD PRESSURES 1-1/2 INCH PER 1000 FEET ABOVE CRITICAL ALTITUDE.

WP-1-1-43-5M

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SPEC. AN-H-8 Dec. 18, 1942 FORM ASC-510

AIRPLANE MODELS
B-17F

TAKE-OFF, CLIMB & LANDING CHART

ENGINE MODELS
R-1820-97

TAKE-OFF DISTANCE (IN FEET)

GROSS WEIGHT (IN LBS.)	HEAD WIND MPH	HEAD WIND KNOTS	HARD SURFACE RUNWAY AT SEA LEVEL GROUND RUN	HARD SURFACE RUNWAY AT SEA LEVEL TO CLEAR 50' OBJ.	HARD SURFACE RUNWAY AT 3,000 FT. GROUND RUN	HARD SURFACE RUNWAY AT 3,000 FT. TO CLEAR 50' OBJ.	HARD SURFACE RUNWAY AT 6,000 FT. GROUND RUN	HARD SURFACE RUNWAY AT 6,000 FT. TO CLEAR 50' OBJ.	SOD-TURF RUNWAY AT SEA LEVEL GROUND RUN	SOD-TURF RUNWAY AT SEA LEVEL TO CLEAR 50' OBJ.	SOD-TURF RUNWAY AT 3,000 FT. GROUND RUN	SOD-TURF RUNWAY AT 3,000 FT. TO CLEAR 50' OBJ.	SOD-TURF RUNWAY AT 6,000 FT. GROUND RUN	SOD-TURF RUNWAY AT 6,000 FT. TO CLEAR 50' OBJ.	SOFT SURFACE RUNWAY AT SEA LEVEL GROUND RUN	SOFT SURFACE RUNWAY AT SEA LEVEL TO CLEAR 50' OBJ.	SOFT SURFACE RUNWAY AT 3,000 FT. GROUND RUN	SOFT SURFACE RUNWAY AT 3,000 FT. TO CLEAR 50' OBJ.	SOFT SURFACE RUNWAY AT 6,000 FT. GROUND RUN	SOFT SURFACE RUNWAY AT 6,000 FT. TO CLEAR 50' OBJ.
65,000	0	0	3350	4400	4000	5100	4600	5800	3950	5000	4500	5600	5600	6800	8950	10,000				
	20	17	2300	3200	2750	3700	3250	4300	2500	3600	3150	4100	3950	5000	7100	8000				
	40	35	1400	2050	1750	2500	2250	3050	1850	2500	2150	2900	2600	3600	5350	6000				
57,000	0	0	2350	3000	2700	3400	3100	3900	2650	3300	3100	3800	3600	4400	4350	5000	5500	6200	7000	7800
	20	17	1600	2100	1800	2400	2150	2800	1800	2300	2100	2700	2550	3200	3100	3000	4000	4600	5150	5900
	40	35	1000	1400	1100	1550	1350	1850	1100	1500	1350	1800	1600	2100	2100	2500	2750	3200	3700	4200
50,000	0	0	1700	2250	1900	2500	2100	2750	1850	2400	2100	2700	2350	3000	2650	3200	3200	3800	3650	4300
	20	17	1150	1600	1300	1750	1400	1900	1250	1700	1450	1900	1600	2100	1800	2250	2250	2700	2600	3100
	40	35	650	1000	800	1150	850	1250	750	1100	900	1250	1000	1400	1100	1450	1450	1900	1600	2000

NOTE: INCREASE DISTANCE 10 % FOR EACH 10°C ABOVE 0°C (10 % FOR EACH 20°F ABOVE 32°F)

ENGINE LIMITS FOR TAKE-OFF 2500 RPM & 46 IN. HG

COMBAT MISSIONS USE 2300 RPM & 38 IN. HG

CLIMB DATA

FERRY MISSIONS USE 2300 RPM & 38 IN. HG

GROSS WEIGHT IN LBS.	TYPE OF CLIMB	S.L. TO 5,000 FT. ALT. BEST I.A.S. MPH	BEST I.A.S. KNOTS	FT/MIN	TIME FROM S.L.	10,000 FT. ALT. BEST I.A.S. MPH	BEST I.A.S. KNOTS	FT/MIN	TIME FROM S.L.	FUEL FROM S.L. U.S.	FUEL FROM S.L. IMP.	15,000 FT. ALT. BEST I.A.S. MPH	BEST I.A.S. KNOTS	FT/MIN	TIME FROM S.L.	FUEL FROM S.L. U.S.	FUEL FROM S.L. IMP.	25,000 FT. ALT. BEST I.A.S. MPH	BEST I.A.S. KNOTS	FT/MIN	TIME FROM S.L.	FUEL FROM S.L. U.S.	FUEL FROM S.L. IMP.	30,000 FT. ALT. BEST I.A.S. MPH	BEST I.A.S. KNOTS	FT/MIN	TIME FROM S.L.	FUEL FROM S.L. U.S.	FUEL FROM S.L. IMP.	BLOWER CHANGE
65,000	COMBAT	135	117	575		135	117	500	18	125	104	135	117	400	29	200	167	135	117	140	65	450	375							RE-DUCE MP.
	FERRY																													
57,000	COMBAT	135	117	830		135	117	750	13	90	75	135	117	660	20	140	117	135	117	360	39	270	225	135	117	170	55	380	317	1-1/2" DR. (1000 FT.
	FERRY																													
50,000	COMBAT	135	117	1060		135	117	990	9	60	50	135	117	890	15	105	87	135	117	600	28	195	162	135	117	380	38	260	217	ABOVE 30,000 FT.
	FERRY																													

FUEL INCLUDES WARM-UP AND TAKE-OFF ALLOWANCE

LANDING DISTANCE (IN FEET)

GROSS WEIGHT IN LBS.	BEST I.A.S. APPROACH MPH	BEST I.A.S. APPROACH KNOTS	HARD DRY SURFACE AT SEA LEVEL TO CLEAR 50' OBJ.	HARD DRY SURFACE AT SEA LEVEL GROUND ROLL	HARD DRY SURFACE AT 3,000 FT. TO CLEAR 50' OBJ.	HARD DRY SURFACE AT 3,000 FT. GROUND ROLL	HARD DRY SURFACE AT 6,000 FT. TO CLEAR 50' OBJ.	HARD DRY SURFACE AT 6,000 FT. GROUND ROLL	FIRM DRY SOD AT SEA LEVEL TO CLEAR 50' OBJ.	FIRM DRY SOD AT SEA LEVEL GROUND ROLL	FIRM DRY SOD AT 3,000 FT. TO CLEAR 50' OBJ.	FIRM DRY SOD AT 3,000 FT. GROUND ROLL	FIRM DRY SOD AT 6,000 FT. TO CLEAR 50' OBJ.	FIRM DRY SOD AT 6,000 FT. GROUND ROLL	WET OR SLIPPERY AT SEA LEVEL TO CLEAR 50' OBJ.	WET OR SLIPPERY AT SEA LEVEL GROUND ROLL	WET OR SLIPPERY AT 3,000 FT. TO CLEAR 50' OBJ.	WET OR SLIPPERY AT 3,000 FT. GROUND ROLL	WET OR SLIPPERY AT 6,000 FT. TO CLEAR 50' OBJ.	WET OR SLIPPERY AT 6,000 FT. GROUND ROLL
50,000	110	98	3500	1950	3800	2150	4100	2350	4150	2800	4500	2850	4900	3150	5450	3900	5900	4300	6450	4700

REMARKS

LEGEND
I.A.S. Indicated Air Speed
M.P.H. Miles Per Hour
S.L. Sea Level
U.S. U.S. Gallons
IMP. Imperial Gallons
NOTE All Distances are Average
RED FIGURES HAVE NOT BEEN FLIGHT CHECKED

Take-Off, Climb and Landing Chart

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FORM ASC-511A

MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 1 OF 7 SHEETS

GR. WT. 65,000 TO 60,000 POUNDS

EXTERNAL LOAD ITEMS
NONE

CONDITION	R.P.M.	M.P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	DURATION IN MIN.	U.S. G.P.H.	IMP. G.P.H.
TAKE-OFF	2500	46	-	A.R.	5	608	-
MILITARY POWER	2500	46	-	A.R.	5	608	-
ENGINE (S)	R-1820-97						

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. **NOTES:** (A) Avoid continuous cruising in Column I except in emergency. (B) Columns (II, III, IV & V) toward the right progressively give increase in range at sacrifice in speed. (C) Manifold Pressure (M.P.), Gallons Per Hour (G.P.H.), are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES STATUTE AT S.L.	I RANGE IN AIR MILES STATUTE AT 25,000	I RANGE IN AIR MILES NAUTICAL AT S.L.	I RANGE IN AIR MILES NAUTICAL AT 25,000	FUEL U.S. GALS. (2)	II RANGE IN AIR MILES STATUTE	II RANGE IN AIR MILES NAUTICAL	III RANGE IN AIR MILES STATUTE	III RANGE IN AIR MILES NAUTICAL	IV RANGE IN AIR MILES STATUTE	IV RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. (2)	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V (MAX. RANGE) RANGE IN AIR MILES NAUTICAL
				3612	212 U.S. GALLONS NOT AVAILABLE IN FLIGHT						3612		
	1910		1660	3400	2240	1940	2480	2150	2750	2390	3400	3070	2660
	1800		1560	3200	2110	1830	2330	2020	2590	2250	3200	2890	2500
	1690		1470	3000	1980	1720	2190	1900	2430	2110	3000	2700	2350
	1580		1370	2800	1840	1600	2040	1770	2270	1970	2800	2520	2190
	1460		1270	2300	1710	1480	1900	1650	2100	1820	2600	2350	2040
	1350		1170	2400	1580	1370	1750	1520	1940	1680	2400	2160	1880
	1240		1080	2200	1450	1260	1600	1390	1780	1540	2200	1980	1720
	1130		980	2000	1320	1150	1460	1270	1620	1410	2000	1810	1570
	1010		880	1800	1190	1030	1310	1140	1460	1270	1800	1620	1410
	900		780	1600	1050	910	1170	1010	1300	1130	1600	1440	1250

CONTINUED ON SHEET 2

I OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	(1) DENSITY ALT. IN FEET	II OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	III OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	IV OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	(1) DENSITY ALT. IN FEET	V OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.
2300	156	A.R.	38	413		30000																			30000	BELOW 20,000 FT. SET RPM TO MAINTAIN 150 IAS WITH 29 INCHES HG: ABOVE 20,000 FT. USE 140 MPH IAS AND 29 + 1 INCH HG. IF SPEED CANNOT BE OBTAINED UP TO 2000 RPM AND 29 INCH HG. USE HIGHER RPM AND RECOMMENDED MAN. PRESSURES. USE AUTO-LEAN MIXTURE WHEN AT OR BELOW 2100 RPM.
2300	172	A.R.	38	413		25000																			25000						
2300	183	A.R.	38	413		20000	2150	151	A.R.	31.5	284														20000						
2300	192	A.R.	38	413		15000	2150	168	A.R.	32	296		2100	160	A.L.	31	253		2050	149	A.L.	30	210		15000						
2300	197	A.R.	38	413		12000	2150	172	A.R.	32	290		2100	167	A.L.	31	252		2050	157	A.L.	30	212		12000						
2300	202	A.R.	38	413		9000	2150	175	A.R.	31.5	280		2100	173	A.L.	31	250		2050	164	A.L.	30	212		9000						
2300	207	A.R.	38	413		6000	2100	179	A.L.	31.5	273		2100	178	A.L.	31	245		2050	170	A.L.	30	211		6000						
2300	210	A.R.	38	413		3000	2100	183	A.L.	31.5	269		2100	182	A.L.	31	240		2050	175	A.L.	29.5	208		3000						
2300	214	A.R.	38	413		S.L.	2100	187	A.L.	31	262		2100	185	A.L.	30.5	235		2050	179	A.L.	29	203		S.L.						

LEGEND

(1) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE.

(2) ALLOW 212 U.S. GALS. — IMP. GALS. FOR WARM UP.
TAKE-OFF AND CLIMB TO 5,000 FEET ALTITUDE
RETURN FUEL FLOWS TO TANK
USE FUEL FROM TANKS IN THE FOLLOWING ORDER

REFER TO "SPECIFIC ENGINE FLIGHT CHART" FOR ADDITIONAL ENGINE OPERATION DATA.

BOLD NUMBERS: Use Auto-Rich
LIGHT NUMBERS Use Auto-Lean
WITH TWO SPEED BLOWER Use high blower above heavy line only

I.A.S. Indicated Air Speed
M.P. Manifold Pressure (In. Hg)
U.S.G.P.H. U.S. Gallons Per Hour
IMP.G.P.H. Imperial Gallons Per Hour
F.T. Full Throttle
S.L. Sea Level

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (no external load) 7 Sheets

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FORM ASC-511A

MODEL (S)
B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 2 OF 7 SHEETS

GR. WT. 65,000 TO 60,000 POUNDS

EXTERNAL LOAD ITEMS
NONE

CONDITION	R.P.M.	M.P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	DURATION IN MIN.	U.S. G.P.H.	IMP. G.P.H.
TAKE-OFF	2500	46	-	A.R.	5	608	-
MILITARY POWER	2500	46	-	A.R.	5	608	-

ENGINE (S) R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. *NOTES:* (A) Avoid continuous cruising in Column I except in emergency. (B) Columns (II, III, IV & V) toward the right progressively give increase in range at sacrifice in speed. (C) Manifold Pressure (M.P.), Gallons Per Hour (G.P.H.), are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES STATUTE AT S.L.	I STATUTE AT 25,000	I NAUTICAL AT S.L.	I NAUTICAL AT 25,000	FUEL U.S. GALS. (2)	II RANGE IN AIR MILES STATUTE	II NAUTICAL	III RANGE IN AIR MILES STATUTE	III NAUTICAL	IV RANGE IN AIR MILES STATUTE	IV NAUTICAL	FUEL IMP. GALS. (2)	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V NAUTICAL
					CONTINUED FROM		SHEET 1						
	900		*780*	1600	1050	*910*	1170	*1020*	1300	*1130*	1600	1450	*1260*
	790		*690*	1400	920	*800*	1020	*890*	1140	*990*	1400	1270	*1100*
	680		*590*	1200	790	*690*	870	*760*	970	*850*	1200	1090	*950*
	560		*490*	1000	660	*570*	730	*630*	810	*700*	1000	900	*780*
	450		*390*	800	530	*460*	580	*510*	650	*560*	800	720	*630*

OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	(1) DENSITY ALT. IN FEET	II OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	III OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	IV OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	(1) DENSITY ALT. IN FEET	V OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.
2300	156	A.R.	38	413		30000																			30000	BELOW 20,000 FT. SET RPM TO MAINTAIN 150 IAS WITH 29±IHG. ABOVE 20,000 FT. USE 140 IAS WITH 29±1 IN. HG. IF SPEED CANNOT BE OBTAINED UP TO 2000 RPM ADD 25 IN. HG. USE HIGHER RPM AND RECOMMENDED MANIFOLD PRESSURES. USE AUTO-LEAN MIXTURE WHEN AT OR BELOW 2100 RPM.					
2300	172	A.R.	38	413		25000																			25000						
2300	183	A.R.	38	413		20000	2150	151	A.R.	31.5	284														20000						
2300	192	A.R.	38	413		15000	2150	168	A.R.	32	296		2100	160	A.L.	31	253		2050	149	A.L.	30	210		15000						
2300	197	A.R.	38	413		12000	2150	172	A.R.	32	290		2100	167	A.L.	31	252		2050	157	A.L.	30	212		12000						
2300	202	A.R.	38	413		9000	2150	175	A.R.	31.5	280		2100	173	A.L.	31	250		2050	164	A.L.	30	212		9000						
2300	207	A.R.	38	413		6000	2100	179	A.L.	31.5	273		2100	178	A.L.	31	245		2050	170	A.L.	30	211		6000	ABOVE RANGES APPLY UP TO 10,000 FT. ONLY.					
2300	210	A.R.	38	413		3000	2100	183	A.L.	31.5	269		2090	182	A.L.	31	240		2050	175	A.L.	29.5	208		3000						
2300	214	A.R.	38	413		S.L.	2100	187	A.L.	31	262		2080	185	A.L.	30.5	235		2050	179	A.L.	29.5	203		S.L.						

LEGEND

(1) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE.

(2) ALLOW ____ - ____ U. S. GALS. ____ - ____ IMP. GALS. FOR WARM UP.
TAKE-OFF AND CLIMB TO ____ - ____ FEET ALTITUDE
RETURN FUEL FLOWS TO TANK ____
USE FUEL FROM TANKS IN THE FOLLOWING ORDER ____

BOLD NUMBERS: Use Auto-Rich
LIGHT NUMBERS: Use Auto-Lean
WITH TWO SPEED BLOWER: Use high blower above heavy line only

I.A.S.: Indicated Air Speed
M.P.: Manifold Pressure (In. Hg)
U.S.G.P.H.: U. S. Gallons Per Hour
IMP.G.P.H.: Imperial Gallons Per Hour
F.T.: Full Throttle
S.L.: Sea Level

REFER TO "SPECIFIC ENGINE FLIGHT CHART" FOR ADDITIONAL ENGINE OPERATION DATA.

WF-1-1-43-5M

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (no external load) 7 Sheets

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MODEL (S)
B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 3 OF 7 SHEETS

60,000 TO 55,000 POUNDS GR WT

EXTERNAL LOAD ITEMS
NONE

CONDITION	R.P.M.	M.P.	BLOWER POSITION	M. PROP. POSITION	DURATION OF RATING	U.S. GAL. PER HR.	TRIP OR OPT.
TAKE OFF	2500	46	-	A.R.	5	608	-
MILITARY POWER	2500	46	-	A.R.	5	608	-

ENGINE III R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read operating conditions. NOTES: (A) Avoid continuous cruising in Column I except in emergency. (B) Columns I, II, III, & IV present the most economical cruising conditions. (C) Manifold Pressure (M.P.) Limited For Rich (G.F.) use approximate maximum values for reference. (D) For quiet reference take-off and minimum power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES		FUEL U.S. GALS.	II RANGE IN AIR MILES		III RANGE IN AIR MILES		IV RANGE IN AIR MILES		FUEL U.S. GALS.	V (MAX. RANGE) RANGE IN AIR MILES	
STATUTE	NAUTICAL		STATUTE	NAUTICAL	STATUTE	NAUTICAL	(Statute)	NAUTICAL		(Statute)	NAUTICAL
[illegible]	[illegible]	(1) 2770	175 U.S. GALLONS NOT AVAILABLE IN FLIGHT						(1) 2770		
1530	[illegible]	2600	1900	[illegible]	2050	[illegible]	2280	[illegible]	2600	2470	[illegible]
1410	[illegible]	2400	1760	[illegible]	1930	[illegible]	2100	[illegible]	2400	2290	[illegible]
1300	[illegible]	2200	1610	[illegible]	1770	[illegible]	1920	[illegible]	2200	2090	[illegible]
1180	[illegible]	2000	1470	[illegible]	1610	[illegible]	1750	[illegible]	2000	1900	[illegible]
1060	[illegible]	1800	1320	[illegible]	1450	[illegible]	1580	[illegible]	1800	1710	[illegible]
940	[illegible]	1600	1170	[illegible]	1290	[illegible]	1400	[illegible]	1600	1520	[illegible]
820	[illegible]	1400	1030	[illegible]	1130	[illegible]	1230	[illegible]	1400	1330	[illegible]
710	[illegible]	1200	880	[illegible]	970	[illegible]	1050	[illegible]	1200	1140	[illegible]
580	[illegible]	1000	730	[illegible]	810	[illegible]	880	[illegible]	1000	950	[illegible]
470	[illegible]	800	580	[illegible]	640	[illegible]	700	[illegible]	800	780	[illegible]

CONTINUED ON SHEET 4

OPERATING DATA						(1) WEIGHT ALT. (In Feet)	OPERATING DATA							OPERATING DATA							OPERATING DATA						(1) WEIGHT ALT. (In Feet)	OPERATING DATA					
R.P.M.	I.A.S. (M.P.H.)	NIE-TUBE	M.P. (In. Hg.)	U.S. G.P.H.	U.S. G.P.H.		R.P.M.	I.A.S. (M.P.H.)	NIE-TUBE	M.P. (In. Hg.)	U.S. G.P.H.	U.S. G.P.H.		R.P.M.	I.A.S. (M.P.H.)	NIE-TUBE	M.P. (In. Hg.)	U.S. G.P.H.	U.S. G.P.H.		R.P.M.	I.A.S. (M.P.H.)	NIE-TUBE	M.P. (In. Hg.)	U.S. G.P.H.	U.S. G.P.H.		R.P.M.	I.A.S. (M.P.H.)	NIE-TUBE	M.P. (In. Hg.)	U.S. G.P.H.	U.S. G.P.H.
2300	184	A.R.	36	413		30000																						30000						
2300	176	A.R.	38	413		25000	2150	148	A.R.	31.6	278		2100	143	A.L.	31	252												25000						
2300	188	A.R.	36	413		20000	2150	158	A.R.	31.5	274		2100	157	A.L.	31	245		2050	150	A.L.	30	2.6						20000						
2300	194	A.R.	38	413		15000	2150	167	A.R.	31	265		2100	165	A.L.	30.5	238		2050	159	A.L.	30	208						15000						
2300	199	A.R.	38	413		12000	2150	173	A.R.	31	252		2100	168	A.L.	30.5	234		2050	164	A.L.	29.5	207						12000						
2300	203	A.R.	36	413		9000	2100	176	A.L.	31	264		2100	173	A.L.	30.5	228		2050	168	A.L.	29.5	202						9000						
2300	209	A.R.	36	413		6000	2100	182	A.L.	31	262		2100	177	A.L.	30	222		2050	172	A.L.	29.5	190						6000						
2300	213	A.R.	38	413		3000	2100	188	A.L.	31	248		2050	181	A.L.	30	217		2000	174	A.L.	29	182						3000						
2300	217	A.R.	38	413		S.L.	2100	189	A.L.	30.5	238		2050	184	A.L.	30	201		2000	177	A.L.	29	187						S.L.						

(1) INDICATED ALTITUDE CORRECTED FOR STD. AIR TEMPERATURE.

(1) ALLOW 370 U.S. GALS. FOR WARM UP, TAKE-OFF AND CLIMB TO 5000 FEET ALTITUDE.

[illegible]

USE FUEL FROM TANKS IN THE FOLLOWING ORDER

[illegible]

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (no external load) 7 Sheets

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MODEL (S)
B-17 F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 4 OF 7 SHEETS

60,000 TO 55,000 POUNDS

EXTERNAL LOAD ITEMS
NONE

INSTRUCTIONS FOR USING CHART: Select figure in first column above to obtain total amount of fuel to be carried. Move horizontally to the right to the first figure equal to or greater than the air miles to be flown. Normally, below and opposite desired cruising altitude read cruising condition (RPM) (IAS) and continue reading in Column I

except in emergency. (B) Columns (I, II, III & IV) toward the right give progressively less economical cruising conditions (C) Manifold Pressure (M.P.) Options for Most (C.R.C.) are approximately maximum values for reference (D) For speed reference lists all and various power data see tables in the upper left corner of chart

ALTERNATE CRUISING CONDITIONS

NO WIND — (NO RESERVE) FUEL ALLOWANCE

I NORMAL RATED MAX CONT. RANGE IN AIR MILES STATUTE	I NORMAL RATED MAX CONT. RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. ②	II RANGE IN AIR MILES STATUTE	II RANGE IN AIR MILES NAUTICAL	III RANGE IN AIR MILES STATUTE	III RANGE IN AIR MILES NAUTICAL	IV RANGE IN AIR MILES STATUTE	IV RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. ②	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V (MAX. RANGE) RANGE IN AIR MILES NAUTICAL
			CONTINUED FROM SHEET 3								
460	4[illegible]	MCD	530	[illegible]	610	[illegible]	700	[illegible]	800	760	660
310	[illegible]	POC	340	[illegible]	390	[illegible]	450	[illegible]	500	570	500
200	[illegible]	MCD	290	[illegible]	320	[illegible]	360	[illegible]	400	380	330
21[illegible]	[illegible]	POC	150	[illegible]	160	[illegible]	180	[illegible]	200	190	170

OPERATING DATA I RPM	I I.A.S.	I MIX. FUEL	I M.P.	I ⅄	I ⅄	DENSITY ALT. (feet)	OPERATING DATA II RPM	II I.A.S.	II MIX. FUEL	II M.P.	II ⅄	II ⅄	OPERATING DATA III RPM	III I.A.S.	III MIX. FUEL	III M.P.	III ⅄	III ⅄	OPERATING DATA IV RPM	IV I.A.S.	IV MIX. FUEL	IV M.P.	IV ⅄	IV ⅄	③ DENSITY ALT. (feet)	OPERATING DATA
2300	18[illegible]	A.R.	38	413		30000																			30000	WITH 12,000 FT. SET RPM TO MAINTAIN 150 MPH. FOR HIGHER TO INCREASE [illegible] INCH MP ABOVE 20,000 FT. USE 160 MPH AS AND TO INCREASE 1 INCH MP. IF SPEED [illegible] AT DATA ADD UP TO 2000 RPM AND 25 MPH. USE A 6000 RPM AND RECOMMENDED MP'S. USE ZERO (Fuel mixture auto at or below 2100 RPM.
2300	175	A.R.	30	413		25000	2150	149	A.R.	34.5	276		2100	143	A.L.	31	242									25000	
2300	168	A.R.	34	413		20000	2150	155	A.R.	34.5	276		2100	157	A.L.	31	245		2050	150	A.L.	30	216			20000	
2300	194	A.R.	39	412		15000	2150	167	A.R.	31	265		2100	165	A.L.	30.5	238		2050	58	A.L.	30	239			15000	
2300	198	A.R.	38	413		12000	2150	173	A.R.	31	252		2100	158	A.L.	30.5	234		2050	94	A.L.	28.5	207			12000	
2300	203	A.R.	34	413		9000	2100	178	A.L.	3[illegible]	250		2100	173	A.L.	30.5	229		2050	168	A.L.	28.5	202			9000	
2300	208	A.R.	30	413		6000	2100	182	A.L.	31	252		2100	177	A.L.	30	222		2050	172	A.L.	28.5	198			6000	
2300	213	A.R.	28	413		3000	2100	196	A.L.	31	248		2050	181	A.L.	30	217		2000	174	A.L.	26	192			3000	
2300	217	A.R.	28	412		S.L.	2100	188	A.L.	30.5	238		2050	184	A.L.	30	20[illegible]		2000	177	A.L.	26	187			S.L.	

LEAD ① INDICATED [illegible] SUBSTITUTE FOR [illegible] PLANE PAYLOAD
② ALLOW 170 U.S. GALS. [illegible] FOR WARM UP TAKE-OFF AND CLIMB TO 5000 FEET ALTITUDE
RETURN FUEL RESERVE TO BE USE FUEL FROM TANKS IN THE FOLLOWING ORDER

[illegible] WEIGHT ENGINE START EVERY FOR ADDITIONAL [illegible] OPERATION DATA

BOLD NUMBERS Via Batteries
ITALIC NUMBERS Via Auxiliary
WITH AND WITHOUT AUXILIARY Fuel Tanks
Above Same Tanks Empty

[illegible] Indicated Airspeed
[illegible] Manifold Pressure (In Hg)
[illegible] Gallons Per Hour
[illegible] Approximate
[illegible] Statute
[illegible] Nautical

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

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MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 5 OF 7 SHEETS

GR WT 55,000 TO 50,000 POUNDS

EXTERNAL LOAD ITEMS
NONE

CONDITION	R.P.M.	M.P. (in Hg)	BOOST (Pounds)	IN FLIGHT (Minutes)	DURATION (In Hrs)	G.P.H.	G.P.M.
TAKE-OFF	2500	48	-	A.R.	5	636	-
EMERGENCY POWER	2500	40		A.R.	5	598	

WEIGHT AT A 1820 97

INSTRUCTIONS FOR USING CHART: Select figure in last column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read operating conditions. NOTE: (A) Allow continuous cruising at Diamond

except in emergency. (B) Cowl flaps II / 3 & 4 toward the right are recommended positions for engine cooling. (C) Manifold Pressure (M.P.) Gallons Per Hour (G.P.H.) are approximate economy values for reference. (D) For quick reference note of cruise and emergency power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO RESERVE FUEL ALLOWANCES)

I (NORMAL RATED IMAG CONT.)

FUEL U.S. GALLS. (D)	RANGE IN AIR MILES STATUTE	RANGE IN AIR MILES NAUTICAL	ALT. (in feet)	R.P.M.	I.A.S. (MPH)	MIX. TYPE	M.P. (in Hg)	G.P.H.	G.P.M.
2360	100 U.S. GALLONS NOT AVAILABLE IN FLIGHT		30000	2450	147	A.R.	32	292	
2200	1690	1465	25000	2150	158	A.R.	32	290	
2000	1510	1315	20000	2150	166	A.R.	31.5	277	
1800	1360	1175	15000	2100	173	A.L.	31	265	
1600	1210	1050	12000	2100	177	A.L.	31	260	
1400	1060	920	9000	2100	182	A.L.	31	255	
1200	910	790	6000	2100	186	A.L.	31	247	
1000	760	660	3000	2100	188	A.L.	31	240	
800	610	530	S.L.	2100	191	A.L.	30.5	233	
600	460	400							
400	300	260							
200	150	130							

II

FUEL U.S. GALLS. (D)	RANGE IN AIR MILES STATUTE	RANGE IN AIR MILES NAUTICAL	ALT. (in feet)	R.P.M.	I.A.S. (MPH)	MIX. TYPE	M.P. (in Hg)	G.P.H.	G.P.M.
2360	100 U.S. GALLONS NOT AVAILABLE IN FLIGHT		30000	2100	141	A.L.	31	251	
2200	1840	1600	25000	2100	154	A.L.	31	252	
2000	1680	1460	20000	2100	164	A.L.	31	247	
1800	1510	1310	15000	2100	171	A.L.	30.5	238	
1600	1340	1165	12000	2100	175	A.L.	30.5	231	
1400	1170	1015	9000	2050	178	A.L.	30	226	
1200	1010	875	6000	2050	181	A.L.	30	219	
1000	840	730	3000	2050	183	A.L.	30	211	
800	670	580	S.L.	2050	185	A.L.	30	204	
600	500	435							
400	340	295							
200	170	145							

III

FUEL U.S. GALLS. (D)	RANGE IN AIR MILES STATUTE	RANGE IN AIR MILES NAUTICAL	ALT. (in feet)	R.P.M.	I.A.S. (MPH)	MIX. TYPE	M.P. (in Hg)	G.P.H.	G.P.M.
2360	100 U.S. GALLONS NOT AVAILABLE IN FLIGHT		30000						
2050	1740	1515	25000	2050	147	A.L.	30	219	
1870	1590	1385	20000	2050	157	A.L.	30	212	
1800	1530	1330	15000	2050	164	A.L.	30	205	
1600	1360	1185	12000	2050	168	A.L.	29.5	200	
1400	1190	1035	9000	2000	171	A.L.	29	193	
1200	1020	890	6000	2000	174	A.L.	29	188	
1000	850	740	3000	2000	175	A.L.	29	180	
800	680	590	S.L.	2000	175	A.L.	29	173	
600	510	445							
400	340	295							
200	170	145							

V (MAX RANGE)

FUEL U.S. GALLS. (D)	RANGE IN AIR MILES STATUTE	RANGE IN AIR MILES NAUTICAL	ALT. (in feet)	R.P.M.	I.A.S. (MPH)	MIX. TYPE	M.P. (in Hg)	G.P.H.	G.P.M.
2350			30000						
2200	2270	1975	25000						
2000	2070	1800	20000						
1800	1850	1610	15000						
1600	1650	1435	12000						
1400	1450	1260	9000						
1200	1240	1080	6000						
1000	1030	895	3000						
800	830	720	S.L.						
600	620	540							
400	420	365							
200	210	180							

NOTES:
(D) ENRICHING MIXTURE REQUIRED FOR HIGH AIR TEMPERATURE
(D) ALLOW 100 U.S. GALLS. FOR WARM UP
TAKE-OFF AND CLIMB TO 5000 FEET ALTITUDE
RETURN FUEL PRESENT TO TANK
USE FUEL FROM TANKS IN THE FOLLOWING ORDER

SEE OPERATING INSTRUCTIONS FOR ADDITIONAL ENGINE OPERATION DATA

Flight Operation Chart (no external load) 7 Sheets

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MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 0 OF 7 SHEETS

GR WT 50,000 TO 45,000 POUNDS

EXTERNAL LOAD ITEMS
NONE

CONDITION	R.P.M.	M.P. in. HG.	MIXTURE POSITION	PROPELLER POSITION	DURATION hr. & min.	U.S. GAL. PER HR.	TAS MPH
MAX. CONT. POWER	2500	46	-	A.R.	5	806	-
MAX. EMERG. POWER	2500	46	-	A.R.	5	806	-

Engine # R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel available. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. NOTES: (A) Avoid continuous cruising in Column I except in emergency. (B) Columns II, III, IV & V spread the right progressively give maximum in range at sacrifice in speed. (C) Manifold Pressure (M.P.) Gallons Per Hour (G.P.H.) are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES STATUTE	I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. ③	II RANGE IN AIR MILES STATUTE	II RANGE IN AIR MILES NAUTICAL	III RANGE IN AIR MILES STATUTE	III RANGE IN AIR MILES NAUTICAL	IV RANGE IN AIR MILES STATUTE	IV RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. ③	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V (MAX. RANGE) RANGE IN AIR MILES NAUTICAL
at S.L. at 25,000	at S.L. at 25,000	1782	132 U.S. GALLONS NOT AVAILABLE IN FLIGHT						1782		
1300	550	1600	1250	1090	1420	1230	1600	1395	1600	1770	1540
1020	985	1400	1100	960	1250	1084	1400	1225	1400	1550	1340
880	770										
770	670	1200	940	810	1070	930	1200	1040	1200	1330	1150
640	550	1000	780	670	880	770	1000	870	1000	1110	960
510	440	800	630	546	710	610	800	700	800	890	770
380	330	600	470	400	530	460	600	520	600	680	590
260	220	400	310	270	380	330	400	340	400	440	380
130	110	200	150	130	180	165	200	170	200	220	190

OPERATING DATA R.P.M.	I.A.S. MPH	MIX. POS.	M.P. in. Hg	G.P.H. U.S. GAL.	T.A.S. MPH	① HEIGHT ALT. in feet
2300	178	A.R.	36	413		30000
2300	185	A.R.	35	413		25000
2300	192	A.R.	33	413		20000
2300	203	A.R.	33	413		15000
2300	206	A.R.	33	413		12000
2300	211	A.R.	30	413		9000
2300	215	A.R.	30	413		6000
2300	220	A.R.	33	413		3000
2300	226	A.R.	36	413		S.L.

OPERATING DATA R.P.M.	I.A.S. MPH	MIX. POS.	M.P. in. Hg	G.P.H. U.S. GAL.	T.A.S. MPH	① HEIGHT ALT. in feet
2150	158	A.R.	32	302		30000
2150	166	A.R.	32	290		25000
2150	173	A.R.	31.5	277		20000
2100	179	A.L.	31	264		15000
2100	184	A.L.	31	253		12000
2100	187	A.L.	31	250		9000
2100	190	A.L.	31	243		6000
2100	193	A.L.	30.5	236		3000
2100	195	A.L.	30.5	229		S.L.

OPERATING DATA R.P.M.	I.A.S. MPH	MIX. POS.	M.P. in. Hg	G.P.H. U.S. GAL.	T.A.S. MPH	① HEIGHT ALT. in feet
2100	151	A.L.	31	265		30000
2100	161	A.L.	31	247		25000
2100	170	A.L.	30.5	239		20000
2050	176	A.L.	30	229		15000
2060	179	A.L.	30	229		12000
2050	181	A.L.	30	213		9000
2050	183	A.L.	30	208		6000
2050	185	A.L.	29.5	189		3000
2000	188	A.L.	29	183		S.L.

OPERATING DATA R.P.M.	I.A.S. MPH	MIX. POS.	M.P. in. Hg	G.P.H. U.S. GAL.	T.A.S. MPH	① HEIGHT ALT. in feet
2050	144	A.L.	30	218		30000
2050	158	A.L.	30	214		25000
2050	162	A.L.	30	204		20000
2000	166	A.L.	29	195		15000
2000	170	A.L.	29	188		12000
2000	172	A.L.	29	181		9000
1800	175	A.L.	29	176		6000
1800	176	A.L.	28	160		3000
1850	178	A.L.	29	181		S.L.

OPERATING DATA R.P.M.	I.A.S. MPH	MIX. POS.	M.P. in. Hg	G.P.H. U.S. GAL.	T.A.S. MPH	① HEIGHT ALT. in feet
BELOW 25,000 FT. SET RPM TO MAINTAIN 150 MPH IAS AFTER THEN MP. ABOVE 30,000 FT. USE 180 MPH AND 29 in/sq. ft. MAP. IF SPEED DROPS BELOW 150 MPH IN 2300 RPM AND 28 INCHES USE FIGHTER RPM'S AND RICHTHROTTLED MP'S. USE MANUAL MIXTURE WHEN AT OR ABOVE FIGHTER RPM.						30000
						25000
						20000
						15000
						12000
						9000
						6000
SHORTS SHOWN ABOVE APPLY AT 10 25,000 FT. ONLY.						3000
						S.L.

① INDICATED ALTITUDE CORRECTED FOR FREE A-A TEMPERATURE

② ALLOW 132 U.S. GALS. FOR TAKE-OFF AND CLIMB TO 5000. FEET ALTITUDE

RETURN FUEL FLOWS TO TANK

USE I.E. FUEL TANKS IN THE FOLLOWING ORDER

BOLD NUMBERS: Use Auxiliary

LIGHT NUMBERS: Use Main Tanks

WHEN TWO SIZES SHOWN, Use High Volume Tanks First

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (no external load) 7 Sheets

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MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 7 OF 7 SHEETS

GR WT 45,000 TO 40,000 POUNDS

EXTERNAL LOAD ITEMS
NONE

CONDITION	R.P.M.	M.P.	BOOST	MIXTURE	PROPELLER	C.S.	I.M.
TAKE-OFF	2600	45	-	A.R.	5	BOR	-
EMERGENCY RATING	2600	46	-	A.R.	5	BOR	-

Ground Idle R 1E2C 07

INSTRUCTIONS FOR USING CHART: Select figure in the column next to the left which is equal to or less than total amount of fuel to be used. Move horizontally to the right to left and select a figure equal to or greater than the altitude to be flown. Vertically below and opposite desired cruising altitude read approximate cruising conditions. NOTE: (a) Avoid continuous cruising in Column I except in emergency. (b) Columns II, III, IV & V, toward the right progressively give increase in range at sacrifice in speed. (c) Manifold Pressure (M.P.) Gallons Per Hour (G.P.H.) are approximate maximum values for reference. (d) For specific performance, tests of aircraft and engines are recommended.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED MAX. CONT.

RANGE IN AIR MILES

FUEL U.S. GALS. (1)	STATUTE	NAUTICAL	ALT.	R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P.	C.S.	I.M.
			30000	2300	179	A.R.	36	413	
			25000	2300	186	A.R.	36	413	
			20000	2300	198	A.R.	36	413	
			15000	2300	208	A.R.	38	413	
			12000	2300	209	A.R.	38	413	
			9000	2300	214	A.R.	38	413	
			6000	2300	217	A.R.	38	413	
			3000	2300	221	A.R.	38	413	
			S.L.	2300	226	A.R.	38	413	

II

RANGE IN AIR MILES

FUEL U.S. GALS. (1)	STATUTE	NAUTICAL	ALT.	R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P.	C.S.	I.M.
1000	860	750	30000	2160	159	A.L.	32	281	
800	690	590	25000	2100	165	A.L.	31	297	
600	520	450	20000	2100	173	A.L.	31	286	
400	340	290	15000	2100	180	A.L.	31	295	
200	170	150	12000	2100	183	A.L.	30.5	239	
			9000	2100	188	A.L.	30.5	231	
			6000	2050	189	A.L.	30	223	
			3000	2050	191	A.L.	30	215	
			S.L.	2050	193	A.L.	30	209	

III

RANGE IN AIR MILES

STATUTE	NAUTICAL	ALT.	R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P.	C.S.	I.M.
960	840	30000	2100	185	A.L.	31	245	
770	670	25000	2100	193	A.L.	30.5	235	
680	590	20000	2050	170	A.L.	30	224	
380	330	15000	2050	176	A.L.	30	214	
190	170	12000	2050	178	A.L.	30	207	
		9000	2000	180	A.L.	29	198	
		6000	2000	182	A.L.	29	193	
		3000	2000	184	A.L.	29	188	
		S.L.	1960	186	A.L.	29	179	

IV

RANGE IN AIR MILES

FUEL U.S. GALS. (1)	STATUTE	NAUTICAL	ALT.	R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P.	C.S.	I.M.
1060	920	30000	2050	160	A.L.	30	215		
850	740	25000	2050	168	A.L.	30	208		
650	570	20000	2000	154	A.L.	29	199		
420	370	15000	2000	158	A.L.	29	185		
210	180	12000	1950	170	A.L.	29	178		
		9000	1900	172	A.L.	29	171		
		6000	1900	173	A.L.	29	165		
		3000	1800	173	A.L.	29	158		
		S.L.	1800	173	A.L.	29	151		

V (MAX. RANGE)

RANGE IN AIR MILES

FUEL U.S. GALS. (1)	STATUTE	NAUTICAL	ALT.	R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P.	C.S.	I.M.
1000	1190		30000						
800	980		25000						
600	770		20000						
400	560		15000						
200	330		12000						
			9000						
			6000						
			3000						
			S.L.						

(1) INCLUDES 45 GALS. CORRECTED FOR TAXI AND WARM-UP

(2) ALLOW 0.5 GALS. PER MIN. FOR TAXI AND WARM-UP

1 STATUTE MILE EQUALS 0.87 NAUTICAL MILE

1 NAUTICAL MILE EQUALS 1.15 STATUTE MILES

USE AUTO-RICH LEVER IN THE NORMAL ONES

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart: no external load: 7 Sheets

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MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 1 OF 3 SHEETS

GR. WT. 65,000 TO 60,000 POUNDS

EXTERNAL LOAD ITEMS
(2) 2000 LB. BOMBS

CONDITION	R.P.M.	M.A.P.	AUTOMATIC RESTRICTIONS	MANUAL RESTRICTIONS	MIXTURE CONTROL	I.A.S.	M.P.
TAKE-OFF	2600	48	-	A.R.	5	80B	-
MILITARY POWER	2500	46	-	A.R.	5	80B	-

ENGINE M R-1820-97

INSTRUCTIONS FOR USING CHART Select figure in last column which is equal to or less than total amount of fuel in gallons. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Normally below and opposite desired cruising altitude read operation cruising conditions. NOTES: (A) Avoid continuous cruising in Column I except in emergency. (B) Columns II, III, IV & V toward the right progressively more available in range at increased speed. (C) Manifold Pressure (M.P.) & Gallons Per Hour (G.P.H.) are approximate maximum values for reference. (D) For quick reference take-off and military power data outlined in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

FUEL U.S. GALS. (B)	I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES STATUTE	I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES NAUTICAL	II RANGE IN AIR MILES STATUTE	II RANGE IN AIR MILES NAUTICAL	III RANGE IN AIR MILES STATUTE	III RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. (B)	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V (MAX. RANGE) RANGE IN AIR MILES NAUTICAL
2282			162 U.S. GALLONS NOT AVAILABLE IN FLIGHT				2282		
2100	1140	990	1280	1110	1410	1220	2100	1720	1490
2000	1050	910	1200	1040	1340	1160	2000	1630	1410
1800	950	820	1080	940	1210	1050	1800	1470	1270
1600	840	730	960	830	1090	940	1600	1310	1130
1400	740	640	840	730	940	810	1400	1150	990
1200	630	550	720	620	810	700	1200	980	850
1000	530	460	600	520	670	580	1000	820	710
800	420	370	480	420	540	470	800	650	560
600	320	280	360	310	400	350	600	490	420
400	210	180	240	210	270	230	400	330	280
200	100	90	120	100	130	110	200	160	140

OPERATING DATA							(C) PRESS. ALT.	OPERATING DATA							OPERATING DATA							OPERATING DATA							(C) PRESS. ALT.	OPERATING DATA						
R.P.M.	I.A.S.	MIX. CONT.	M.P.	G.P.H.	T.A.S.	I.M.P.		R.P.M.	I.A.S.	MIX. CONT.	M.P.	G.P.H.	T.A.S.	I.M.P.	R.P.M.	I.A.S.	MIX. CONT.	M.P.	G.P.H.	T.A.S.	I.M.P.	R.P.M.	I.A.S.	MIX. CONT.	M.P.	G.P.H.	T.A.S.	I.M.P.		R.P.M.	I.A.S.	MIX. CONT.	M.P.	G.P.H.	T.A.S.	I.M.P.
2300		A.R.	39	413			30000																						30000						
2300	168	A.R.	38	413			25000																						25000						
2300	172	A.R.	36	413			20000	2200	148	A.R.	32.5	310																	20000						
2300	179	A.R.	38	413			15000	2200	159	A.R.	32.5	307			2100	150	A.L.	31	255			2050	142	A.L.	30	218			15000						
2300	181	A.R.	38	413			12000	2150	164	A.R.	32	304			2100	156	A.L.	31	255			2050	147	A.L.	30	218			12000						
2300	188	A.R.	38	413			9000	2150	168	A.R.	32	298			2100	162	A.L.	31	253			2050	154	A.L.	30	217			9000						
2300	192	A.R.	38	413			6000	2150	172	A.R.	32	290			2100	166	A.L.	31	250			2050	160	A.L.	30	216			6000						
2300	199	A.R.	36	413			3000	2150	175	A.R.	31.5	286			2100	170	A.L.	31	248			2050	164	A.L.	30	212			3000						
2300	203	A.R.	38	413			S.L.	2150	179	A.R.	31.5	275			2100	174	A.L.	30.5	238			2050	168	A.L.	30	208			S.L.						

(C) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE.

(B) ALLOW 182 U.S. GALS. FOR CLIMB UP TAKE-OFF AND CLIMB TO 5000 FEET ALTITUDE

RETURN FUEL FLOWS TO TANK

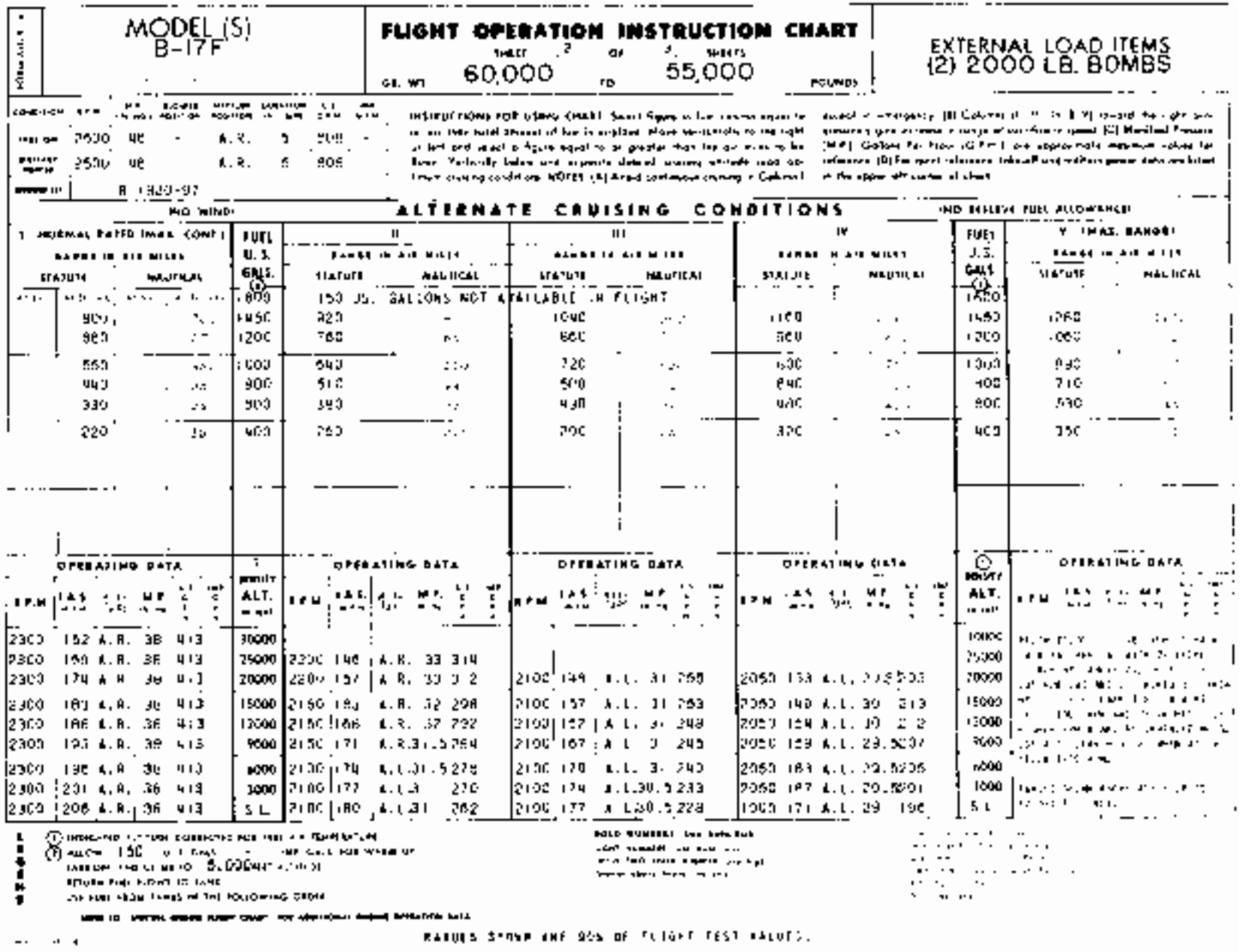
USE FUEL FROM TANKS IN THE FOLLOWING ORDER

SEE TO "SPECIFIC ENGINE FLIGHT CHART" FOR ADDITIONAL ENGINE OPERATION DATA.

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (external load - two 2000-pound bombs) 3 Sheets

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MODEL (S)
B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 2 OF 3 SHEETS

GR. WT 60,000 TO 55,000 POUNDS

EXTERNAL LOAD ITEMS
(2) 2000 LB. BOMBS

CONDITION | R.P.M. | M.P. | BOOST ADJUSTMENT | MIXTURE POSITION | DURATION IN MIN. | I.A.S. | ALT.
TAKE OFF | 2600 | 48 | - | A.R. | 5 | 2000 | -
EMERGENCY WAR | 2500 | 48 | | A.R. | 5 | 2000 |

ENGINE(S) R-1820-97

INSTRUCTIONS FOR USING CHART: Speed figures in the various columns refer to an indicated airspeed of two miles per hour. Where necessary to read right or left and select a figure equal to or greater than the one given to be sure. Vertically below and opposite desired cruising altitude read cruise control cruising condition. NOTES: (A) Avoid continuous cruising in Columns I and II between 1000 and 1500 R.P.M. (B) Columns II, III, IV and V record the right and wrong way to cruise at various altitudes and speeds. (C) Manifold Pressure (M.P.) Gallons Per Hour (G.P.H.) are approximate maximum values for reference. (D) For more extensive take-off and emergency power data see latest of the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

NO WIND

I (NORMAL RATED IMEP. CONT.) RANGE IN AIR MILES STATUTE	NAUTICAL	FUEL U.S. GALS. (A)	II RANGE IN AIR MILES STATUTE	NAUTICAL	III RANGE IN AIR MILES STATUTE	NAUTICAL	IV RANGE IN AIR MILES STATUTE	NAUTICAL	FUEL U.S. GALS. (A)	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	NAUTICAL
		800	150 GAL. GALLONS NOT AVAILABLE IN FLIGHT						1400		
900		1450	920	-	1040		1160		1450	1260	
980		1200	780		860		960		1200	1060	
860		1000	640		720		800		1000	860	
940		800	510		570		640		800	710	
880		600	390		430		480		600	530	
220		400	260		290		320		400	350	

OPERATING DATA

R.P.M.	I.A.S.	A.L.	M.P.	G.P.H.	H.P.	DENSITY ALT.	R.P.M.	I.A.S.	A.L.	M.P.	G.P.H.	H.P.
2300	152	A.R.	38	413		30000						
2300	159	A.R.	38	413		25000	2200	146	A.R.	33	314	
2300	174	A.R.	38	413		20000	2200	157	A.R.	33	312	
2300	181	A.R.	38	413		15000	2150	163	A.R.	32	298	
2300	186	A.R.	38	413		12000	2150	166	A.R.	32	292	
2300	193	A.R.	38	413		9000	2150	171	A.R.	31.5	284	
2300	198	A.R.	38	413		6000	2100	174	A.L.	31.5	278	
2300	201	A.R.	38	413		3000	2100	177	A.L.	31	270	
2300	206	A.R.	38	413		S.L.	2100	180	A.L.	31	262	

R.P.M.	I.A.S.	A.L.	M.P.	G.P.H.	H.P.
2100	149	A.L.	31	268	
2100	157	A.L.	31	263	
2100	167	A.L.	31	248	
2100	167	A.L.	31	245	
2100	170	A.L.	31	240	
2100	174	A.L.	30.5	233	
2100	177	A.L.	30.5	228	

R.P.M.	I.A.S.	A.L.	M.P.	G.P.H.	H.P.
2050	153	A.L.	29.5	203	
2050	160	A.L.	30	213	
2050	164	A.L.	30	212	
2050	169	A.L.	29.5	207	
2050	183	A.L.	29.5	206	
2050	187	A.L.	29.5	201	
2050	191	A.L.	29	196	

DENSITY ALT.

R.P.M.	I.A.S.	A.L.	M.P.	G.P.H.	H.P.
10000					
25000					
20000					
15000					
12000					
9000					
6000					
3000					
S.L.					

(A) ENGINE AND FUEL CONSUMPTION CORRECTIONS FOR TEMPERATURE

(B) ALLOW 150 U.S. GALS. FOR WARM UP

TAKE-OFF GROSS WT. 60,000 LBS.

RETURN FUEL ALLOWANCE

FOR FUEL FROM TANKS OF THE FOLLOWING GROUP

BOLD NUMBERS

VALUES SHOWN ARE 95% OF FLIGHT TEST VALUES.

Flight Operation Chart (external load - two 2000-pound bombs) 3 Sheets

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MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 3 OF 3 SHEETS

GR WT. 55,000 TO 50,000

EXTERNAL LOAD ITEMS (2) 2000 LB. BOMBS

	R.P.M.	M.P.	BOOST POSITION	MIXTURE POSITION	PROPELLER CONTROL SETTING IN. MM.	OIL OUT °C	OIL OUT °F
TAKE-OFF	2500	48	-	A.R.	5	80A	-
MILITARY POWER	2500	48	-	A.R.	5	80B	-

ENGINE No. R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in last column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. NOTES: (A) Avoid continuous cruising in Column I except in emergency. (B) Columns II, III, IV & V to avoid the right propeller only gate increase in range at specific airspeed. (C) Manifold Pressure (M.P.) Gauges For Mass (G.P.M.) are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I (NORMAL RATED I.M.E.P. CONT.) RANGE IN AIR MILES		FUEL U.S. GALS. ①	II RANGE IN AIR MILES		III RANGE IN AIR MILES		IV RANGE IN AIR MILES		FUEL U.S. GALS. ①	V (MAX. RANGE) RANGE IN AIR MILES	
STATUTE	NAUTICAL		STATUTE	NAUTICAL	STATUTE	NAUTICAL	STATUTE	NAUTICAL		STATUTE	NAUTICAL
350	[illegible]	520	410	[illegible]	470	[illegible]	530	[illegible]	620	580	[illegible]
230	[illegible]	400	270	[illegible]	300	[illegible]	340	[illegible]	400	380	[illegible]
120	[illegible]	200	130	[illegible]	150	[illegible]	170	[illegible]	200	180	[illegible]

OPERATING DATA							② BOOST ALT. IN. MM.	OPERATING DATA							OPERATING DATA							OPERATING DATA							② BOOST ALT. IN. MM.	OPERATING DATA						
R.P.M.	I.A.S. M.P.H.	HIGH/LOW	M.P. IN. MM.	°C	°F		R.P.M.	I.A.S. M.P.H.	HIGH/LOW	M.P. IN. MM.	°C	°F		R.P.M.	I.A.S. M.P.H.	HIGH/LOW	M.P. IN. MM.	°C	°F		R.P.M.	I.A.S. M.P.H.	HIGH/LOW	M.P. IN. MM.	°C	°F		R.P.M.	I.A.S. M.P.H.	HIGH/LOW	M.P. IN. MM.	°C	°F	
2300	158	A.R.	38	413			30000	2200	141	A.R.	33	316																				
2300	170	A.R.	38	413			25000	2200	154	A.R.	33	318		2100	144	A.L.	31	267		2050	137	A.L.	30	220			25000	[illegible]				
2300	160	A.R.	38	413			20000	2200	161	A.R.	32	306		2100	153	A.L.	31	258		2050	147	A.L.	30	248			20000	[illegible]				
2300	188	A.R.	38	413			15000	2150	156	A.R.	32	261		2100	161	A.L.	31	248		2050	151	A.L.	30	215			15000	[illegible]				
2300	191	A.R.	32	413			12000	2150	170	A.R.	31	.5283		2100	165	A.L.	31	241		2050	158	A.L.	30	208			12000	[illegible]				
2300	195	A.R.	38	413			9000	2150	173	A.R.	31	.5275		2100	168	A.L.	30	.5236		2050	163	A.L.	26	.5202			9000	[illegible]				
2300	186	A.R.	38	413			6000	2100	179	A.L.	31	269		2100	173	A.L.	30	.5231		2050	168	A.L.	26	196			6000					
2300	206	A.R.	30	413			3000	2100	190	A.L.	31	262		2050	178	A.L.	30	.5226		2000	188	A.L.	20	192			3000	[illegible]				
2300	210	A.R.	38	413			S.L.	2100	184	A.L.	31	250		2050	190	A.L.	30	218		2000	172	A.L.	20	187			S.L.	[illegible]				

NOTES:

① INDICATES ALTITUDE CORRECTION FOR FUEL AT TEMPERATURE

② ALLOW [illegible] U.S. GALS. [illegible] FOR GALS FOR WARM UP TAKE-OFF AND CLIMB TO [illegible] FEET ALTITUDE

RETURN FUEL TO BE [illegible] GALS.

USE FUEL NOW TAKEN IN THE FOLLOWING ORDER: [illegible]

REFER TO "ENGINE OPERATING CHART" FOR ADDITIONAL ENGINE OPERATING DATA

BOLD NUMBERS ARE ALTITUDES AT WHICH POWER OR ALTITUDE MIXTURE CONTROL SHOULD BE USED

I.A.S. = INDICATED AIRSPEED

[illegible]

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (external load - two 2000 - pound bombs) 3 Sheets

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FORM ASC-511A

MODEL (S)
B-17 F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 1 OF 2 SHEETS

GR. WT. 65,000 TO 60,000 POUNDS

EXTERNAL LOAD ITEMS
(2) 4000 LB. BOMBS

CONDITION	R.P.M.	M.P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	DURATION IN MIN.	U.S. G.P.H.	IMP. G.P.H.
TAKE-OFF	2500	46	-	A.R.	5	608	-
MILITARY POWER	2500	46	-	A.R.	5	608	-
ENGINE (S)	R-1820-97						

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. *NOTES:* (A) Avoid continuous cruising in Column I except in emergency. (B) Columns (II, III, IV & V) toward the right progressively give increase in range at sacrifice in speed. (C) Manifold Pressure (M.P.), Gallons Per Hour (G.P.H.), are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

(NO WIND) **ALTERNATE CRUISING CONDITIONS** (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES STATUTE AT S.L.	I STATUTE AT 25,000	I NAUTICAL AT S.L.	I NAUTICAL AT 25,000	FUEL U.S. GALS. (2)	II RANGE IN AIR MILES STATUTE	II NAUTICAL	III RANGE IN AIR MILES STATUTE	III NAUTICAL	IV RANGE IN AIR MILES STATUTE	IV NAUTICAL	FUEL U.S. GALS. (2)	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V NAUTICAL
				1732	132 U.S. GALLONS NOT AVAILABLE IN FLIGHT						1732		
	790		690	1600	910	790	1040	900			1600	1170	1020
	690		600	1400	800	700	910	790			1400	1030	900
	590		510	1200	690	600	780	680			1200	880	770
	490		430	1000	570	500	650	570			1000	730	640
	400		350	800	460	400	520	450			800	590	510
	300		260	600	350	300	390	340			600	440	380
	200		170	400	230	200	260	230			400	300	260

OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	(1) DENSITY ALT. IN FEET	II OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	III OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	IV OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	(1) DENSITY ALT. IN FEET	V OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. (IN. Hg)	U.S. G.P.H.	IMP. G.P.H.
						30000																			30000	BELOW 20,000 FT. SET RPM TO MAINTAIN 145 MPH IAS WITH 29± INCH MP. ABOVE 20,000 FT. USE 135 MPH IAS AND 29 INCHES ± 1 INCH MP. IF SPEED CANNOT BE OBTAINED UP TO 2000 RPM AND 29 INCHES, USE HIGHER RPM'S AND RECOMMENDED MP'S. USE AUTO-LEAN MIXTURE WHEN AT OR BELOW 2100 RPM. RANGES SHOWN ABOVE APPLY UP TO 6,000 FT. ONLY.					
2300	150	A.R.	38	413		25000																			25000						
2300	162	A.R.	38	413		20000																			20000						
2300	171	A.R.	38	413		15000	2150	149	A.R.	32	299														15000						
2300	175	A.R.	38	413		12000	2150	154	A.R.	32	295		2100	145	A.L.	31	241								12000						
2300	179	A.R.	38	413		9000	2150	157	A.R.	31.5	290		2100	141	A.L.	31	241								9000						
2300	183	A.R.	38	413		6000	2150	160	A.R.	31.5	283		2100	156	A.L.	31	239								6000						
2300	189	A.R.	38	413		3000	2150	164	A.R.	31	276		2100	161	A.L.	30.5	236								3000						
2300	194	A.R.	38	413		S.L.	2150	168	A.R.	31	270		2100	166	A.L.	30.5	234								S.L.						

LEGEND

(1) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE.
(2) ALLOW 132 U. S. GALS. - IMP. GALS. FOR WARM UP.
TAKE-OFF AND CLIMB TO 5000 FEET ALTITUDE
RETURN FUEL FLOWS TO TANK
USE FUEL FROM TANKS IN THE FOLLOWING ORDER

BOLD NUMBERS: Use Auto-Rich
LIGHT NUMBERS: Use Auto-Lean
WITH TWO SPEED BLOWER: Use high blower above heavy line only

I.A.S.: Indicated Air Speed
M.P.: Manifold Pressure (In. Hg)
U.S.G.P.H.: U. S. Gallons Per Hour
IMP.G.P.H.: Imperial Gallons Per Hour
F.T.: Full Throttle
S.L.: Sea Level

REFER TO "SPECIFIC ENGINE FLIGHT CHART" FOR ADDITIONAL ENGINE OPERATION DATA.

WF-1-1-43-3M

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (external load - two 4000- pound bombs) 2 Sheets

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FORM ASC-511A

MODEL (S) B-17F

FLIGHT OPERATION INSTRUCTION CHART

SHEET 2 OF 2 SHEETS

GR. WT. 60,000 TO 55,000 POUNDS

EXTERNAL LOAD ITEMS
(2) 4000 LB. BOMBS

CONDITION	R.P.M.	M.P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	DURATION IN MIN.	U.S. G.P.H.	IMP. G.P.H.
TAKE-OFF	2500	46	-	A.R.	5	608	-
MILITARY POWER	2500	46	-	A.R.	5	608	-

ENGINE (S): R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. **NOTES:** (A) Avoid continuous cruising in Column I except in emergency. (B) Columns (II, III, IV & V) toward the right progressively give increase in range at sacrifice in speed. (C) Manifold Pressure (M.P.), Gallons Per Hour (G.P.H.), are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES				FUEL U.S. GALS. (2)	II RANGE IN AIR MILES		III RANGE IN AIR MILES		IV RANGE IN AIR MILES		FUEL U.S. GALS. (2)	V (MAX. RANGE) RANGE IN AIR MILES	
STATUTE		NAUTICAL			STATUTE	NAUTICAL	STATUTE	NAUTICAL	STATUTE	NAUTICAL		STATUTE	NAUTICAL
AT S.L.	AT 25,000	AT S.L.	AT 25,000										
	410		360	800	470	410	530	460	590	510	800	650	560
	310		270	600	350	300	400	350	440	380	600	490	430
	210		180	400	240	210	270	230	300	260	400	320	280
	100		90	200	120	100	130	110	150	130	200	160	140

OPERATING DATA						(1) DENSITY ALT. IN FEET	OPERATING DATA						OPERATING DATA						OPERATING DATA						(1) DENSITY ALT. IN FEET	OPERATING DATA					
R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.		R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.		R.P.M.	I.A.S. M.P.H.	MIX-TURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.
						30000																			30000	BELOW 20,000 FT. SET RPM TO MAINTAIN 145 MPH IAS WITH 29 INCHES ± 1 INCH MP. ABOVE 20,000 FT. USE 135 MPH IAS AND 29 INCHES ± 1 INCH MP. IF SPEED CANNOT BE OBTAINED UP TO 2000 RPM AND 29 INCHES, USE HIGHER RPM'S AND RECOMMENDED MP'S. USE AUTO-LEAN MIXTURE WHEN AT OR BELOW 2100 RPM. RANGES SHOWN ABOVE APPLY UP TO 6000 FT. ONLY.
						25000	2200	140	A.R.	33	320														25000					
2300	156	A.R.	38	413		20000	2200	148	A.R.	33	316														20000					
2300	166	A.R.	38	413		15000	2150	156	A.R.	32.5	306		2100	148	A.L.	31	259		2050	144	A.L.	30.5	224		15000					
2300	176	A.R.	38	413		12000	2150	160	A.R.	32	300		2100	153	A.L.	31	256		2050	148	A.L.	30	220		12000					
2300	179	A.R.	38	413		9000	2150	163	A.R.	32	293		2100	158	A.L.	31	251		2050	152	A.L.	30	218		9000					
2300	184	A.R.	38	413		6000	2150	166	A.R.	31.5	285		2100	162	A.L.	31	248		2050	156	A.L.	30	213		6000					
2300	187	A.R.	38	413		3000	2100	169	A.L.	31.5	277		2100	166	A.L.	31	242		2050	160	A.L.	30	210		3000					
2300	192	A.R.	38	413		S.L.	2100	172	A.L.	31	270		2100	170	A.L.	31	237		2050	164	A.L.	30	206		S.L.					
2300	196	A.R.	38	413																										

LEGEND

(1) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE.

(2) ALLOW _____ - _____ U. S. GALS. _____ - _____ IMP. GALS. FOR WARM UP.
TAKE-OFF AND CLIMB TO _____ - _____ FEET ALTITUDE
RETURN FUEL FLOWS TO TANK _____
USE FUEL FROM TANKS IN THE FOLLOWING ORDER _____

BOLD NUMBERS: Use Auto-Rich
LIGHT NUMBERS: Use Auto-Lean
WITH TWO SPEED BLOWER: Use high blower above heavy line only

I.A.S.: Indicated Air Speed
M.P.: Manifold Pressure (In. Hg)
U.S.G.P.H.: U. S. Gallons Per Hour
IMP.G.P.H.: Imperial Gallons Per Hour
F.T.: Full Throttle
S.L.: Sea Level

REFER TO "SPECIFIC ENGINE FLIGHT CHART" FOR ADDITIONAL ENGINE OPERATION DATA.

WF-1-1-43-5M

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (external load - two 4000- pound bombs) 2 Sheets

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MODEL (S) B-17F 3 ENGINE OPERATION

FLIGHT OPERATION INSTRUCTION CHART

SHEET 1 OF 4 SHEETS

GR. WT. 60,000 TO 55,000 POUNDS

EXTERNAL LOAD ITEMS: 1 FEATHERED PROPELLER

CONDITION	R.P.M.	M.P.	[illegible]	MIXTURE POSITION	[illegible]	[illegible]	[illegible]
TAKE-OFF	2500	46	-	A.R.	5	458	-
MILITARY POWER	2500	46	-	A.R.	5	458	-

ENGINE(S): R-1820-97

INSTRUCTIONS FOR USING CHART: [illegible]

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) Statute	I Nautical	FUEL U.S. GALS. (2)	II Statute	II Nautical	III Statute	III Nautical	IV Statute	IV Nautical	FUEL U.S. GALS. (2)	V (MAX. RANGE) Statute	V Nautical
		2770	170 U.S. GALLONS NOT AVAILABLE IN FLIGHT.						2770		
1550	[illegible]	2800	1720	[illegible]					2800	1730	[illegible]
1430	[illegible]	2400	1590	[illegible]					2400	1750	[illegible]
1310	[illegible]	2200	1460	[illegible]					2200	1600	[illegible]
1190	[illegible]	2000	1320	[illegible]					2000	1480	[illegible]
1070	[illegible]	1800	1190	[illegible]					1800	1310	[illegible]
950	[illegible]	1600	1080	[illegible]					1600	1170	[illegible]
840	[illegible]	1400	530	[illegible]					1400	1020	[illegible]
720	[illegible]	1200	800	[illegible]					1200	880	[illegible]
600	[illegible]	1000	650	[illegible]					1000	730	[illegible]
480	[illegible]	800	530	[illegible]					800	580	[illegible]

OPERATING DATA

I R.P.M.	I.A.S. M.P.H.	MIX. TURE	M.P. IN. HG.	[illegible]	[illegible]	PRESS. ALT. (1) FEET	II R.P.M.	I.A.S. M.P.H.	MIX. TURE	M.P. IN. HG.	[illegible]
						30000					
						25000					
						20000					
2300	159	A.R.	38	310		15000					
2300	165	A.R.	38	310		12000					
2300	171	A.R.	38	310		9000					
2300	178	A.R.	38	310		6000	2150	152	A.R.	37.5	230
2300	181	A.R.	38	310		3000	2150	157	A.R.	32	227
2300	185	A.R.	38	310		S.L.	2150	152	A.R.	31.5	211

Columns III, IV and V (R.P.M., I.A.S., MIX-TURE, M.P.): no values legible.

V notes (altitude column 30000 … S.L.): [illegible] … RANGES SHOWN ABOVE APPLY UP TO [illegible]

NOTES: (1) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE. (2) ALLOW 170 U.S. GALS. [illegible] GALS. FOR WARM-UP, TAKE-OFF AND CLIMB TO 5000 FEET ALTITUDE [illegible]

BOLD NUMBERS [illegible]

REFER TO [illegible] FOR ADDITIONAL [illegible] DATA

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES

Flight Operation Chart (one propeller feathered; 4 Sheets

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MODEL (S) B-17F 3 ENGINE OPERATION

FLIGHT OPERATION INSTRUCTION CHART

SHEET 2 OF 11 SHEETS

GR. WT. 55,000 TO 50,000 POUNDS

EXTERNAL LOAD ITEMS
I FEATHERED PROPELLER

CONDITION	R.P.M.	M.P. (IN. HG.)	BLOWER POSITION	CARBURETOR POSITION	OUTLET SHUTTER	I.A.S. (M.P.H.)	M.P. (M.P.H.)
TAKE-OFF	2500	45	-	A.R.	5	120	
MAXIMUM CRUISE	2500	45	-	A.R.	5	156	

ENGINE # R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or next less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read operating conditions. NOTES: (A) Avoid continuous cruising in Column I except in emergency. (B) Columns II, III, IV & V should be used approximately one minute. A range of ten M.P.H. indicated speed. (C) Maximum Range (M.P.H. Gallons Per Hour (G.P.H.)) are approximate maximum values for reference. (D) Fuel allowance for take-off and warm-up data are listed in the upper left corner of chart.

NO WIND

ALTERNATE CRUISING CONDITIONS

NO RESERVE FUEL ALLOWANCE

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES STATUTE	I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. (1)	II RANGE IN AIR MILES STATUTE	II RANGE IN AIR MILES NAUTICAL	III RANGE IN AIR MILES STATUTE	III RANGE IN AIR MILES NAUTICAL	IV RANGE IN AIR MILES STATUTE	IV RANGE IN AIR MILES NAUTICAL	FUEL U.S. GALS. (2)	V (MAX. RANGE) RANGE IN AIR MILES STATUTE	V (MAX. RANGE) RANGE IN AIR MILES NAUTICAL
		2850	150 U.S. GALLONS NOT AVAILABLE IN FLIGHT.						2300		
1400		2800	1560		1710				2200	1900	
1270		2000	1420		1620				2000	1730	
1140		1800	1280		1420				1800	1560	
1020		1600	1140		1260				1600	1360	
890		1400	1000		1100				1400	1210	
780		1200	860		960				1200	1040	
640		1000	710		790				1000	870	
510		800	570		630				800	690	
380		600	430		470				600	520	
280		400	290		320				400	350	
130		200	140		160				200	170	

OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIT. TUBE	M.P. IN. HG.	G.P.H. U.S. GALS.	MIXT. CONTROL	(3) MINIMUM ALT. IN M.P.	OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIT. TUBE	M.P. IN. HG.	G.P.H. U.S. GALS.	MIXT. CONTROL	OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIT. TUBE	M.P. IN. HG.	G.P.H. U.S. GALS.	MIXT. CONTROL	OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIT. TUBE	M.P. IN. HG.	G.P.H. U.S. GALS.	MIXT. CONTROL	(3) MINIMUM ALT. IN M.P.	OPERATING DATA R.P.M.	I.A.S. M.P.H.	MIT. TUBE	M.P. IN. HG.	G.P.H. U.S. GALS.	MIXT. CONTROL
2300	-	A.R.	38	810		30000																			30000					
2300	139	A.R.	38	810		25000																			25000					
2300	156	A.R.	36	810		20000																			20000					
2300	168	A.R.	38	810		15000	2200	148	A.R.	33	839														15000					
2300	171	A.R.	38	810		12000	2200	152	A.R.	33	836														12000					
2300	178	A.R.	38	810		9000	2150	157	A.R.	32.5	832		2100	140	A.L.	31	191								9000					
2300	182	A.R.	38	810		6000	2150	160	A.R.	32	827		2100	150	A.L.	31	192								6000					
2300	188	A.R.	38	810		3000	2150	163	A.R.	32	821		2100	155	A.L.	31	188								3000					
2300	200	A.R.	38	810		S.L.	2150	164	A.R.	31	813		2100	158	A.L.	31	185								S.L.					

NOTES

(1) INDICATED ALTITUDE CORRECTED FOR ARI AIR TEMPERATURE

(2) ALLOW 160 U.S. GALS. FOR WARM UP, TAKE OFF AND CLIMB TO 5000 FEET ALTITUDE. 150 GALS. FOR WARM UP, TAKE OFF AND CLIMB TO 5000 FEET ALTITUDE. RETURN FUEL FROM ELEVATED TANKS. USE FUEL FROM TANKS IN THE FOLLOWING ORDER

(3) REFER TO SPECIAL FLIGHT CHART FOR ADDITIONAL ENGINE OPERATING DATA

ADJUSTMENT NUMBERS for Autolean (Set number for desired mixture then spin rotor knob to that number)

Flight Operation Chart (one propeller feathered) 4 Sheets

RESULTS SHOWN ARE 90% OF FLIGHT TEST VALUES.

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MODEL (S) B-17F
3 ENGINE OPERATION

FLIGHT OPERATION INSTRUCTION CHART

SHEET 3 OF 4 SHEETS

GR WT 50,000 TO 45,000 POUNDS

EXTERNAL LOAD ITEMS
1 FEATHERED PROPELLER

CONDITION	R.P.M.	M.P.	BOOST	MIXTURE	FUEL GALS.	OIL GALS.	CYL. TEMP.
TAKE-OFF	2500	48	-	A.R.	5	456	-
MAXIMUM CRUISE	2500	49	-	A.R.	5	456	-

ENGINES: R-1820-97

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right on that and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read operating data. NOTES: (A) Avoid continuous cruising in Lateral range of engine speeds. (B) Columns II, III, IV & V) Ground the right propeller in emergency. (C) Operate in range of lowest engine speed (C) Manifold Pressure (M.P.) Column For Hour (G.P.H.) are approximate maximum values for reference. (D) For detailed climb data, take-off and military power data see text in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND)

(NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.)

RANGE IN AIR MILES

STATUTE	NAUTICAL	FUEL U.S. GALS. (2)
1170	[illegible]	1732
1080	[illegible]	1600
950	[illegible]	1400
810	[illegible]	1200
680	[illegible]	1000
540	[illegible]	800
410	[illegible]	600
270	[illegible]	400
140	120	200

OPERATING DATA

R.P.M.	I.A.S.	H.T. (Hdg)	M.P.	G.P.H.	(1) HEIGHT ALT.
2300	151	A.R.	38	310	30000
2300	163	A.R.	38	310	25000
2300	171	A.R.	38	340	20000
2300	178	A.R.	38	340	15000
2300	183	A.R.	38	310	12000
2300	187	A.R.	38	310	9000
2300	192	A.R.	38	310	6000
2300	196	A.R.	38	310	3000
2300	196	A.R.	38	310	S.L.

II

RANGE IN AIR MILES

STATUTE	NAUTICAL
132 U.S. GALLONS NOT AVAILABLE IN FLIGHT.	
1220	[illegible]
1070	[illegible]
920	[illegible]
760	[illegible]
610	[illegible]
490	[illegible]
300	[illegible]
150	[illegible]

OPERATING DATA

R.P.M.	I.A.S.	H.T. (Hdg)	M.P.	G.P.H.	(1) HEIGHT ALT.
2150	151	A.R.	32.5	230	15000
2150	167	A.R.	32.5	228	12000
2150	181	A.R.	32	224	9000
2150	183	A.R.	32	216	6000
2150	187	A.R.	31	211	3000
2150	191	A.R.	31	208	S.L.

III

RANGE IN AIR MILES

STATUTE	NAUTICAL
1060	[illegible]
1190	[illegible]
1020	[illegible]
850	[illegible]
690	[illegible]
510	[illegible]
340	[illegible]
170	[illegible]

OPERATING DATA

R.P.M.	I.A.S.	H.T. (Hdg)	M.P.	G.P.H.	(1) HEIGHT ALT.
2100	141	A.L.	31	193	15000
2100	148	A.L.	31	193	12000
2100	155	A.L.	31	193	9000
2100	160	A.L.	31	189	6000
2100	164	A.L.	31	185	3000
2100	168	A.L.	31	182	S.L.

IV

RANGE IN AIR MILES

STATUTE	NAUTICAL
1500	[illegible]
1320	1260
1130	980
940	[illegible]
750	[illegible]
570	[illegible]
380	[illegible]
190	[illegible]

OPERATING DATA

R.P.M.	I.A.S.	H.T. (Hdg)	M.P.	G.P.H.	(1) HEIGHT ALT.
2050	141	A.L.	30	156	15000
2050	148	A.L.	30	156	12000
2050	155	A.L.	30	156	9000
2050	158	A.L.	30	163	6000
2050	162	A.L.	30	159	3000
2050	162	A.L.	30	159	S.L.

V (MAX. RANGE)

RANGE IN AIR MILES

STATUTE	NAUTICAL	FUEL U.S. GALS. (2)
		1732
1540	[illegible]	1600
1440	[illegible]	1400
		1200
		1000
		800
		600
		400
		200

OPERATING DATA

R.P.M.	I.A.S.	H.T. (Hdg)	M.P.	G.P.H.	(1) HEIGHT ALT.
					30000
					25000
					20000
					15000
					12000
					9000
					6000
					3000
					S.L.

NOTES:

(1) HEIGHTS WITHOUT CORRECTION FOR AIR TEMPERATURE

(2) ALLOW 132 U.S. GALS. [illegible] GAL. FOR WARM UP TAKE OFF AND CLIMB TO 5,000 FEET ALTITUDE BEFORE YOU BEGIN TO TAKE YOU FUEL FROM TANKS IN THE FOLLOWING ORDER

[illegible] FLIGHT CHART FOR APPROXIMATE [illegible] DATA

BOLD NUMBERS ARE RANGES [illegible]

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (one propeller feathered) 4 Sheets

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FORM ASC-511A

MODEL (S) B-17F
3 ENGINE OPERATION

FLIGHT OPERATION INSTRUCTION CHART

SHEET 4 OF 4 SHEETS

GR. WT. 45,000 TO 40,000 POUNDS

EXTERNAL LOAD ITEMS
1 FEATHERED PROPELLER

CONDITION	R.P.M.	M.P. (IN. HG.)	BLOWER POSITION	MIXTURE POSITION	DURATION IN MIN.	U.S. G.P.H.	IMP. G.P.H.
TAKE-OFF	2500	46	-	A.R.	5	456	-
MILITARY POWER	2500	46	-	A.R.	5	456	-
ENGINE (S)	R-1820-97						

INSTRUCTIONS FOR USING CHART: Select figure in fuel column equal to or less than total amount of fuel in airplane. Move horizontally to the right or left and select a figure equal to or greater than the air miles to be flown. Vertically below and opposite desired cruising altitude read optimum cruising conditions. *NOTES:* (A) Avoid continuous cruising in Column I except in emergency. (B) Columns (II, III, IV & V) toward the right progressively give increase in range at sacrifice in speed. (C) Manifold Pressure (M.P.), Gallons Per Hour (G.P.H.), are approximate maximum values for reference. (D) For quick reference, take-off and military power data are listed in the upper left corner of chart.

ALTERNATE CRUISING CONDITIONS

(NO WIND) (NO RESERVE FUEL ALLOWANCE)

I NORMAL RATED (MAX. CONT.) RANGE IN AIR MILES				FUEL U.S. GALS. (2)	II RANGE IN AIR MILES		III RANGE IN AIR MILES		IV RANGE IN AIR MILES		FUEL U.S. GALS. (2)	V (MAX. RANGE) RANGE IN AIR MILES	
STATUTE		NAUTICAL			STATUTE	NAUTICAL	STATUTE	NAUTICAL	STATUTE	NAUTICAL		STATUTE	NAUTICAL
AT S.L.	AT 25,000	AT S.L.	AT 25,000										
	700		610	1000	880	770	980	830	1050	910	1000	1140	990
	560		490	800	700	610	770	670	840	730	800	910	790
	430		370	600	540	470	590	510	640	560	600	690	600
	280		240	400	350	300	340	300	420	370	400	450	390
	140		120	200	180	160	170	160	210	180	200	230	200

OPERATING DATA						(1) DENSITY ALT. IN FEET	OPERATING DATA						OPERATING DATA						OPERATING DATA						(1) DENSITY ALT. IN FEET	OPERATING DATA					
R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.		R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN. Hg	U.S. G.P.H.	IMP. G.P.H.	R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN Hg	U.S. G.P.H.	IMP. G.P.H.								R.P.M.	I.A.S. M.P.H.	MIXTURE	M.P. IN Hg	U.S. G.P.H.	IMP. G.P.H.
2300	144	A.R.	38	310		30000																			30000	BELOW 20,000 FT. SET RPM TO MAINTAIN 145 MPH IAS WITH 29± 1 INCH MP. ABOVE 20,000 USE 135 MPH IAS AND 29 ± 1 INCH MP. IF SPEED CANNOT BE OBTAINED UP TO 2000 RPM AND 29 INCHES USE HIGHER RPM'S AND RECOMMENDED MP'S. USE AUTO-LEAN MIXTURE WHEN AT OR BELOW 2100 RPM. RANGES SHOWN ABOVE APPLY UP TO 10,000 FT. ONLY.					
2300	159	A.R.	38	310		25000																			25000						
2300	169	A.R.	38	310		20000																			20000						
2300	178	A.R.	38	310		15000	2100	149	A.L.	31	199		2100	148	A.L.	31	180		2050	144	A.L.	30	160		15000						
2300	183	A.R.	38	310		12000	2100	155	A.L.	31	196		2100	154	A.L.	31	178		2050	148	A.L.	30	157		12000						
2300	186	A.R.	38	310		9000	2100	161	A.L.	31	194		2100	159	A.L.	31	175		2050	153	A.L.	29.5	155		9000						
2300	190	A.R.	38	310		6000	2100	165	A.L.	31	191		2050	162	A.L.	30.5	171		2000	155	A.L.	29.5	150		6000						
2300	194	A.R.	38	310		3000	2100	169	A.L.	31	186		2050	165	A.L.	30	166		2000	158	A.L.	29	146		3000						
2300	198	A.R.	38	310		S.L.	2100	173	A.L.	31	182		2050	168	A.L.	30	161		2000	161	A.L.	29	142		S.L.						

3 ENGINE OPERATION

LEGEND

(1) INDICATED ALTITUDE CORRECTED FOR FREE AIR TEMPERATURE.

(2) ALLOW ___ - ___ U. S. GALS. ___ - ___ IMP. GALS. FOR WARM UP, TAKE-OFF AND CLIMB TO ___ - ___ FEET ALTITUDE

RETURN FUEL FLOWS TO TANK ___

USE FUEL FROM TANKS IN THE FOLLOWING ORDER ___

BOLD NUMBERS: Use Auto-Rich

LIGHT NUMBERS Use Auto-Lean

WITH TWO SPEED BLOWER Use high blower above heavy line only

I.A.S. Indicated Air Speed
M.P. Manifold Pressure (In. Hg.)
U.S.G.P.H. U.S. Gallons Per Hour
IMP.G.P.H. Imperial Gallons Per Hour
F.T. Full Throttle
S.L. Sea Level

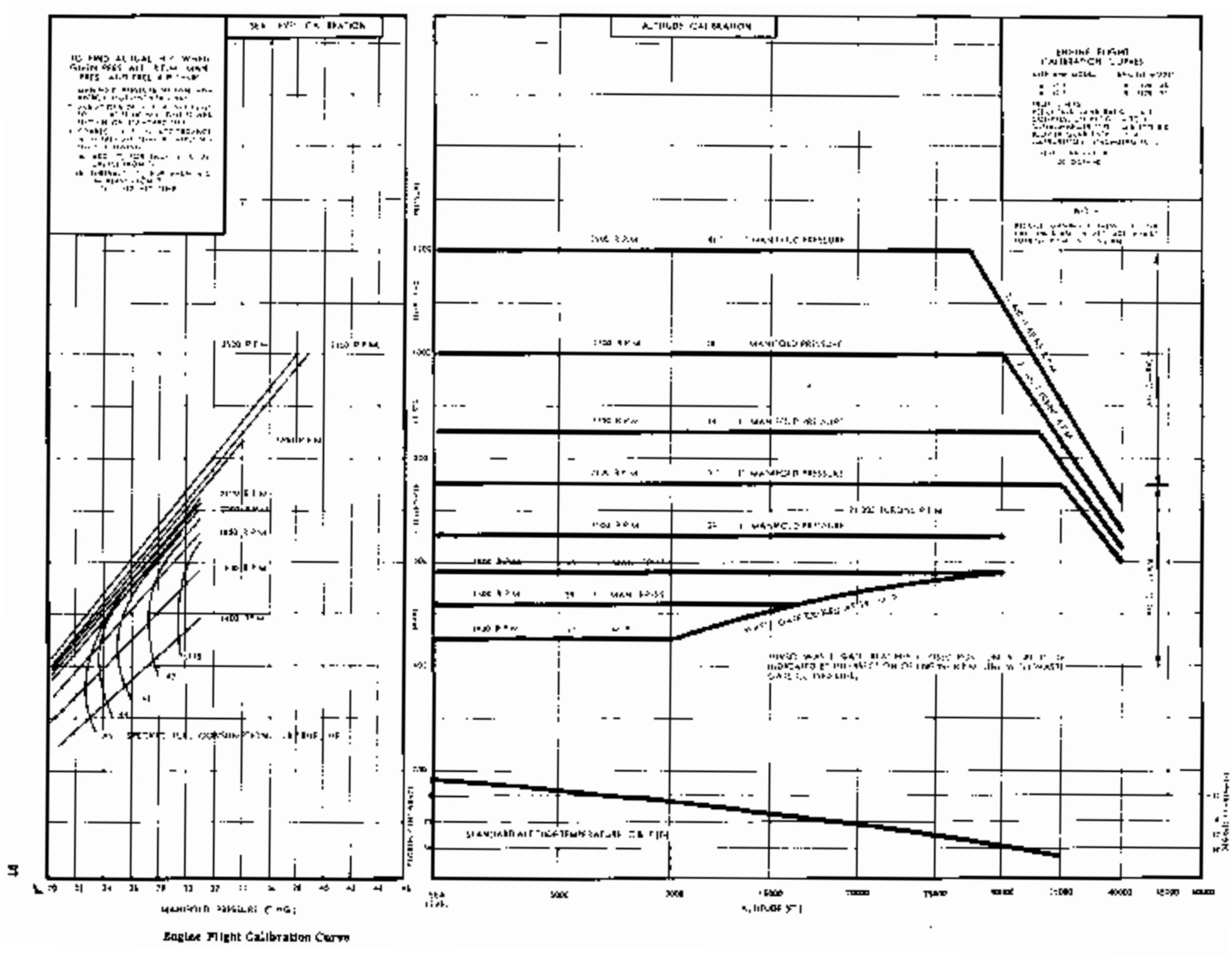
REFER TO "SPECIFIC ENGINE FLIGHT CHART" FOR ADDITIONAL ENGINE OPERATION DATA.

WI-1-1-44-5M

RANGES SHOWN ARE 90% OF FLIGHT TEST VALUES.

Flight Operation Chart (one propeller feathered) 4 Sheets

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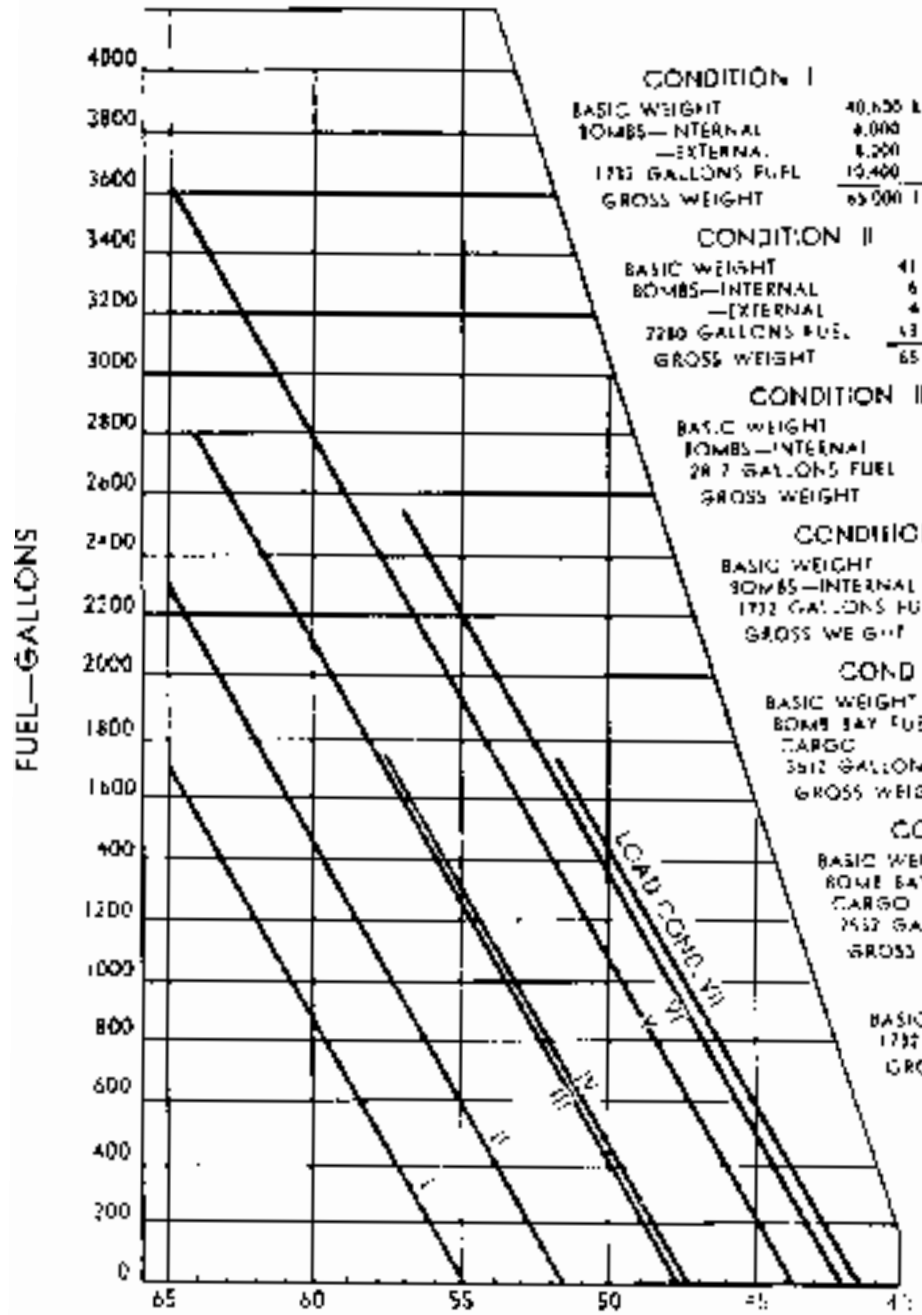


Engine Flight Calibration Curve

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LOAD CONDITIONS INCLUDE IN BASIC WEIGHT:

CREW OF NINE
NINE 50 CALIBER GUNS
3500 ROUNDS AMMUNITION EXCEPT I = 1170 ROUNDS
900 LBS. MISCELLANEOUS EQUIPMENT
144 GALLONS OIL
1500 LBS. EXTRA WING TANKS IN
CONDITIONS I, II, III, IV, V, AND VII.



Loading Chart

CONDITION I

BASIC WEIGHT	40,650 LBS
BOMBS—INTERNAL	6,000
—EXTERNAL	6,000
1732 GALLONS FUEL	10,400
GROSS WEIGHT	65,000 LBS

CONDITION II

BASIC WEIGHT	41,300 LBS.
BOMBS—INTERNAL	6,000
—EXTERNAL	4,000
2280 GALLONS FUEL	13,700
GROSS WEIGHT	65,000 LBS

CONDITION III

BASIC WEIGHT	41,100 LBS
BOMBS—INTERNAL	6,000
2817 GALLONS FUEL	16,900
GROSS WEIGHT	64,200 LBS.

CONDITION IV

BASIC WEIGHT	41,300 LBS
BOMBS—INTERNAL	6,000
1732 GALLONS FUEL	10,400
GROSS WEIGHT	57,700 LBS

CONDITION V

BASIC WEIGHT	41,300 LBS
BOMB BAY FUEL TANKS	900
CARGO	1,100
3512 GALLONS FUEL	21,700
GROSS WEIGHT	65,000 LBS

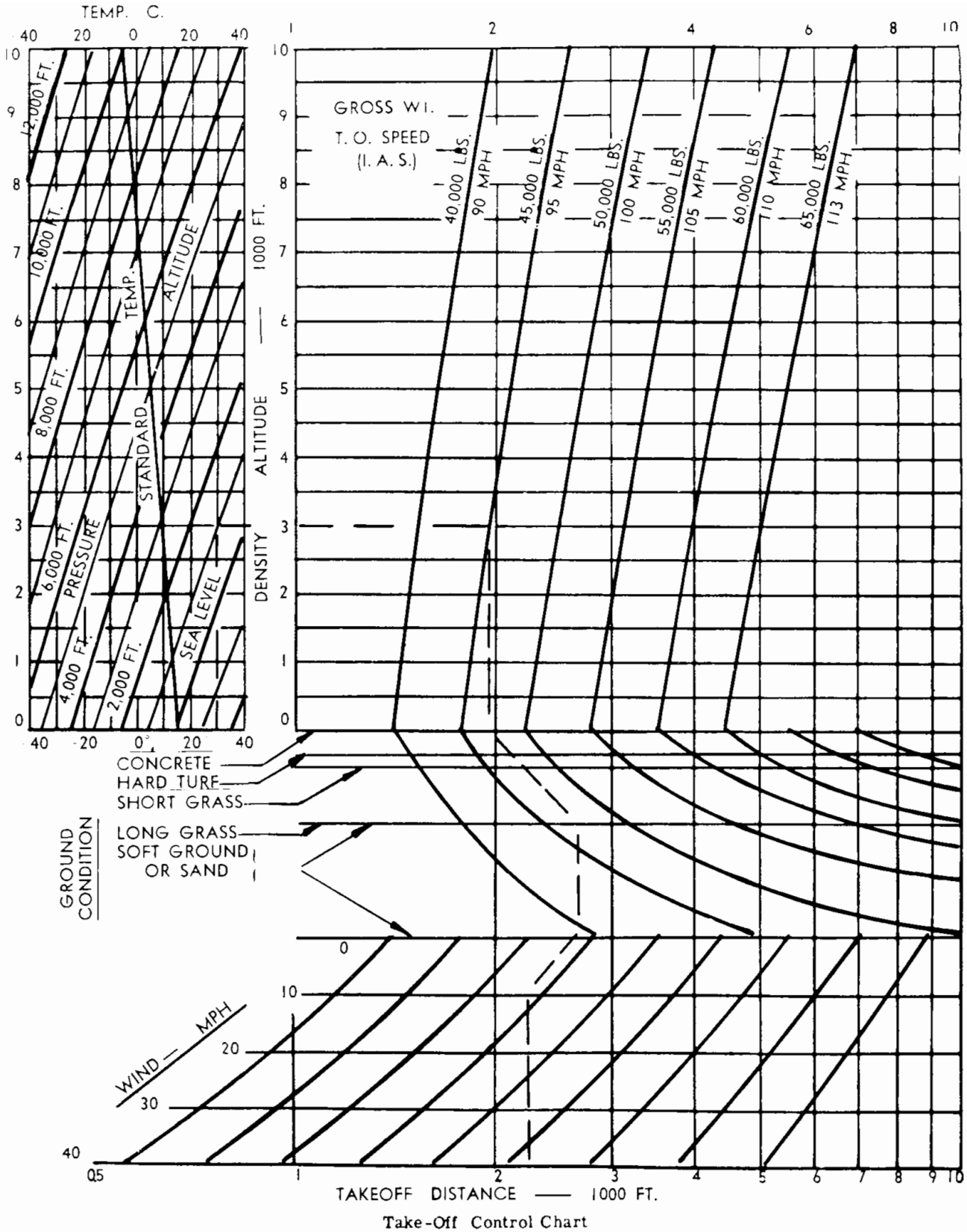
CONDITION VI

BASIC WEIGHT	39,800 LBS
BOMB BAY FUEL TANKS	900
CARGO	1,100
2552 GALLONS FUEL	15,200
GROSS WEIGHT	57,000 LBS

CONDITION VI

BASIC WEIGHT	41,300 LBS
1732 GALLONS FUEL	10,400
GROSS WEIGHT	51,700 LBS

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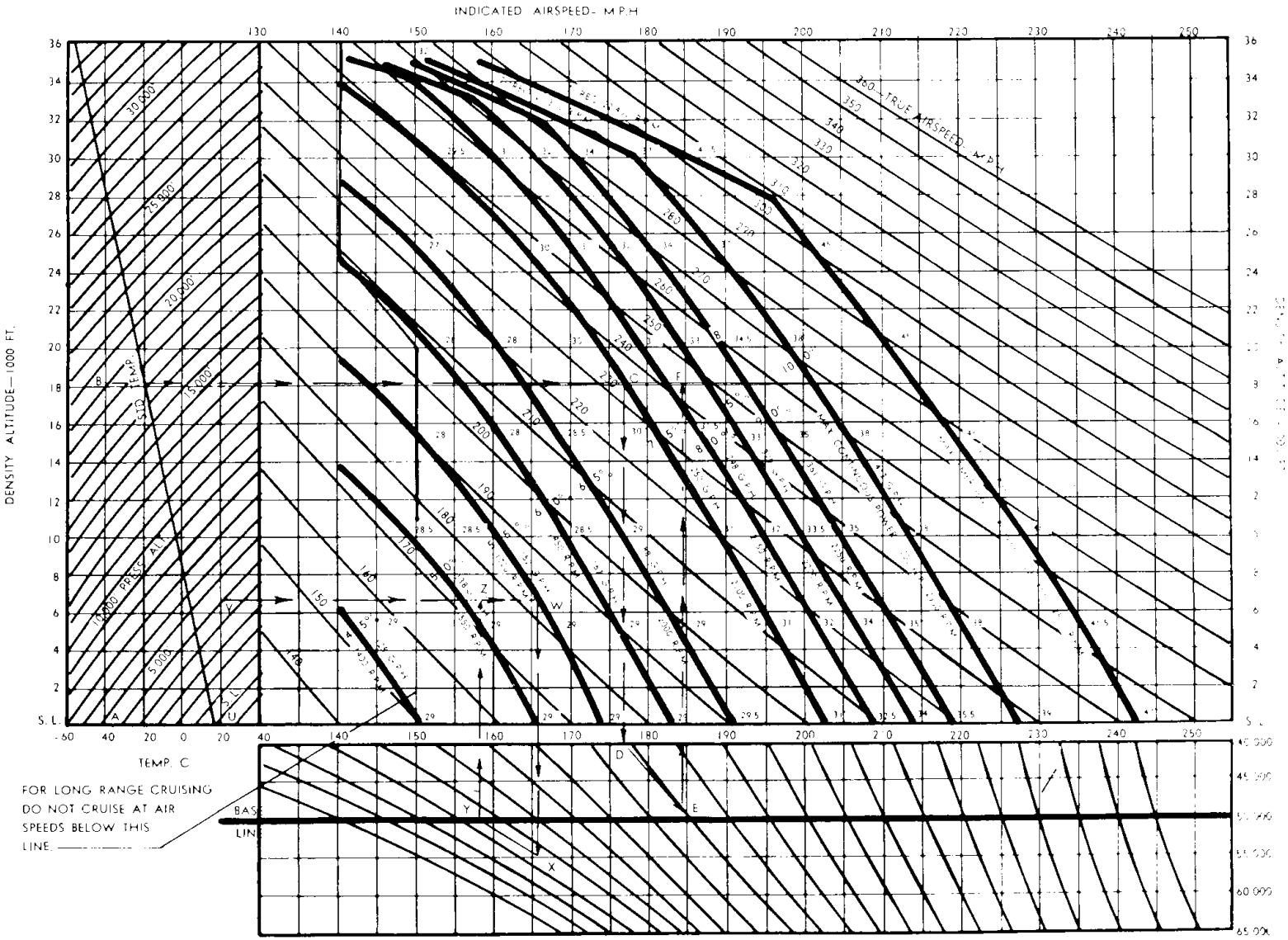


Take-Off Control Chart

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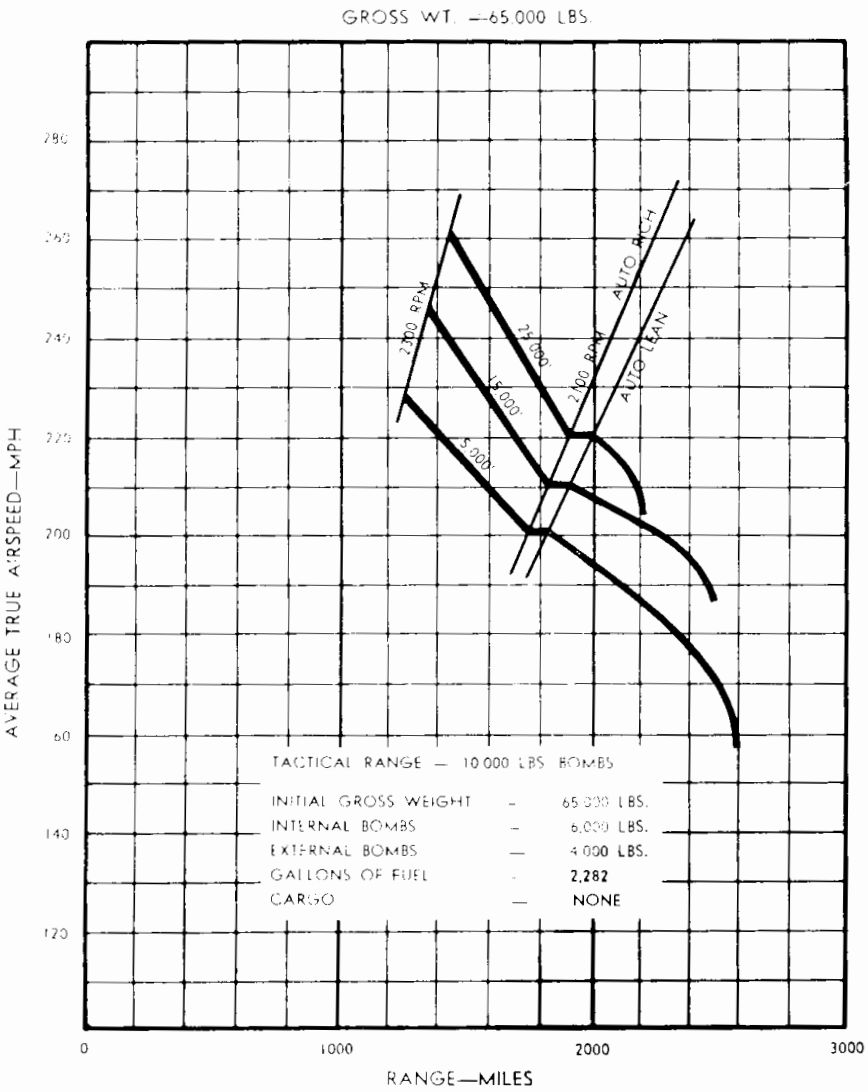
Climb Control Chart

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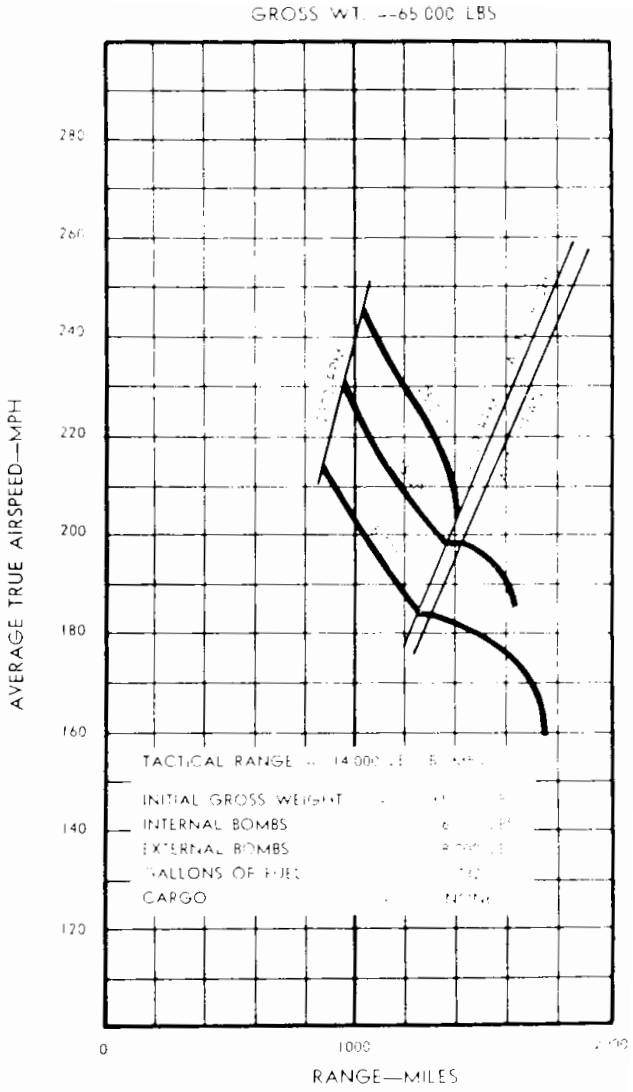
Composite Cruising Control Chart

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NOTE:

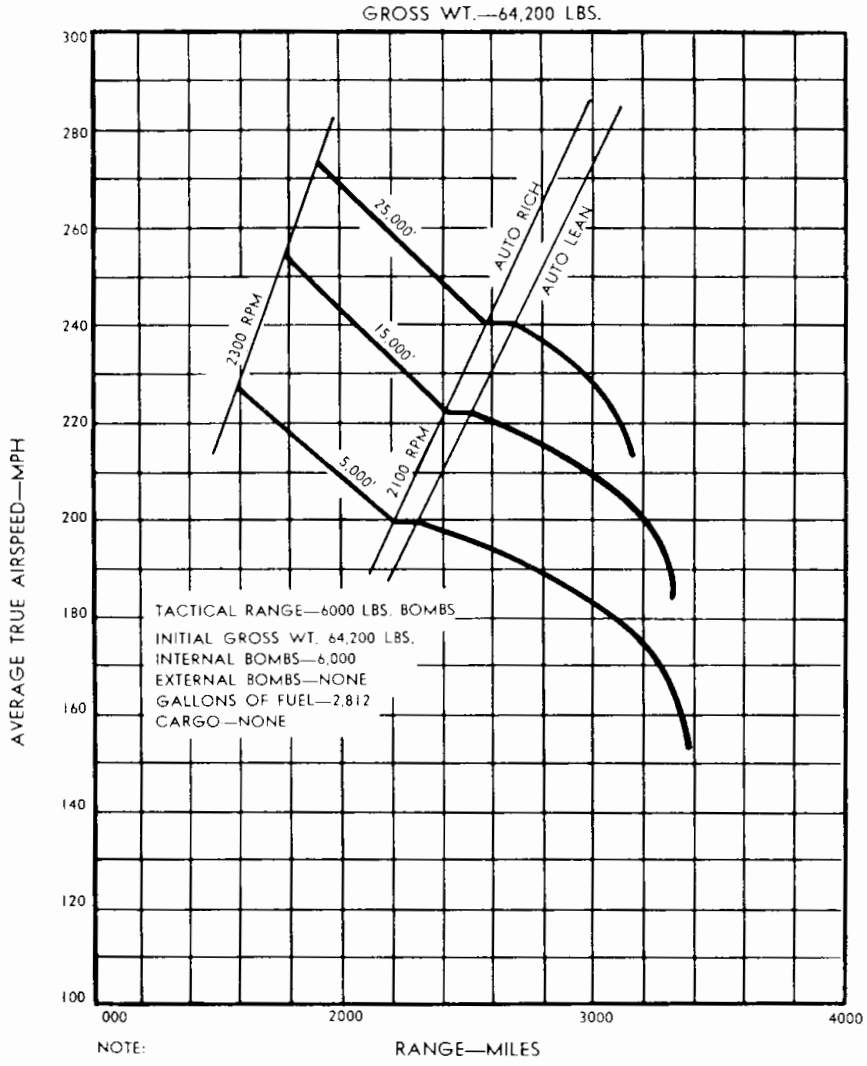
1. THESE RANGE VS. TRUE AIRSPEED CURVES SHOW ABSOLUTE RANGES AND ARE COMPUTED FROM INSTANTANEOUS CRUISING CONDITIONS OF ALTITUDE, POWER, AND FUEL FLOW.
2. NO ALLOWANCE IS MADE FOR WARMUP, TAKEOFF, CLIMB, DESCENT OR HEADWINDS.
3. BOMBS ARE CONSIDERED CARRIED HALF THE DISTANCE OF FLIGHT.

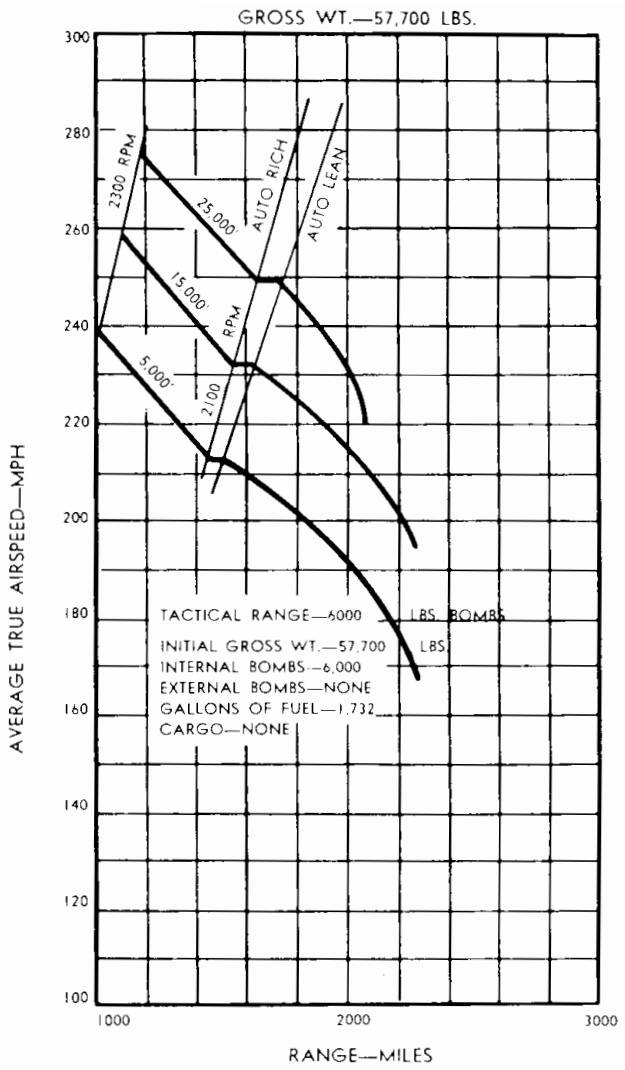


RANGE VS. AVERAGE TRUE AIRSPEED

Tactical Range Charts

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RANGE VS. AVERAGE TRUE AIRSPEED

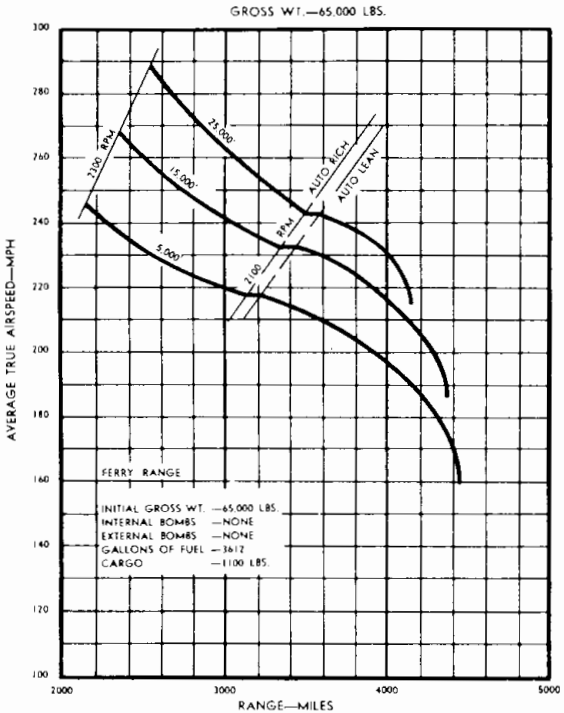
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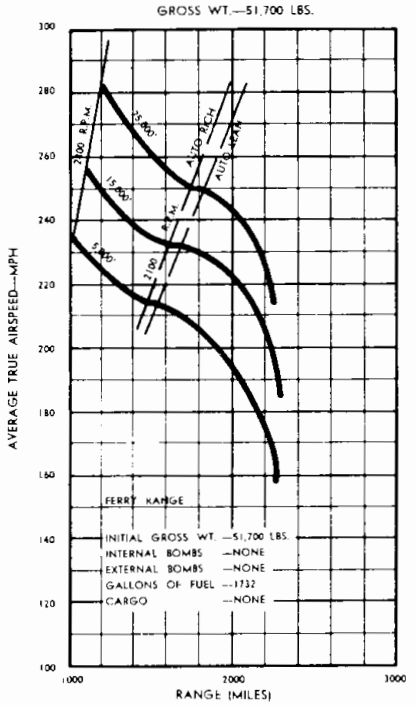
1. THESE RANGE VS. AVERAGE TRUE AIRSPEED CURVES SHOW ABSOLUTE RANGES AND ARE COMPUTED FROM INSTANTANEOUS CRUISING CONDITIONS OF ALTITUDE, POWER, AND FUEL FLOW.

2. NO ALLOWANCE IS MADE FOR WARMUP, TAKEOFF, CLIMB, DESCENT OR HEADWINDS.

3. BOMBS ARE CONSIDERED CARRIED HALF THE DISTANCE OF FLIGHT.

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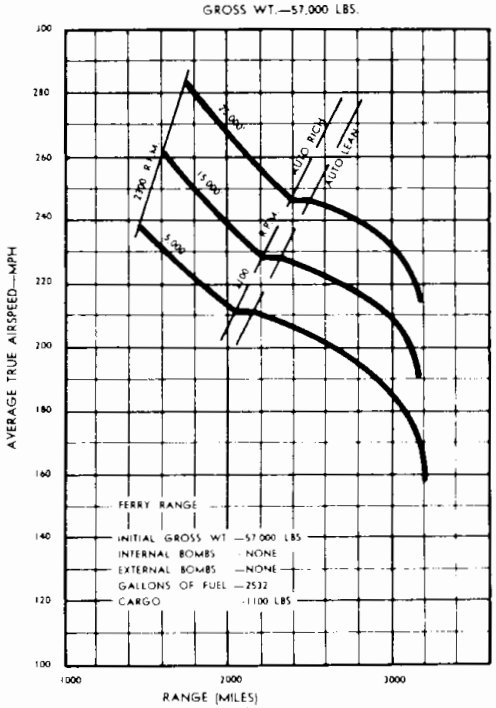


NOTE:

1. THESE RANGE VS. TRUE AIRSPEED CURVES SHOW ABSOLUTE RANGES AND ARE COMPUTED FROM INSTANTANEOUS CRUISING CONDITIONS OF ALTITUDE, POWER, AND FUEL FLOW.

2. NO ALLOWANCE IS MADE FOR WARMUP, TAKEOFF, DESCENT, OR HEADWIND.

3. BOMBS ARE CONSIDERED CARRIED HALF OF THE DISTANCE OF FLIGHT.

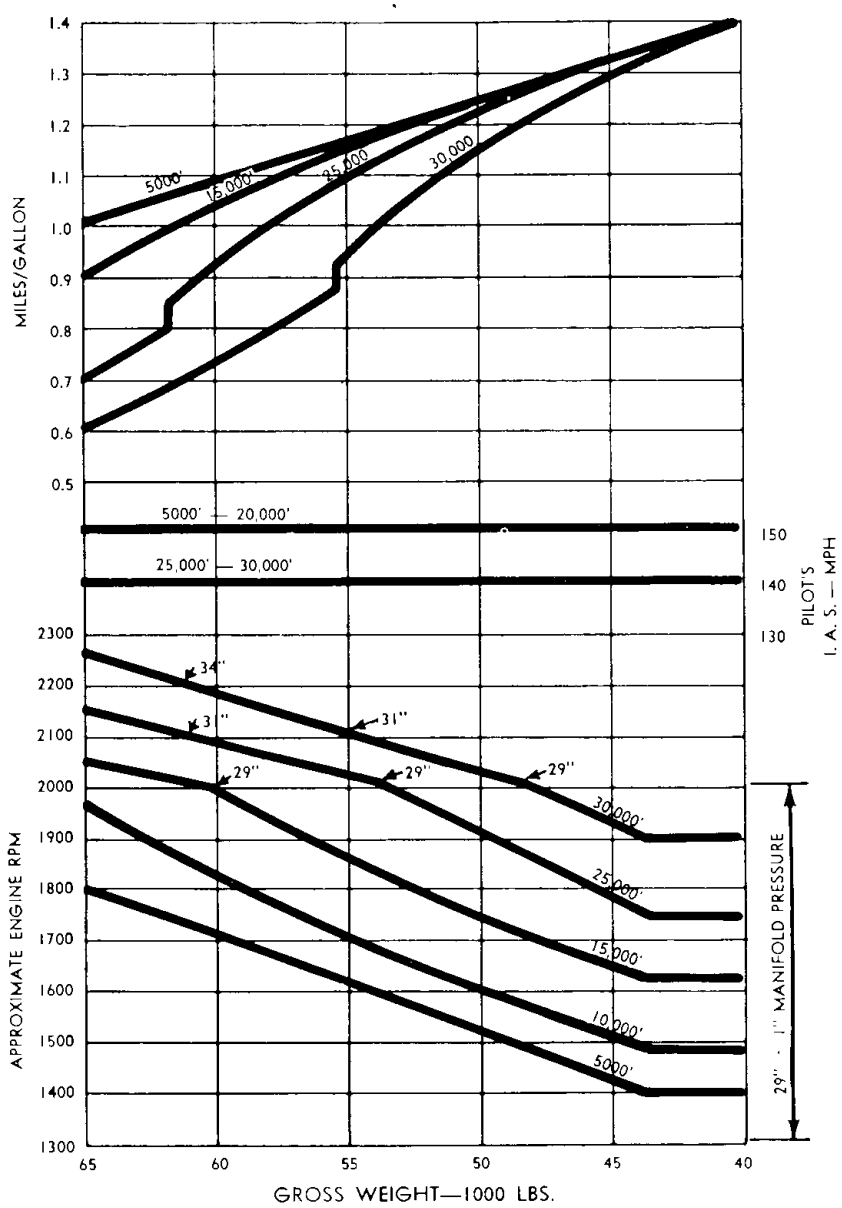


RANGE VS. AVERAGE TRUE AIRSPEED

Ferry Range Charts

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LONG RANGE CRUISING PROCEDURE

(WITH ALL ENGINES OPERATING—NO EXTERNAL BOMBS)

BELOW 20,000' SET RPM TO MAINTAIN 150 MPH PILOT'S INDICATED AIRSPEED WITH 29 INCHES - 1 INCH MANIFOLD PRESSURE ABOVE 20,000' USE 140 MPH PILOT'S INDICATED AND 29 INCHES - 1 INCH. IF SPEED CANNOT BE OBTAINED UP TO 2,000 RPM AND 29 INCHES, USE HIGHER RPM'S AND RECOMMENDED MANIFOLD PRESSURES. USE AUTO-LEAN MIXTURE WHEN AT OR BELOW 2100 RPM. CLOSE COWL FLAPS OR SET TO OBTAIN PROPER CYLINDER TEMPERATURE. HOLD POWER SETTING AND LET AIRSPEED INCREASE AS FUEL IS USED. RESET RPM EVERY THREE HOURS TO MAINTAIN DESIRED CRUISING SPEED.

LONG RANGE CRUSING PROCEDURE—

WITH ONE ENGINE OUT
OR TWO ENGINES OUT
OR TWO 2,000-LB. EXTERNAL BOMBS
OR TWO 4,000-LB. EXTERNAL BOMBS

USE SAME PROCEDURE AS ABOVE EXCEPT FLY AT 145 MPH PILOT'S INDICATED AIRSPEED BELOW 20,000 FEET AND 135 MPH ABOVE 20,000 FEET.

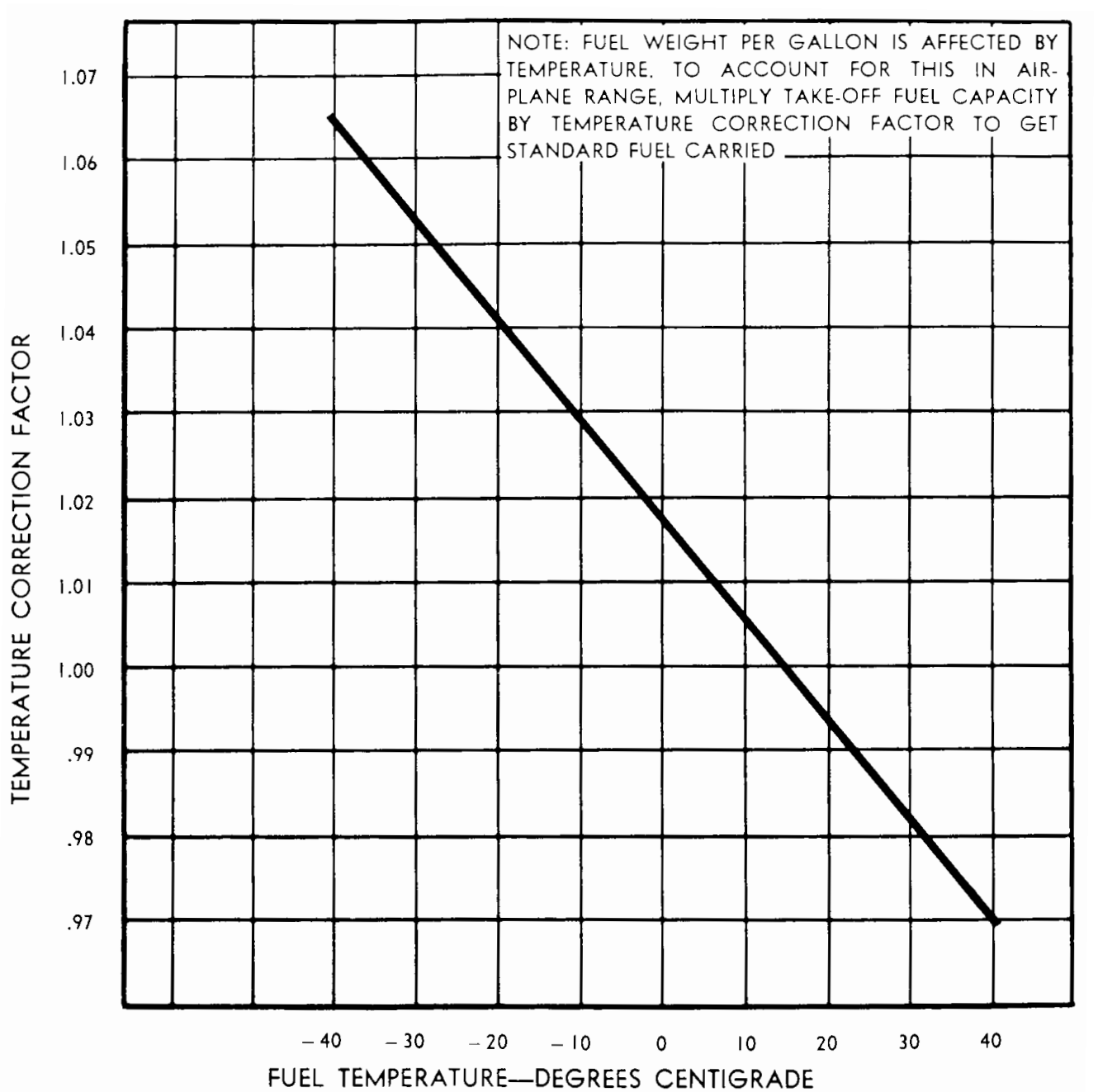
ALWAYS USE ABOVE PROCEDURES FOR LONG RANGE FLYING. VARIATIONS FROM RPM'S SHOWN CAN BE EXPECTED INASMUCH AS AIR TEMPERATURE, COWL FLAP POSITION EXTRA GUNS, EXTRA RADIO EQUIPMENT, OR OPEN SIDE WINDOWS WILL ALL AFFECT THE RPM AT WHICH THE DESIRED AIRSPEED AND MANIFOLD PRESSURE ARE OBTAINED.

PROCEDURE FOR USE OF CHART

ENTER CHART AT GROSS WEIGHT CORRESPONDING TO THAT OF AIRPLANE. PROJECT VERTICALLY TO OBTAIN APPROXIMATE RPM, PILOT'S INDICATED AIRSPEED, AND APPROXIMATE MILES PER GALLON OF FUEL. TO DETERMINE GALLONS PER HOUR OF FUEL REFER TO FUEL CONSUMPTION CHART, PAGE 31

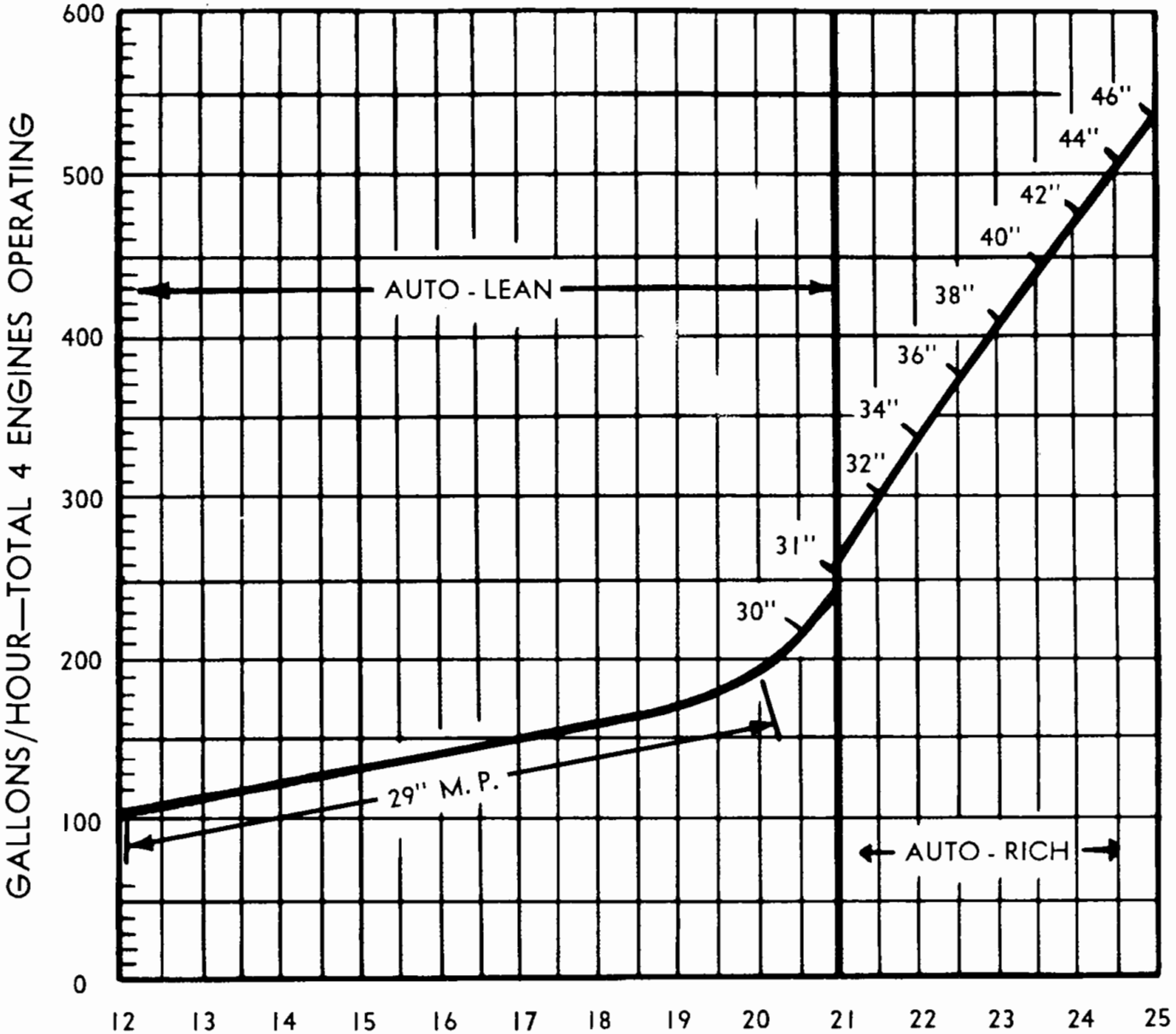
Long Range Cruise Control Charts

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Fuel Temperature Correction Curve

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Fuel Consumption Curve

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APPENDIX III

RESTRICTIONS WITH USE OF 91 OCTANE FUEL

CONDITIONS FOR OPERATION OF ENGINE R-1820-97 ON SPECIFICATION NO. AN-F-26 GRADE 91 FUEL

TAKE-OFF OR MAXIMUM CONDITIONS OF OPERATION

Horsepower	RPM	Manifold Pressure	Mixture Setting
1100	2500	43.5 inches Hg	Full rich

NORMAL RATED POWER

Horsepower	RPM	Manifold Pressure in Hg	Mixture Setting
900	2300	37.0	Auto-rich

MAXIMUM CRUISING — MAXIMUM CRUISE BMEP

Horsepower	RPM	Manifold Pressure in Hg	Mixture Setting
675	2020	31.0	Auto-rich

DESIRED CRUISING

Horsepower	RPM	Manifold Pressure in Hg	Mixture Setting
450	1500	28.0	Auto-lean

Do Not Use Turbo

Although the use of turbosuperchargers is not permitted, if the manifold pressure specified cannot be obtained, the supercharger may be used to obtain the necessary manifold pressure for take-off, but extreme care must be exercised to avoid exceeding the specified limits.

The lightest loads possible will be carried when operating aircraft in accordance with these instructions. Take-off with normal load may not be possible with the restrictions imposed.

The principal concern of operating personnel is the tendency of engines to detonate when operating on fuel of a different grade than that for which the engine was designed. Special care must be taken to see that all spark plugs are operating.

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